

**SPACELAB 1 - METRIC CAMERA**  
**User Handbook and**  
**Data Catalogue**



Prepared for:  
The European Space Agency's Earthnet Programme

by  
Deutsche Forschungs- und Versuchsanstalt  
für Luft- und Raumfahrt E. V.  
Institut für Optoelektronik  
Oberpfaffenhofen, FRG



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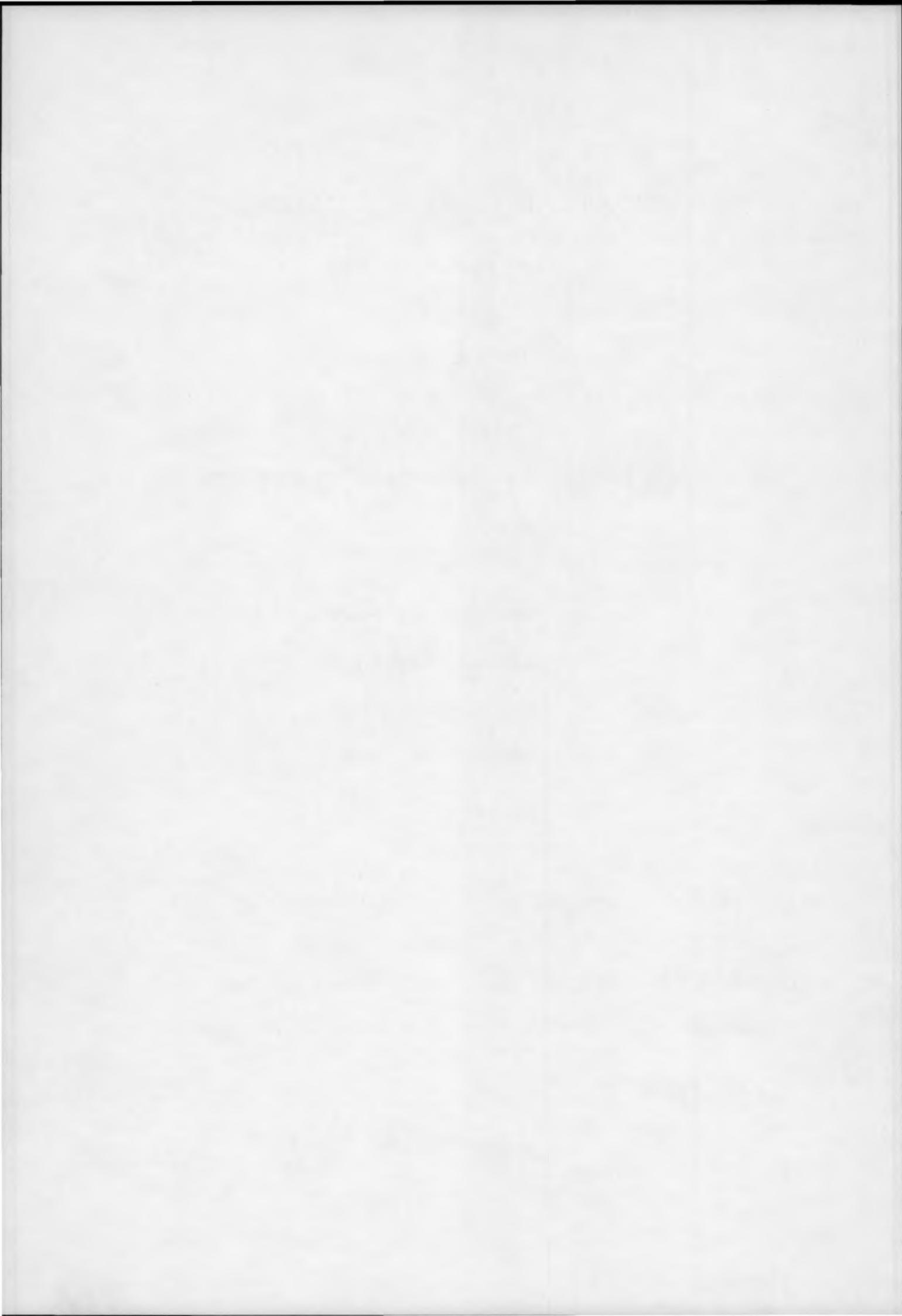
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## 1. Introduction

A Metric Camera was flown on-board ESA/NASA's first Spacelab Mission, which was carried in orbit by the Space Shuttle on STS (Space Transport System) No. 9; the camera was a slightly modified Aerial Survey Camera of the type ZEISS RMK A 30/23.

This Spacelab Mission was originally scheduled for the summer of 1980, but due to various problems with Space Shuttle, the launch was postponed to 1983. Because the camera had to be ready for the 1980 launch date and only a very short period of development was available, no major changes could be introduced to the standard camera system. Also, the waiting period from 1980 to 1983 could not be used for further technical modifications. These were the main reasons why the camera system was not equipped with a Forward Motion Compensation.

Since the launch, originally planned for the summer months, was delayed twice from September to October, and then finally to the end of November, the lighting conditions over the earth became very unfavourable as the sun elevation for all camera operations never exceeded 30°.

The first Spacelab flight actually took place from 28 November to 8 December 1983. The orbit inclination was 57° and the flight altitude 250 km. This altitude and the camera's focal length of 305 mm yields an image scale of approx. 1 : 800 000, which results for the 23 cm × 23 cm image format in a ground coverage of approx. 190 km × 190 km per image frame.

For Metric Camera operations, the Shuttle flew with the open cargo bay orientated towards the earth. In this flight attitude, the camera's optical axis looked exactly vertical down to the earth's surface. Deviations from this vertical observation position were in the order of some tenths of a degree with the exception of Operation No. 32, when the deviations were in the order of several degrees - because this operation was carried out during a Shuttle attitude maneuver.

During the Mission approx. 36 h were flown in this earth oriented attitude, of which 4.5 h were suitable for taking photographs over land. Actually approx. 3 h of these were used for camera operations to expose all the film material that was loaded in the magazines.

Thirty-one (31) camera operations were planned pre-mission of which six had to be skipped due to the launch slip from summer to November. Other pre-planned operations were shortened or deleted during the mission, because cloud cover was predicted over the potential target areas. (Deleted were Operations No. 9 and 10). Operations No. 15-21 were lost due to a film transport problem in one of the magazines. Originally planned to last 8 days, the Mission was extended to 10 days, and gave the opportunity for 4 additional operations (No. 32-35). All together 21 exposure series were taken which varied in duration from 2 to 18 minutes.

One thousand nineteen (1019) photographs were exposed of which 546 were on Colour Infrared (CIR) film and 473 on black and white (B/W) film; the CIR-film was Kodak 2443 - Aerochrome film and the B/W was Kodak Double-X Aerographic film. For both films an orange filter (ZEISS D; 535nm) was used. Images were obtained over 66 countries world wide. Figure 1.1. and Table 1.1 show the ground tracks with respect to the countries over which pictures were acquired during the first week of December 1983. The ground tracks/operations are numbered from 1 to 35. Operations No. 1-13 (Image No. 1-546) used CIR-film, and operations 14 to 35 (Image No. 548-1054) used the B/W-film. The missing numbers indicate how many of the planned operations were deleted due to cloud cover or the film transport problem.

For stereoscopic evaluation, the images were made with at least 60% overlap in the flight direction. Over some regions, 80% overlapping photographs were obtained (Ops. No. 1, 8, 12, 25).

To compensate for the low lighting conditions during the mission, relatively long exposure times of 1/500 to 1/250 sec. had to be used, instead of 1/1000 sec., which was originally planned to reduce the effect of image motion during the time of exposure.

The camera was installed inside the pressurized cabin of the Spacelab Module under controlled environmental conditions. The average cabin temperature was 22°C and the air pressure was approx. 1020 mbar. The air was composed of oxygen and nitrogen to an approximately by normal atmosphere.

During the launch phase - and before reentry - the camera and its magazines were stowed in slide-out drawers in the Module racks. For operation, the camera was unstowed by the crew and mounted over a high quality optical window of 41 mm thickness, through which the photographs were exposed. The operation of the camera was fully automatic, in that the start and stop of every exposure series (operation), as well as, the settings for every single exposure were stored in the on-board master computer and were transferred to the camera via a micro-processor. No exposure meter was used for the exposure control to avoid wrong exposure settings caused by single clouds over the target areas. For operations 32 to 35, the exposure settings and start/stop of the exposure series were made manually by the crew.

During the mission, some operational changes to the preplanned operation sequences were necessary. Exposure of the first film (CIR) went according to plan. However, shortly after the scheduled change-over to the B/W - film magazine, it was detected that the film transport was no longer functioning and a film-jam occurred in the magazine. In a trouble-shooting action between the ground-team and the crew it was possible to clear this film-jam, permitting further operations of the camera.

Following this action it became clear the crew should assist the film take-up by rotating the film transport indicator disk on the outside of the magazine. In doing so they may have moved the film during an exposure event; this probably happened at the end of the Operation No. 25 causing a slight blurring of some image sections and of the fiducial marks.

A post-mission failure analysis revealed that the film-jam was caused by lack of clearance between the take-up spool and the inner wall of the magazine.

At the end of all the possible pre-programmed operations approximately 80% of the available film had been exposed over the planned target areas. The extension of the mission by two days allowed the crew to use the remaining film on a target of opportunity basis. This enabled the four additional Operations (No. 32-35).

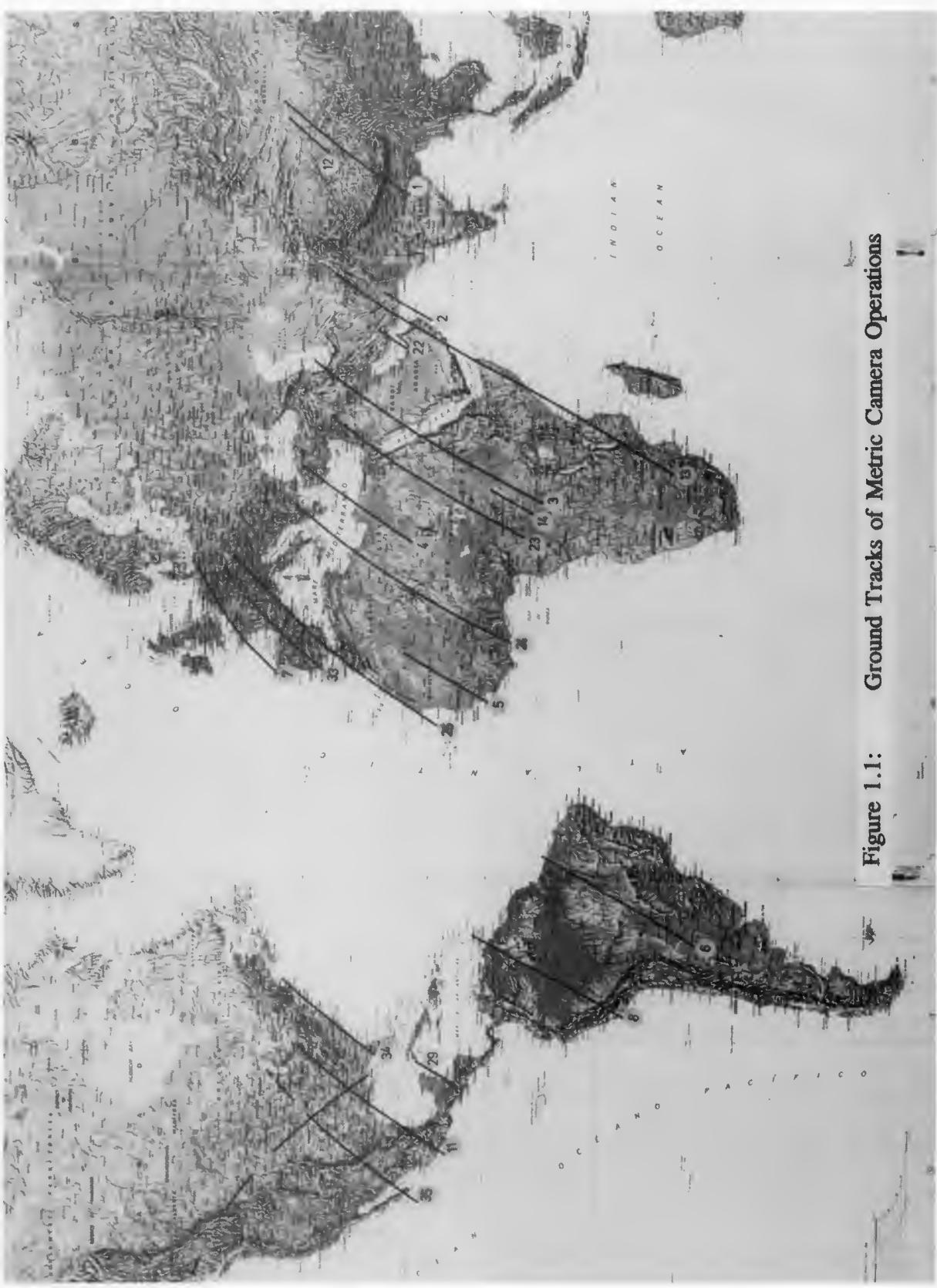


Figure 1.1: Ground Tracks of Metric Camera Operations

**Table 1.1: Metric Camera Operations**

Ops.No	Day	GMT	Film	Image No.	Country
1	336	03:30 to 03:36	CIR	1 - 54	India, Nepal Sikkim,Tibet,China
2		04:57 to 05:00	CIR	56 - 73	Oman, Iran,Pakistan
3		06:21 to 06:33	CIR	74 - 146	Zaire,Central African Republic,Sudan, Saudi Arabia, Iraq, Iran
4		07:58 to 08:03	CIR	150 - 179	Libya, Greece,Turkey
5		09:22 to 09:27	CIR	180 - 209	Guinea, Guinea Bissau,Senegal, Gambia, Mali,Mauritania
6		10:41 to 10:49	CIR	210 - 258	Argentina, Paraguay, Brazil
7		11:03 to 11:08	CIR	259 - 289	Spain, France, Belgium, The Netherlands,Germany
8		12:13 to 12:22	CIR	292 - 358	Peru, Brazil, Venezuela
11		15:22 to 15:30	CIR	359 - 407	Mexico, USA
12	337	03:23 to 03:25	CIR	408 - 431	Tibet, China
13		04:34 to 04:53	CIR	432 - 546	South Africa,Zimbabwe,Mozambique, Malawi,Tanzania,Zanzibar,Somalia, Ethiopia,South Yemen,Oman, Iran,Pakistan,Afghanistan
14		06:11 to 06:13	B/W	548 - 561	Rep. of Congo, Central African Rep.
22	339	04:26 to 04:28	B/W	596 - 607	United Arab Emirates,Iran
23		05:49 to 05:59	B/W	608 - 668	Congo,Centr. African Rep.,Chad,Sudan, Egypt,Israel,Saudi Arabia, Jordan, Syria, Iraq
24		07:19 to 07:30	B/W	669 - 735	Ivory Coast, Ghana, Burkina Faso, Niger,Mali,Algeria,Liberia,Greece
25		08:50 to 09:02	B/W	742 - 889	Mauritania, Western Sahara,Morocco, Spain,France,Italy,Switzerland Austria, Germany
29		13:19 to 13:21	B/W	899 - 911	Honduras, Belize,Mexico,Cuba
32		22:23 to 22:33	B/W	912 - 952	Canada, USA
33	341	08:33 to 08:38	B/W	953 - 976	Portugal, Spain, France, Germany
34		12:58 to 13:03	B/W	983 - 1012	USA
35		14:25 to 14:33	B/W	1013 - 1054	Mexico, USA

## 2. The Camera System

### 2.1 The Components of the Camera System

The camera system is comprised of the following hardware items (Figure 2.1):

- Camera body with optics and mounting brackets.
- Two (2) film magazines containing aerial film of 24 cm width and 150 m length.
- Two (2) Filters (1 Spare)
- Remote Control Unit (RCU)
- Camera Suspension Mount
- Stowage Containers

The camera, film magazines and filters were stowed in special containers in experiment racks during launch and landing (see Fig. 2.1). The camera interfaced to the optically flat High-Quality Window (see 2.1.2.1) via a suspension mount, which was permanently mounted to the window adapter plate. The Remote Control Unit was installed (during pre-launch integration) in an experiment rack and remained there during the whole mission.

In orbit, the camera was assembled on the suspension mount, fitted with a magazine, and electrically connected to the Remote Control Unit. These tasks were carried out on-board by the Payload Specialists. Before the end of the Mission they removed the camera and magazine from the window and stowed them again in the rack for the landing phase.

The camera itself consisted of a modified version of a commercial Aerial Survey Camera of the type Zeiss RMK A 30/23 (Figure 2.2). The characteristics of the camera and the experiment parameters are shown in Table 2.1 and 2.2.

Mechanical and electrical changes of the original RMK A 30/23 were necessary to meet the mission requirements concerning safety, strength, handling under zero gravity and remote control. For mass reduction, the inner lens cone and the focal plane frame were fabricated in cast aluminium instead of steel. The flight camera was equipped with special anti-vibration mounts, locking devices and reinforcing bracketry. The harness was replaced by a non-toxic and non-flammable version. Improved EMC-filters (electro magnetic compatibility) were added and the circuits for remote control were adapted to the Spacelab system requirements.

## THE METRIC CAMERA SYSTEM

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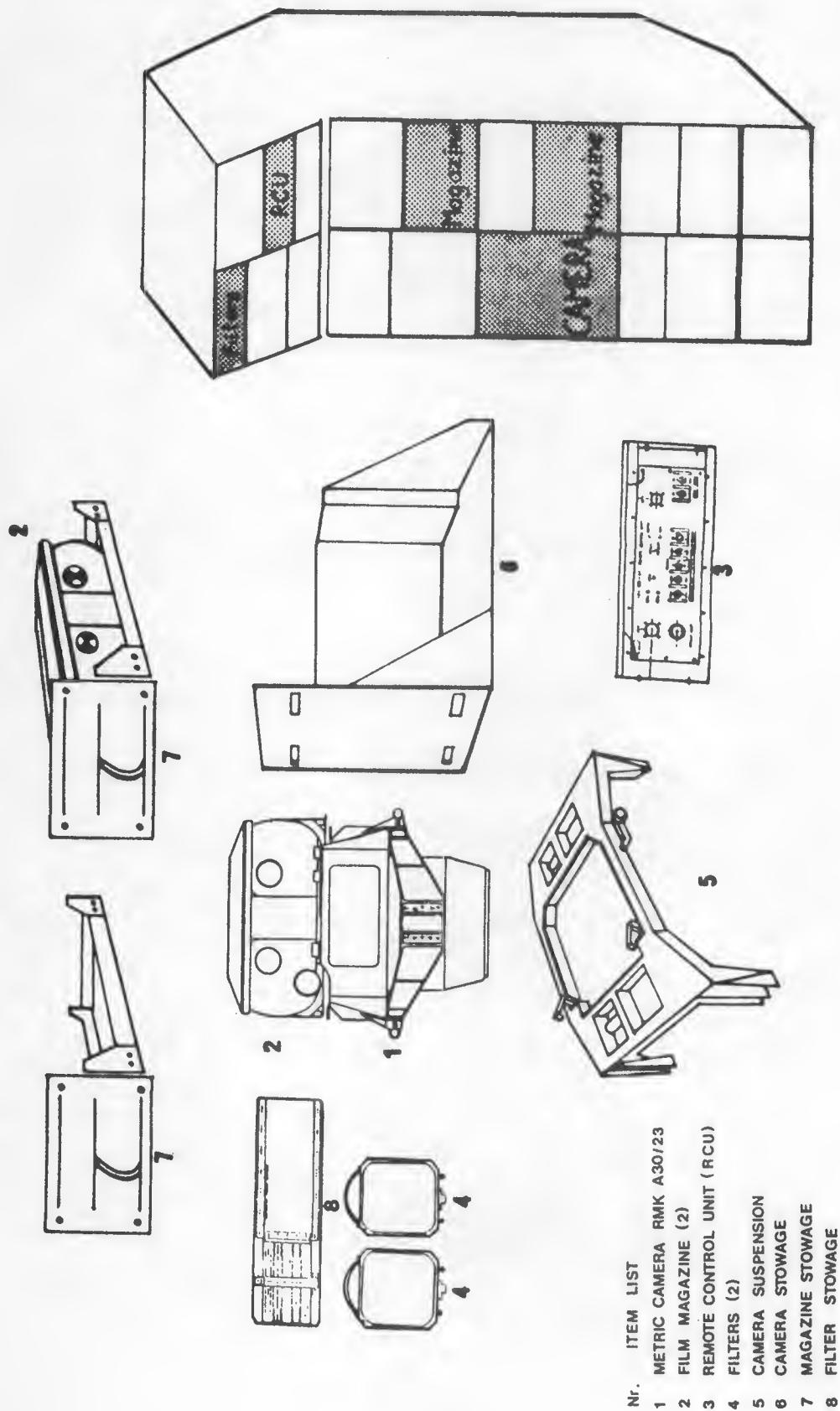


Figure 2.1: Camera System - Components Overview

Table 2.1: Characteristics of the Spacelab/Metric Camera

Type	Modified Zeiss RMK A 30/23
Lens	Topar A 1 with 7 Lens Elements
Calibrated	
Focal Length	305,128 mm (see 2.2)
Max. Distortion	6 Microns (Measured) (see 2.2)
Resolution	39 lp/mm AWAR on Aviphot Pan 30 Film
Film Flattening	By Blower Motor Incorporated in the Camera Body
Shutter	Aerotop Rotating Disk Shutter (Between the Lens Shutter)
Shutter Speed	1/250 sec.- 1/1000 sec. in 31 Steps (see Table 5.4)
f/Stops	5.6 to 11.0 in 31 Steps (see Table 5.4)
Exposure Frequency	4 to 6 sec. and 8 to 12 sec.
Image Format	23 cm x 23 cm
Film Width	24 cm
Film Length	150 m = 550 Image Frames
Dimensions:	
Camera	46 cm x 40 cm x 52 cm
Magazine	32 cm x 23 cm x 47 cm
Mass:	
Camera	54.0 kg
Magazine	24.5 kg (with Film)

Table 2.2: Characteristics of the Experiment

Scales of Images	1 : 770 000 to 1 : 840 000
Image Size	23 cm x 23 cm
Flight Altitude	Approx. 250 km
Ground Coverage of One Image	Approx. 189 km x 189 km
Velocity of Spacecraft above Ground	Approx. 7.5 km/sec.
Exposure Time	1/250 to 1/1000 sec.
Cycling Time	10 sec. for 60% Overlap 5 sec. for 80% Overlap
Image Motion	1/250 sec. : 36 Microns (30 m on ground) 1/500 sec. : 18 Microns (15 m on ground) 1/1000 sec. : 9 Microns ( 8 m on ground)
Base to Height Ratio	1 : 3,30 at 60% Overlap 1 : 1,65 at 80% Overlap (1st and 5th Image)

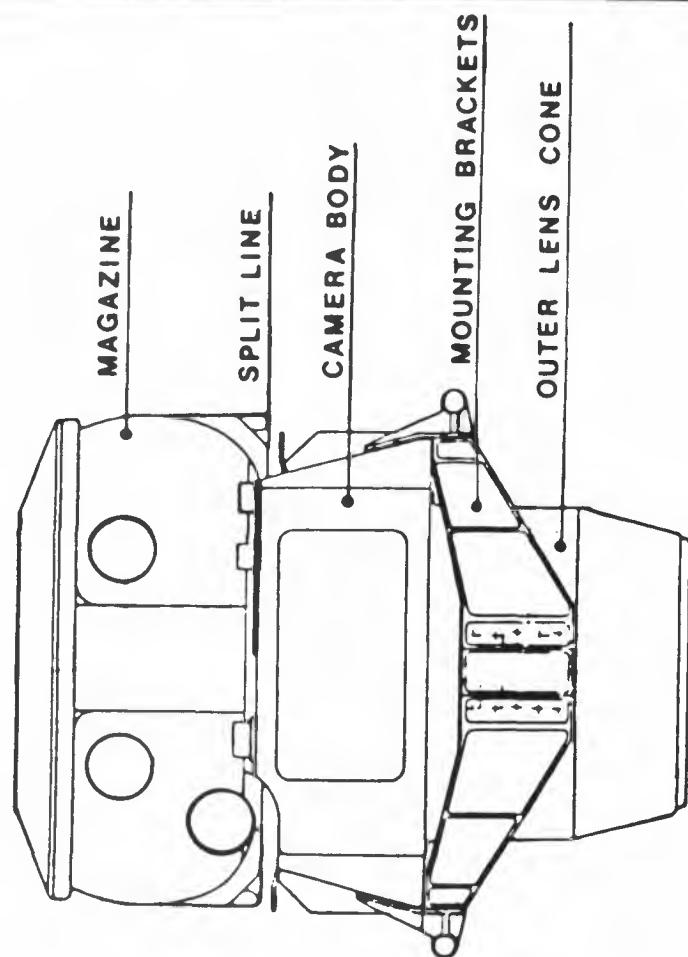
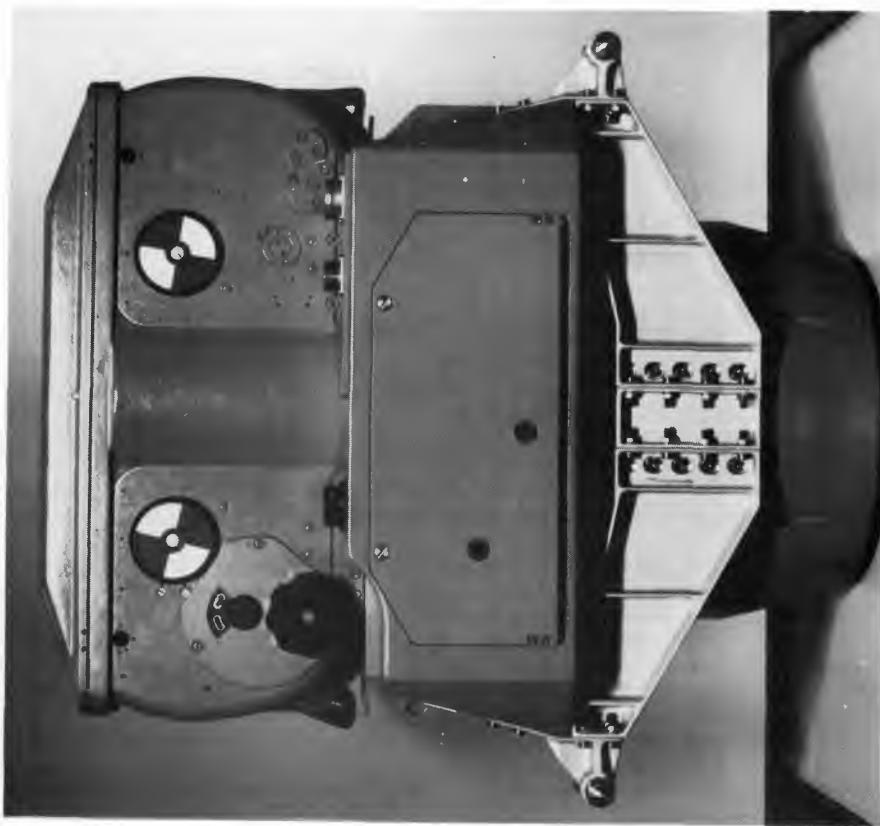


Figure 2.2: Metric Camera Instrument

The camera body consists of a solid casting, and houses the following parts:

- Lens Cone with Shutter and Focal-Plane Frame
- Main Motor with Drive Unit
- Blower Motor
- Display for Auxiliary Data

#### Lens Cone with Shutter and Focal Plane Frame

Together with the focal plane frame, the lens cone forms a single integral unit of high dimensional stability which contains the lens and the shutter. The combination of the lens and focal plane frame in one single unit guarantees an extraordinary constancy of interior orientation, i.e. of the position of the fiducial marks with respect to the projection center of the lens. The lens is the practically distortion free, high-performance Topar A1 f 5.6/305 mm (12").

#### Shutter

The 'Aerotop' shutter (Fig. 2.3) is a rotating-disk type shutter between the front and rear optical elements of the lens. It allows very short exposure times. From the moment the camera is switched on until it is turned off, the shutter blades rotate with a constant speed corresponding to exposure time. From the large number of 'open-time' positions produced during the rotation of the blades (130), one blade (131) which rotates at a lower speed, and a capping disk (162) controlled by the intervalometer, combine to give the desired exposure.

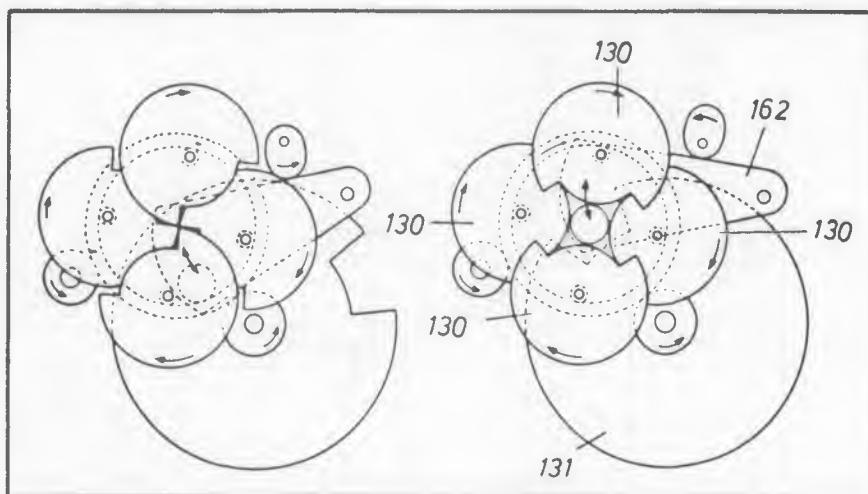


Figure 2.3: The 'Aerotop' rotating Disk Shutter

### Focal Plane Frame

The focal plane frame (Fig. 2.4) serves for holding the film in place at the correct distance from the lens, which is determined by the calibrated focal length. For this purpose, the upper surface of the focal plane frame is polished. Furthermore, the focal plane frame limits the image area to 9" x 9" and carries the four fiducial marks. The point of intersection of the connecting lines between these fiducial marks is adjusted in a way, that will coincide with the principal point of the image, i.e. the point of intersection of the optical axis with the film plane, with an accuracy of about  $\pm 2/100$  mm. The fiducial marks are designed as mechanical pointers and as optical point marks. The distance between two point marks on two opposite sides is 226.00 mm. The checking of this distance on the exposed film is used for the determination of regular film shrinkage. A small 'window' at the side of the focal plane frame is used for recording the auxiliary flight data. These auxiliary data were exposed after the film transportation cycle of each image. The total size of a negative including recorded data is therefore 230 x 250 mm.

### Film Magazine

The film magazine (Fig. 2.5) has the function of providing light-tight accommodation of the film, and to effect film transport, film flattening and accurate location of the film on the focal plane frame. The magazine holds 150m of 4 mils (=0.1 mm) thick film. This corresponds to approximately 550 exposures. The film travels from the supply spool (132) over the movable reversing roller (88), and the stationary reversing roller (87) underneath the platen (89) to the take-up spool (133). Two transport rollers - splined roller (90) and rubber roller (91) - take care of film transport proper and are driven via the coupling from the camera body. The two film spools are of the friction type and are moved according to the amount of film required or supplied, as the case may be. The movable rubber roller (91) is pressed by springs towards the splined roller (90). At the same time, the movable reversing roller (88) actuates the brake provided at the supply spool. This brake is released as soon as the film exercises a slight pull on the roller.

The bottom of the magazine also houses the coupling for the drive mechanism and the connection for the vacuum hose. Both parts are coupled automatically when the magazine is fastened in position or operated.

Between the film spools is the centrally seated pressure plate which presses the film against the focal plane frame during the exposure. It also has the function of keeping the film flat. The surface against which the film is held is drilled with a large number of small orifices connected to a vacuum line so that the film is sucked against this platen. The vacuum is produced by a blower motor within the camera body and delivered to the pressure plate by a hose (101) in the interior of the camera body via the respective magazine coupling.

The film spools are seated between two axle pins (Fig. 2.5) which project through the wall of the magazine. On the outside, the axles are provided with black-and-white signal disks (Fig. 2.5) serving as indicators of film transport during operation. If for instance the film should tear , this malfunction will immediately become visible as only one of the

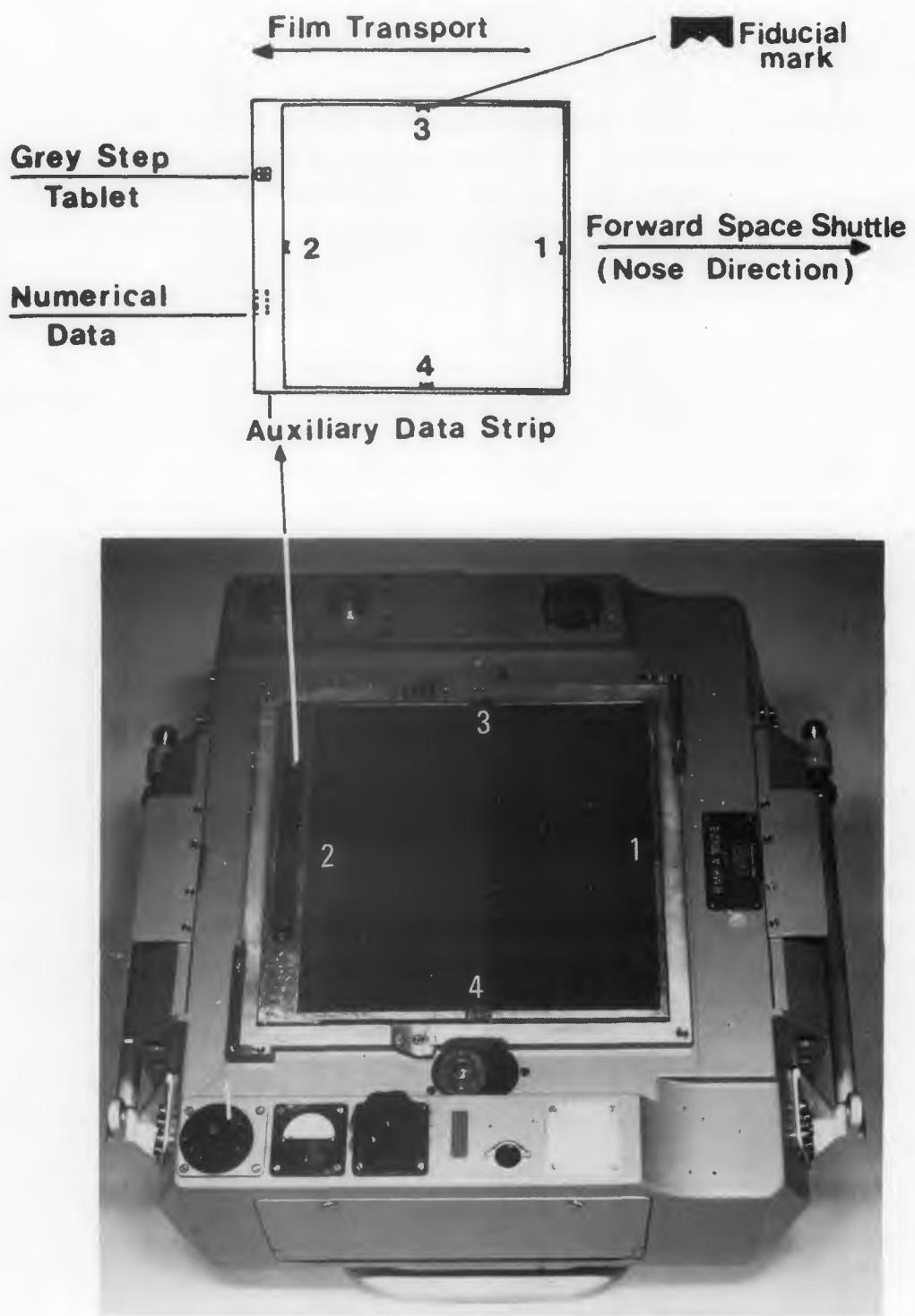


Figure 2.4: Focal Plane Frame

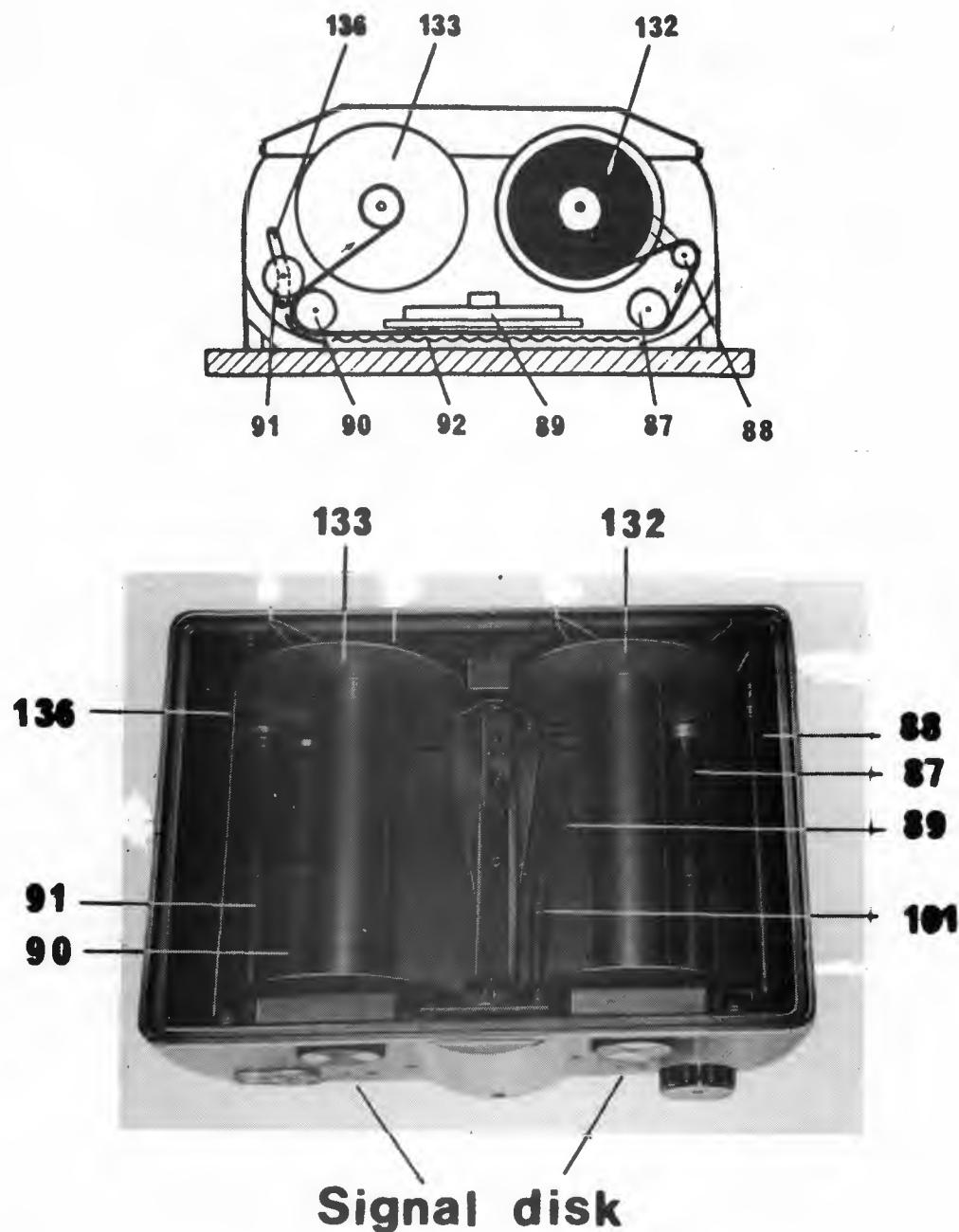


Figure 2.5: Cross Section of the Film Magazine and view of the opened Film Magazine (from above).

136 = Release Bar; 133 = Take-up Spool;  
132 = Supply Spool; 91 = Moving Transport Roller  
90 = Fixed Transport Roller; 92 = Dark Slide;  
89 = Pressure Platen; 87 = Fixed Guide Roller;  
88 = Moving Guide Roller; 101 = Hose

two spool-disks would rotate. (During the Spacelab 1 Mission, after the film jam had been cleared, the crew assisted the film transport after each exposure by rotating the disks of the take-up spool by hand.)

### **Mounting Brackets (Fig. 2.2)**

The Mounting Brackets which were added to the standard camera had three functions:

- To reinforce the lens cone,
- To interface the camera to the suspension mount and,
- To enable better handling of the camera under zero gravity conditions.

### **Auxiliary Data Display**

A new display module with LED's was introduced to print auxiliary data such as image number, time of release event, exposure time and f-number for every image on the film (see 5.2). The exact time of the shutter release is recorded with an accuracy of thousandths of a second. This time signal is triggered by a photodiode which is placed behind the shutter. Beside these auxiliary data, a six-step grey tablet is exposed on every image frame.

The auxiliary data are exposed on the film after shutter release and film transport. The sequence of these three events is as follows:

1. Shutter Release
2. Film Transport
3. Exposure of Auxiliary Data

A positive image with auxiliary data strip, and how it is related to the focal plane frame, and a film negative is shown in Figure 2.6.

### **Remote Control Unit**

The operation of the camera was controlled by a specially developed RCU (Fig. 2.7) which interfaced the Spacelab Command and Data Management System (CDMS) and the Power Distribution System to the camera. The RCU utilized a 16-bit Microprocessor to control the camera operation. The control data sets were loaded from the Mass Memory Unit (MMU) of the CDMS and transferred to the camera via the RCU. The following functions could be software commanded, or manually performed, at the RCU control-panel-switches:

- ON/OFF
- Start/Stop of Serial or Single Exposure
- Adjustment of Overlap (60% or 80%)
- Adjustment of Exposure Time and Aperture

During manual operations the exposure time and aperture had to be set at the camera body using adjustment control knobs. For data recording and retrieval purposes the RCU also performed:

- Tagging of Shutter Release Time
- Conditioning of House-Keeping Data
- Conditioning and Control of Power Supply
- Control of Auxiliary Data Recording

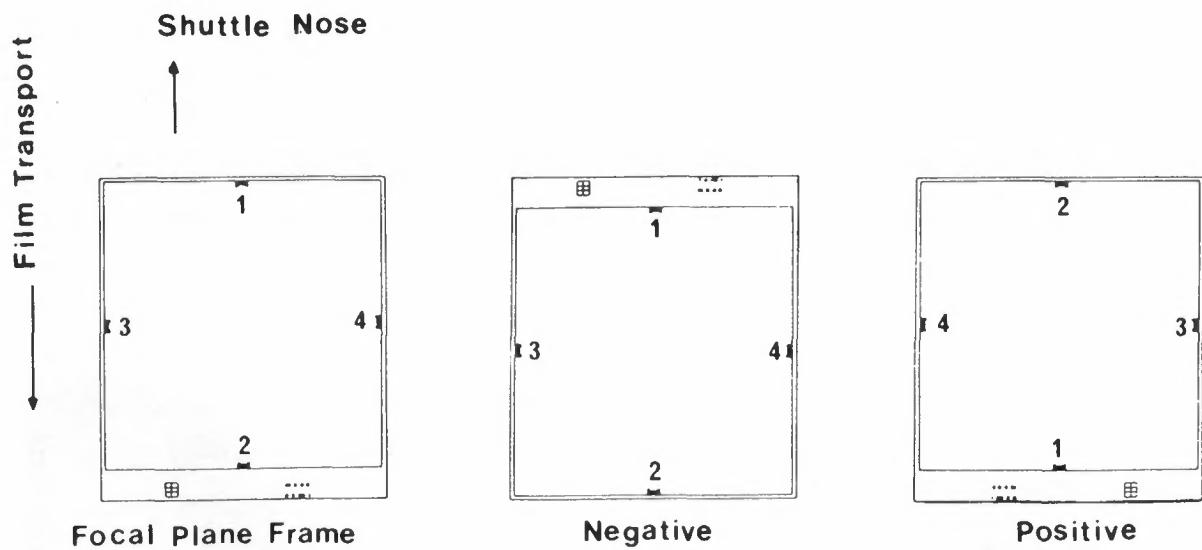


Figure 2.6: Relation between Focal Plane Frame and Negative and Positive Film. (1, 2, 3, 4 are the Fiducial Marks)

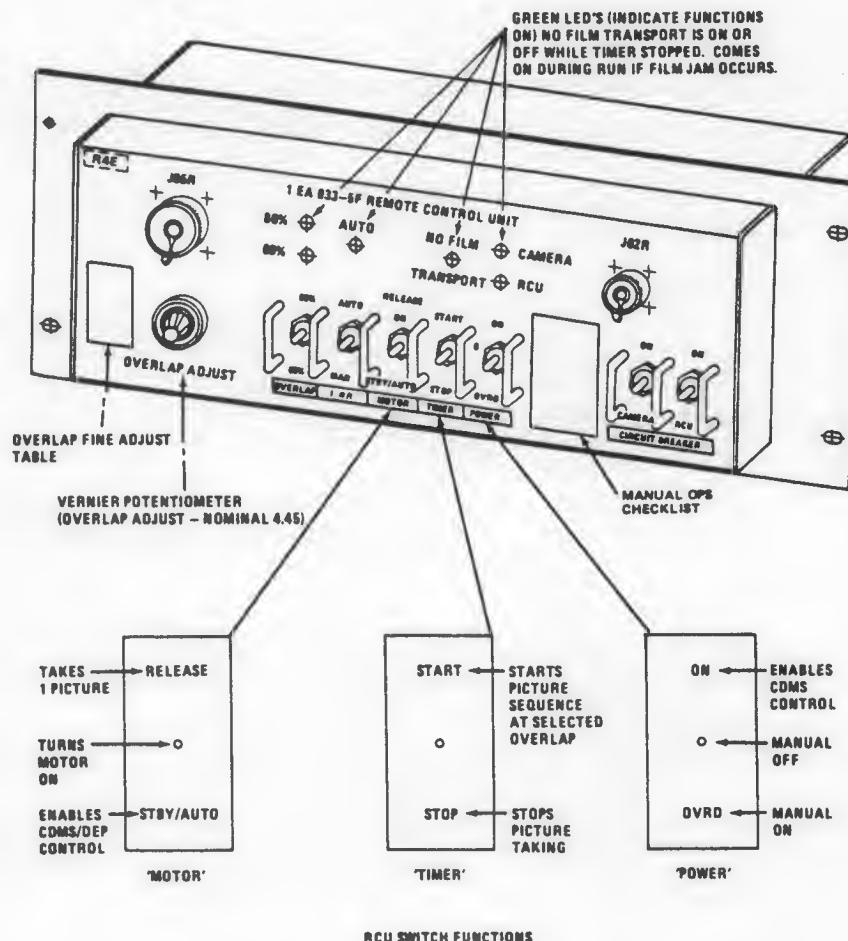


Figure 2.7: Remote Control Unit (RCU)

### 2.1.1 Operation and Control

The Metric Camera is designed to be operated either automatically or manually. The automatic operation mode was the nominal case and used the on-board computer system, whereas the manual mode was foreseen more as a contingency operation mode.

The installation of the camera over the high quality optical window was performed on-board by the crew. Their further duties were the mid-mission exchange of film magazines and stowage of the camera system on completion of the experiment.

The exposure settings for each picture were pre-mission calculated and stored in the on-board computer system. These data were derived from a program which used the sun angle, an average earth reflectance coefficient, film sensitivity, a filter factor and the nominal GMT to compute the f-stop and shutter speed for each picture.

The computer system, part of the Spacelab Command and Data Management System (CDMS), was instructed by a dedicated software (ECAS = Experiment Computer Application Software) to apply the pre-flight defined/stored exposure data for each photo; this consisted of an overlap (60% or 80%), f-stop and exposure instruction in coded form. The camera's performance was also monitored by the CDMS over a range of various parameters, and the data were recorded at the Houston Control Center.

All of the exposure instructions for the automatic operation of the camera were routed through a microprocessor in a Remote Control Unit (RCU) which was installed in a Spacelab rack separate from the camera. An interconnecting cable harness carried the experiment data to and from the CDMS, and supplied the requisite power for the camera and RCU.

The control data for each sequence of exposures, known as Cycle Control Data (CCD) - comprising the exposure data, number of pictures in the respective sequence, and the start GMT - were stored in the Mass Memory Unit (MMU) of the CDMS and retrieved by the ECAS on command from the computer just prior to each picture taking sequence.

The picture sequence or operations contained a number of exposures depending on the target size (geographically) to produce a series of overlapped pictures taken at 5 second (80% overlap) or 10 second (60% overlap) intervals. The path or ground track that these series of pictures covered can be seen in Fig. 1.1.

On an average, a cycle comprised around 50 pictures with the shortest run being covered in 13 consecutive frames and 148 for the longest.

During the mission the camera operations were scheduled in a so-called Mission Timeline (pre-flight computed) which basically allocated the resources needed by the camera (e.g. crew, power, computer time) along with all the other experiments. This computer controlled time-table initiated the start/stop times of the experiment via its computer and the CDMS.

The manual mode of operation was intended more as a contingency mode in cases where the computer system fails or the demand on the computer time (by other experiments) exceed the buffer capacity.

Manual operations were indeed resorted to on the 9th mission day when runs were made during periods for which no nominal camera operations were planned. In these runs the crew had to set in the camera functions by hand, using exposure parameters supplied in the on-board Flight Data File and selecting those data which fitted the sun elevation at the time.

The Metric Camera ground team in the Houston Payload Operations Control Center (POCC) monitored the whole 10 day mission and could interact with the experiment when satellite radio-contact with the Shuttle/ Spacelab was possible. During these so-called Acquisition of Signal (AOS) periods for instance, changes to the on-board stored exposure parameters could be uplinked (via the command system) as well as picture inventory updating (e.g. total picture counter corrected). These changes were needed in some cases to account for deleted or altered exposure sequence due to cloud cover forecasts.

### 2.1.2 Integration of the Metric Camera into Spacelab

Spacelab is a manned and reusable laboratory developed by ESA to be carried into orbit by the Space Shuttle. Spacelab consists of two elements: a pressurized module with several segments and a non-pressurized instrument platform consisting of up to five pallets. During the whole mission Spacelab remains within the cargo bay of the Space Shuttle (Fig. 2.8).

The Metric Camera was part of the payload in the pressurized Spacelab module. For operation, the camera was mounted by the crew over a High Quality Window (HQW) through which the exposures were made. The HQW is an integral part of the Spacelab Window Adapter Assembly (SWAA).

The SWAA (Fig. 2.9) was mounted on the Spacelab module opening, and contained a Skylab S-190A high quality window for photographic or other instrument usage, and a Spacelab Viewport (VP). The assembly includes external shields and a thermal protection, with removable internal safety shields latching and actuation mechanisms for the external shields with latch status indications heaters and temperature sensors/controls for maintaining proper thermal conditions. Two handrails for crew mobility/stability were also provided.

The camera was installed over the HQW on a special developed suspension mount shown in Fig. 2.10 and bolted to the SWAA at four points (Fig. 2.9). This mount was equipped with a tilt mechanism by which the camera could be swiveled and 'parked' in an oblique position for easy exchange of lens cover, filters, and removal/refitting of the inner safety cover of the HQW. In the nominal working position the optical axis of the Camera looked vertically through the window in the  $-Z_B$  - axis of the Orbiter (Fig. 2.8).

The relative position of the camera with respect to the Space Shuttle and the HQW is shown in Figures 2.11 and 2.12.

The intersection of the optical axis of the camera with the inner surface of the window was displaced from the window center towards the  $-X_B$  - direction by 102 mm (Fig. 2.11). The direction of film transport was in the  $-X_B$  - direction.

The distance between the outer lens surface and the window was 19 mm and the distance of the lens' point of divergence and the window was 199 mm (Fig. 2.12).

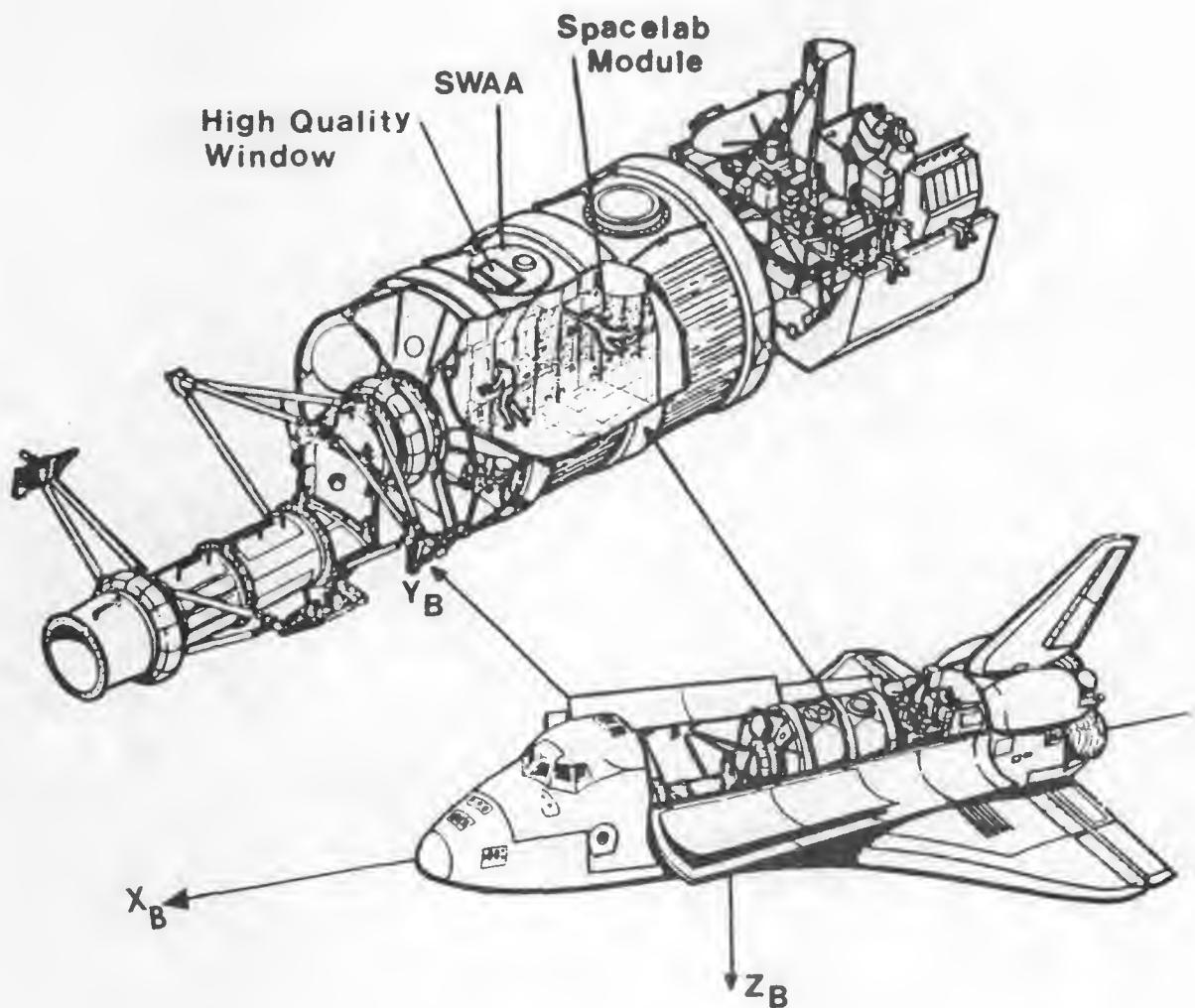


Figure 2.8: Spacelab-1 Mission Configuration

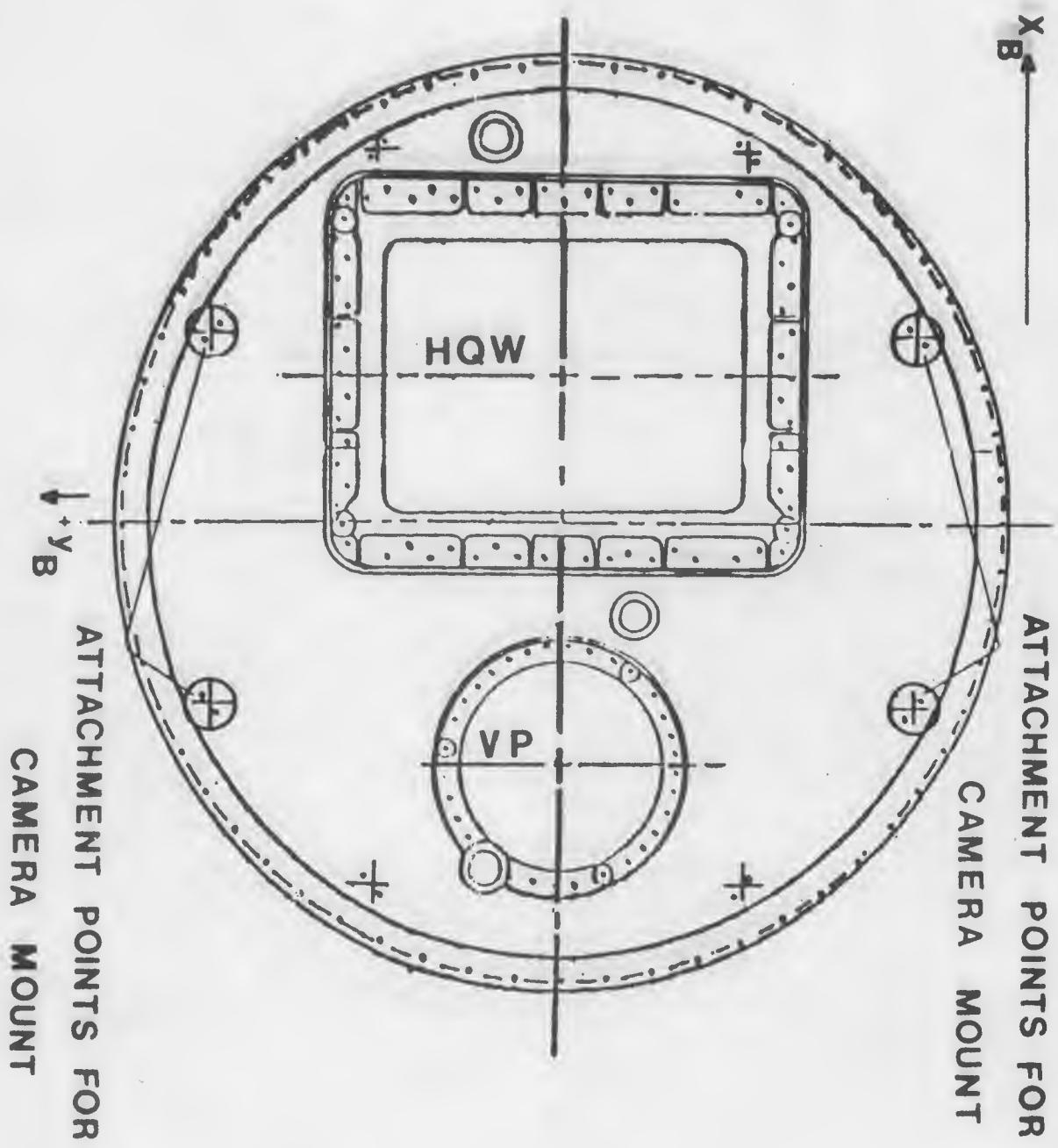
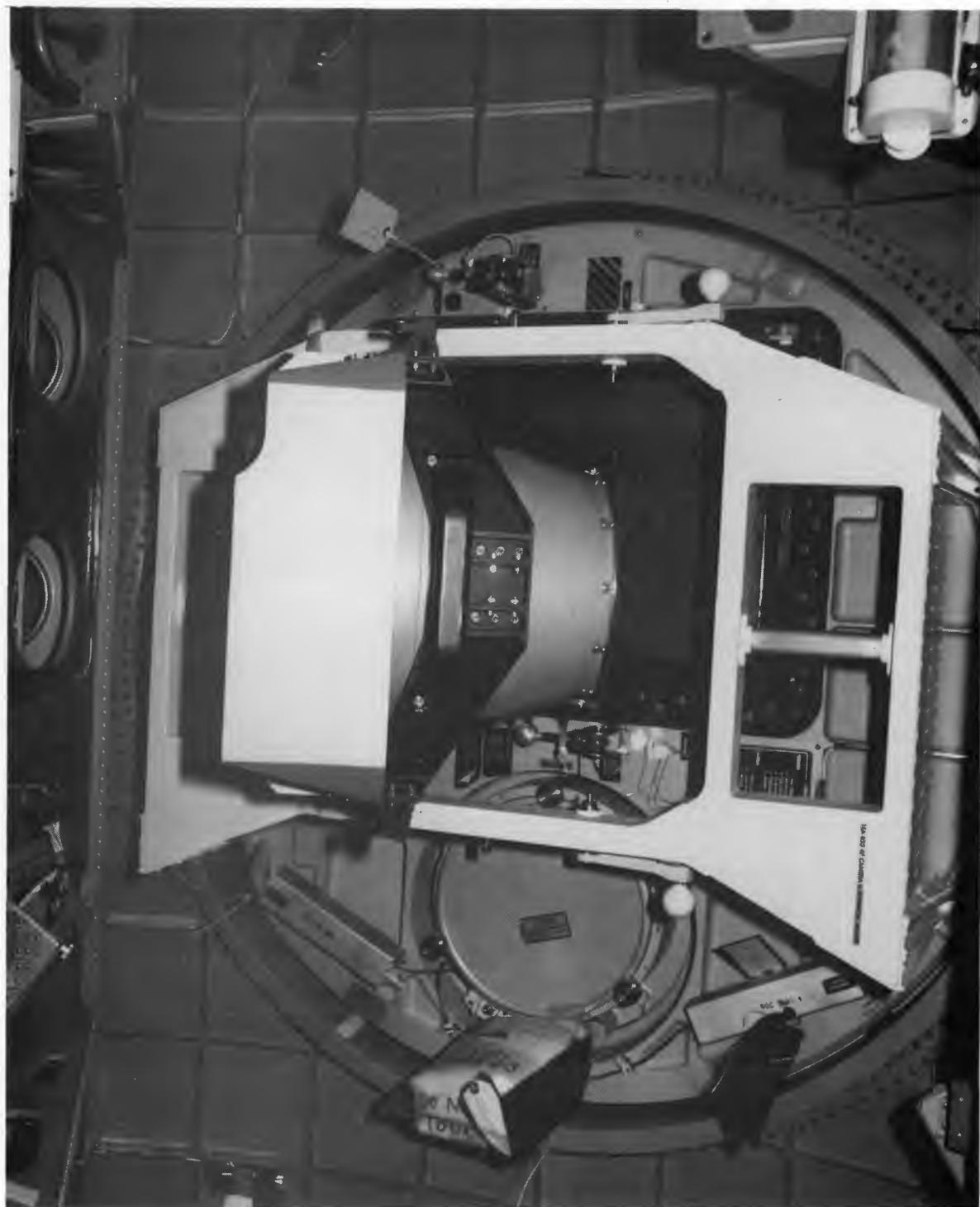


Figure 2.9: Spacelab Window Adapter Assembly. (Internal View)  
HQW: High Quality Window; VP: View Port.



**Figure 2.10:** Camera Suspension Mount with Camera Mock-up  
in tilted position installed over High Quality Window  
in Spacelab (Photo: NASA)

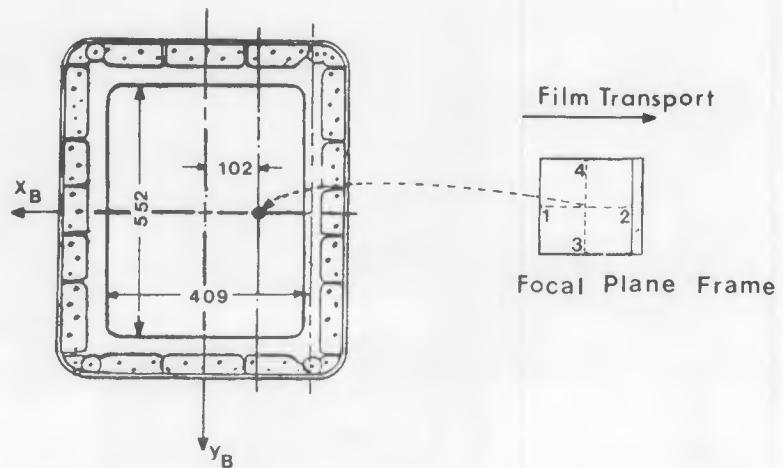


Figure 2.11: Position of the focal plane frame with respect to the HQW and the Space Shuttle Body Coordinate System (Internal view on the window surface). Viewing direction of the Camera is  $-Z_B$ , that is vertical through the paper plane from the observer. (All distances in mm)

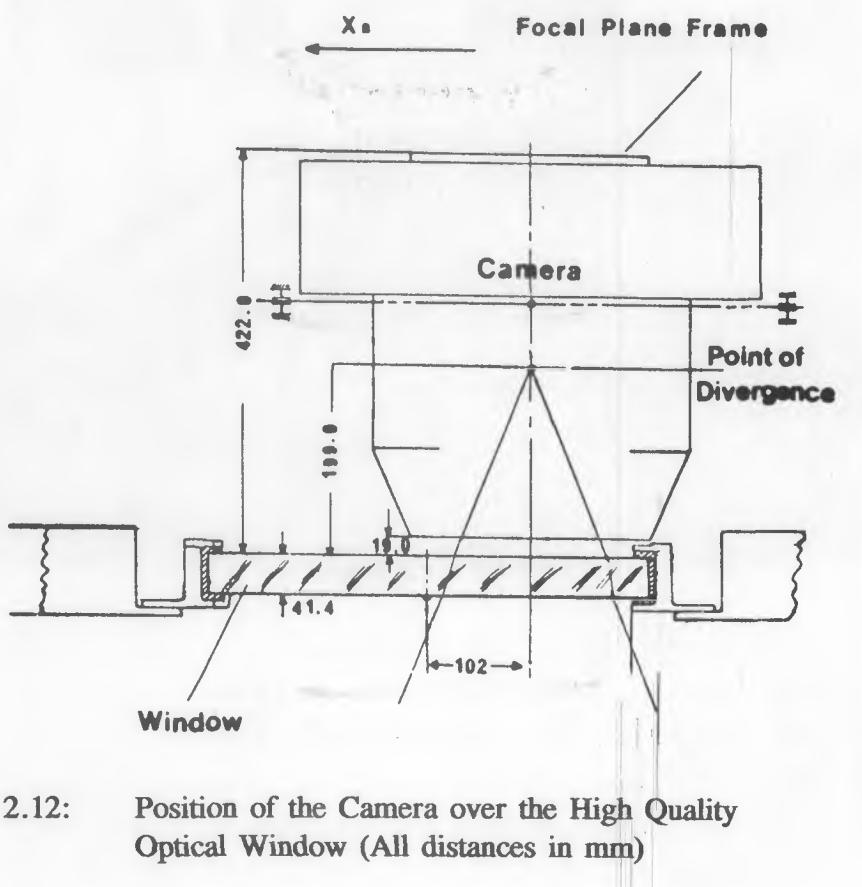


Figure 2.12: Position of the Camera over the High Quality Optical Window (All distances in mm)

### 2.1.2.1 The Optical Window of Spacelab

A spare unit of the Skylab Optical Window (left-over from the Skylab program) was requalified for use with Spacelab. An adapter assembly enabled the mounting of the window assembly in Spacelab and the viewport in the 1.3m diameter opening in the module as shown in Figure 2.9. This assembly is known as the SWAA.

The high quality window itself consists of a single 41 mm thick plane of BK-7 glass of rectangular (41 x 55 cm) shape enclosed in a molded seal and supported by a flexible spring system in an aluminum frame. Window cross section details are shown in Figure 2.13. The window is equipped with a heater system that controls window temperature to minimize thermal gradients across the glass, and thus maintain optical performance. The outer surface of the glass has a 40 W electric conductive film (ECF) heater and two 100 W heater elements are mounted in the frame surrounding the glass.

An external manually operated cover is installed to protect the window glass from radiation, meteoroid impact and contamination during periods when the window is not in use. A removable safety shield is mounted over the inside of the window when the window is not in use.

The window's transmission curve from 300 to 1000 nm are shown in Figure 2.14. Other optical characteristics are given in Table 2.3.

#### Viewing Area:

The window has  $2245 \text{ cm}^2$  ( $348 \text{ in}^2$ ) of viewing area. There are two areas  $1 \times 19 \text{ cm}$ , as shown in Figure 2.13 where temperature sensors and their associated wiring are mounted to the outside of the glass. To avoid interference of the camera's field of view with these sensors the position of the camera was selected as shown in Figure 2.11 and 2.12.

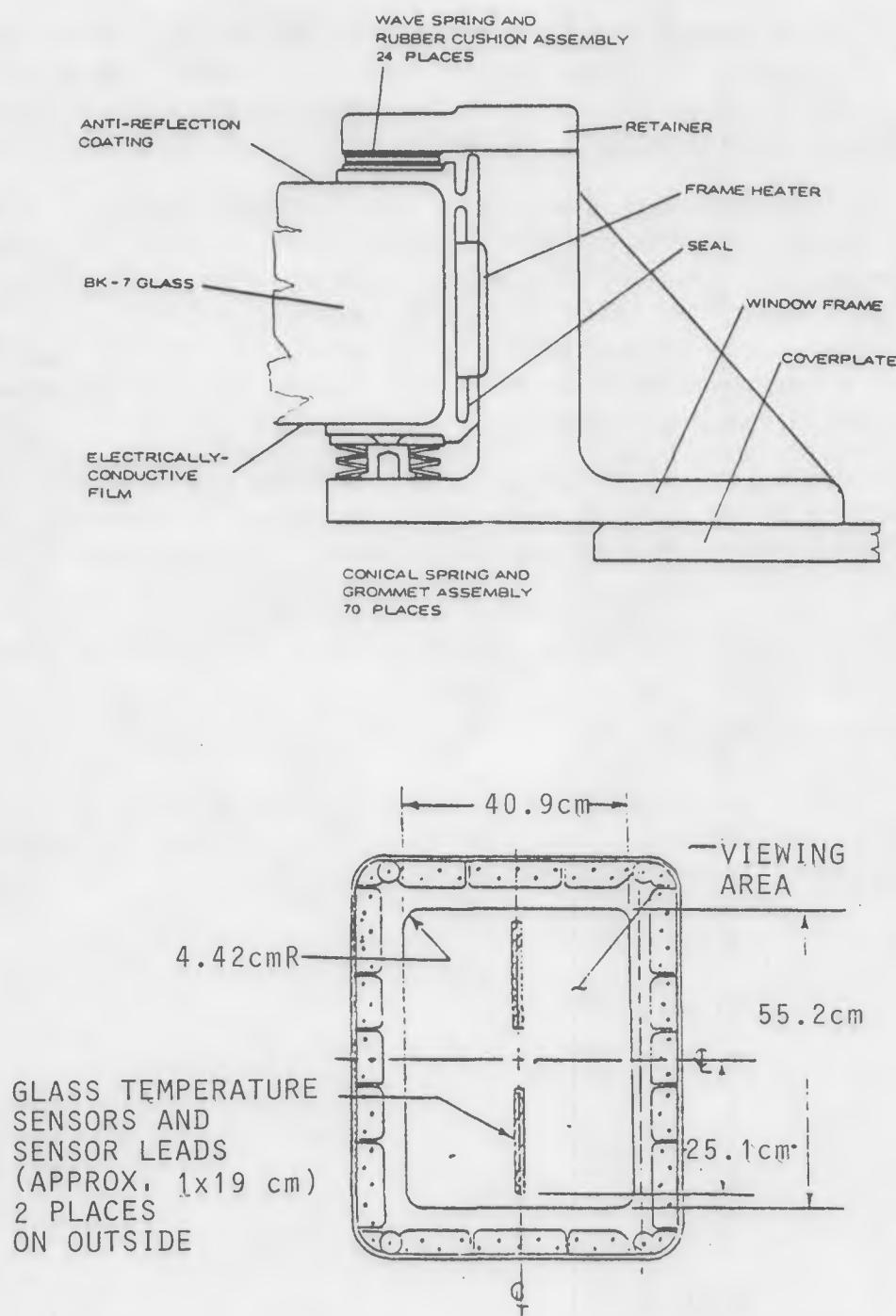


Figure 2.13: Detail Section through the Window Frame and Location of Temperature Sensors

Table 2.3: Optical Characteristics of the Window

OPTICAL CHARACTERISTICS	WINDOW PERFORMANCE
Parallelism	2 arc sec
Reflectance	2 % on Inside 4 % on Outside
Seeds and Bubbles	Total Area: $0,1 \text{ mm}^2 / 100 \text{ cm}^3$ of glass Maximum Dimension: 0,76 mm of Single Imperfection
Surface Quality	60 - 40 or better as defined in MIL-13 830

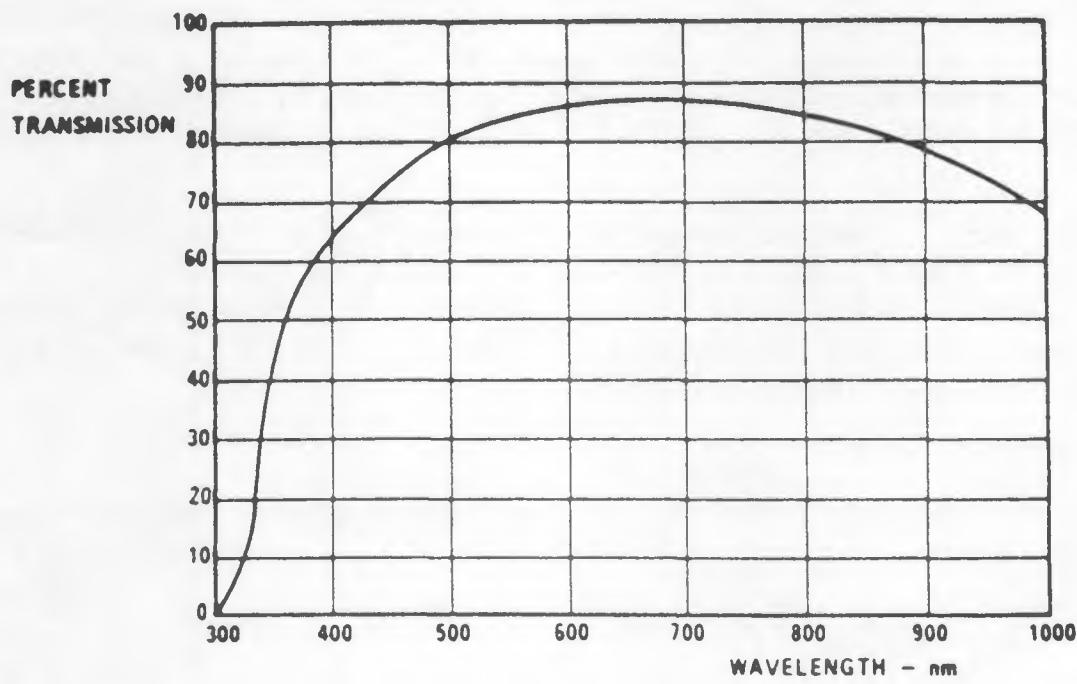


Figure 2.14: Window Spectral Transmission

### 2.1.2.2 Environmental Conditions inside the Spacelab Module

The cabin air was composed of nitrogen and oxygen, and the percentage of each was close to that of normal air on earth with  $O_2$  partial pressure of 0,22 bar. The relative humidity was maintained between 30% and 70%.

The average ambient cabin temperature during camera operations was approx. 22°C. The air pressure varied between 1016 and 1027 mbar.

The actual measured values for cabin temperature and air pressure can be found in Fig. 2.15 and 2.16. Superimposed in these Figures are the camera operation periods, indicated as black bars.

## 2.2 Camera Calibration

The camera was calibrated twice, pre-mission and post-mission.

The pre-mission calibration was carried out in September 1980, about three years before the flight. In the meantime, the camera was kept and stored under controlled conditions. The camera was operated during this time only in the laboratory for functional tests. The camera was shipped from Germany to the USA in August 1982. The post-mission calibration was carried out six months after the mission in May 1984. Between the end of the mission and the calibration the camera was shipped back to Germany and again stored under controlled conditions.

The main difference established between both calibrations, is that the distance between the fiducial marks 3 and 4 increased by 0,029 mm. The camera was tested in accordance with the existing regulations, and the methods used were based on the recommended procedures for Calibrating Photogrammetric Cameras and for Related Optical Tests (International Society of Photogrammetry, 1960, reaffirmed 1964).

### Calibrated Focal Length

The calibrated focal length is chosen so as to minimize the square sum of the radial measured distortion.

### Distortion

The values of radial distortion refer to the calibrated focal length and to the principal point of symmetry. A positive value indicates that the image is further from the center than its distortion free position.

The radial distortion is measured for points of the focal plane separated by 10 mm from the axis for each of the 4 radii A, B, C, and D (Fig. 2.17). AV is the average radial measured distortion at a given radial distance. Measurements are made at maximum aperture on the goniometer by attaching the filter D (cut-off wavelength 535 nm at transmittance 50%). The standard deviation of the distortion values given can be assumed to be less than 0,002 mm.

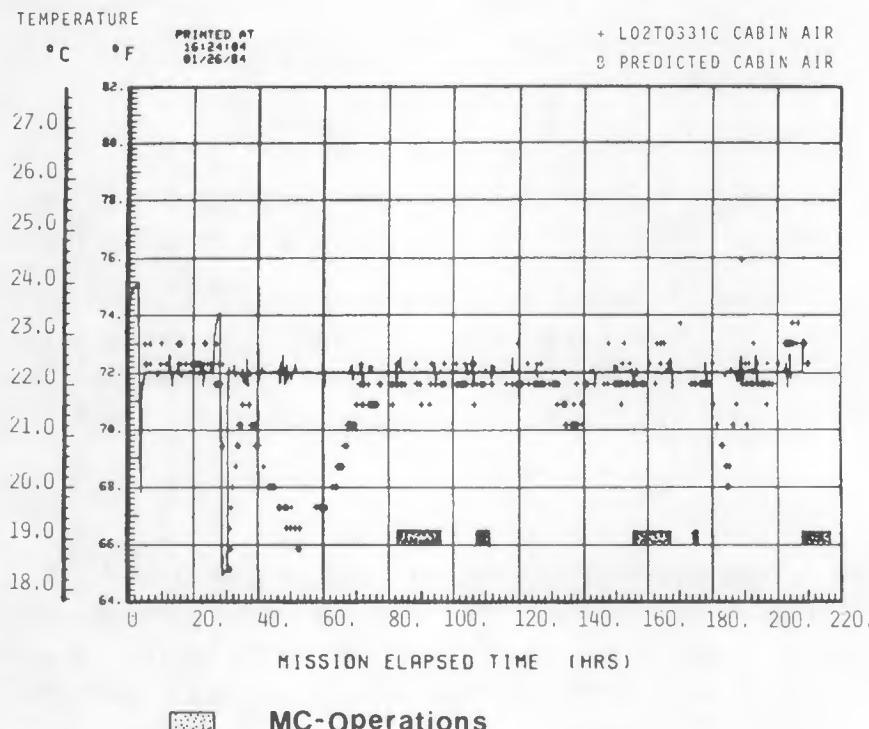


Figure 2.15: Actual and Predicted Cabin Air Temperature

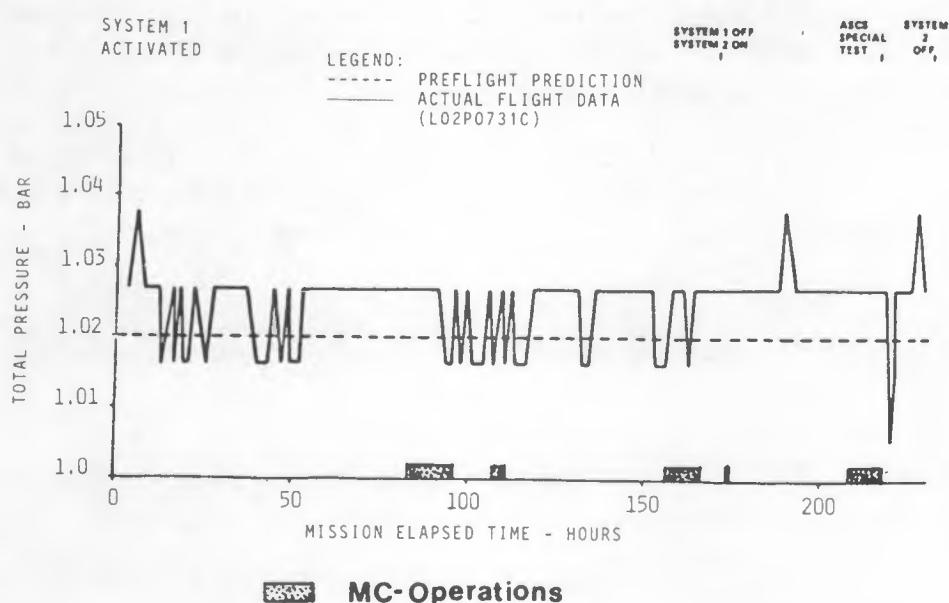


Figure 2.16: Spacelab total Pressure versus Time

The maximum tangential distortion, i.e. the displacement of the central image from a straight line connecting corresponding image points at equal but opposite angular separations from the axis, does not exceed 0.005 mm.

#### **Principal Point and Fiducial Center**

The positions of the principal point of auto-collimation and of the fiducial center are given in a rectangular coordinate system as shown in Fig. 2.17, with the principal point as origin.

Regarding the origin for distortion values it must be realized that in the photogrammetric process, the asymmetry due to a displacement of that point is eliminated together with the asymmetry introduced by camera tilt. The principal point of symmetry is chosen as origin for distortion, because only this residual asymmetry cannot be eliminated by single compensation.

#### **Fiducial Marks**

Coordinates of the fiducial marks are given in a rectangular system as shown in Fig. 2.17, with the principal point of symmetry as origin. Fiducial marks are numbered from 1 to 4 (Fig. 2.17). The lines joining opposite pairs of fiducial marks intersect at an angle within 30 seconds of  $90^\circ$ . The point of intersection (fiducial center) is with 0.02 mm of the principal point of auto-collimation. The location of the fiducial marks can be assumed to be accurate within 0.005 mm.

#### **Photographic Resolving Power**

The resolving power is obtained by photographing a series of three line test figures. The difference of log luminance between the lines and the background is 1.6. The photographs are taken under the recommended standard illumination by using the filter B (cut-off wavelength 480 nm at transmittance 50%) in parallel light. The camera is used at full aperture.

The resulting image is examined with a low power stereoscopic microscope to find the spatial frequency of the smallest pattern resolved. The values of resolving power are reduced to the image plane and refer to the focus setting as used for determining the calibrated focal length.

#### **Filters**

The two surfaces of the filter listed in the certificate are within 5 seconds of being parallel.

#### **Magazine Platen**

The platen mounted in FK 24/120 film magazine, serial no. as indicated in the certificate (2.2.1 and 2.2.2), does not depart from a true plane by more than 0.010 mm.

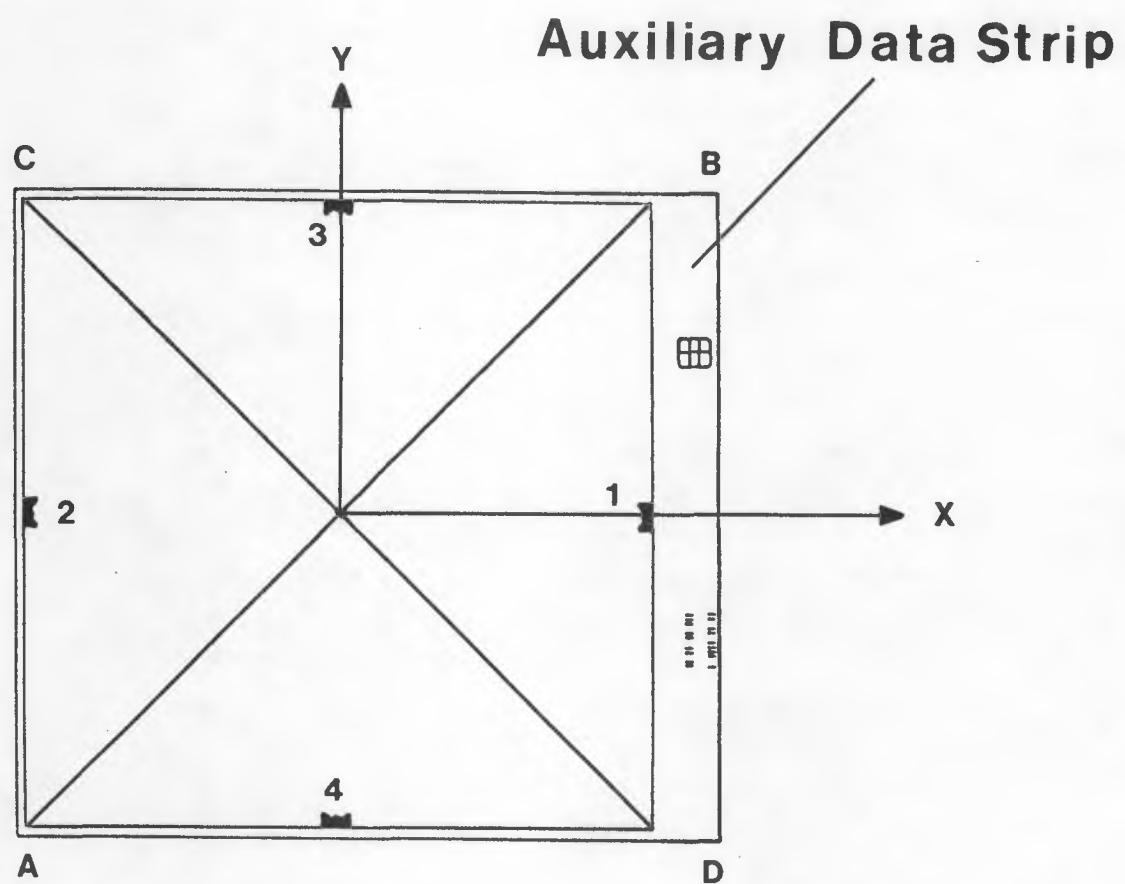


Figure 2.17: Metric Camera Image (Positive) with Fiducial Marks  
1,2,3,4 and four radii A,B,C and D along which the distortion  
was measured

### 2.2.1 Pre-Mission Calibration

C A R L Z E I S S  
O B E R K O C H E N / W U E R T T .

#### C A L I B R A T I O N   C E R T I F I C A T E F O R P H O T O G R A M M E T R I C   C A M E R A S

C A M E R A   T Y P E : R M K   A   3 0 / 2 3  
L E N S   T Y P E : T O P A R   A 1  
M A X . A P E R T U R E : F / 5 . 6

S E R I A L   N O .   1 2 4 5 0 1  
S E R I A L   N O .   1 2 4 5 1 5  
N O M .   F O C A L   L E N G T H : 3 0 5   M M

1) C A L I B R A T E D   F O C A L   L E N G T H = 3 0 5 . 1 2 8   M M

2) D I S T O R T I O N / 0 . 0 0 1   M M , R E F E R R I N G   T O   P . P .   O F   S Y M M E T R Y   P P S

S/M M	0	10	20	30	40	50	60	70	80	90	100	110	120	130	140	150
A	0	2	2	1	3	1	-1	1	-2	-2	-2	-2	0	-1	0	1
B	0	-1	0	-2	0	-2	-1	-3	-4	-4	-3	-6	-1	0	4	3
C	0	2	4	2	2	1	3	2	-1	-1	0	0	2	2	2	1
D	0	0	0	-1	-2	-3	-1	-4	-2	-4	-2	-4	2	2	4	6
A V.	0	1	2	0	1	-1	0	-1	-2	-3	-2	-3	1	1	3	3

3) P . P .   O F   A U T O C O L L I M A T I O N   A N D   F I D U C I A L   C E N T R E , R E F E R R I N G   T O   P P S

P . P .   O F   A U T O C O L L I M A T I O N   P P A   X = . 0 4 2   Y = . 0 3 7   M M  
F I D U C I A L   C E N T R E   F C   X = . 0 3 4   Y = . 0 4 5   M M

4) F I D U C I A L   M A R K S , R E F E R R I N G   T O   P P S

X 1 = 1 1 3 . 0 3 3   X 2 = - 1 1 2 . 9 6 6   X 3 = . 0 3 4   X 4 = . 0 3 5   M M  
Y 1 = . 0 4 5   Y 2 = . 0 4 6   Y 3 = 1 1 3 . 0 3 5   Y 4 = - 1 1 2 . 9 5 8   M M  
D I S T A N C E S   1 - 2 = 2 2 5 . 9 9 9   3 - 4 = 2 2 5 . 9 9 3   M M

5) P H O T O G R A P H I C   R E S O L V I N G   P O W E R , I N   C Y C L E S   P E R   M M

A R E A   W E I G H T E D   A V E R A G E   R E S O L U T I O N   3 9

F I E L D   A N G L E / D E G = 0   7   1 4   2 4

R A D I A L   L I N E S   4 5   4 4   3 4   3 6  
T A N G E N T I A L   L I N E S   4 5   4 4   4 2   3 7

F I L M : A V I P H O T   P A N   3 0   S P E E D   2 1   D I N  
D E V E L O P E D   I N   U L T R A F I N   1 + 1 5

6) F I L T E R S

K L ( C L E A R )   N O .   ---  
B ( Y E L L O W )   N O .   1 2 4 5 3 5  
D ( O R A N G E )   N O .   1 2 4 5 4 7

7) M A G A Z I N E   P L A T E N

F K   2 4 / 1 2 0   N O .   1 2 4 8 3 1 , 1 2 7 6 3 0 , 1 2 7 6 4 8

A B T E I L U N G   F U E R   G E O D A E S I E   U N D   P H O T O G R A M M E T R I E

I . A .   *R. Schwebel*

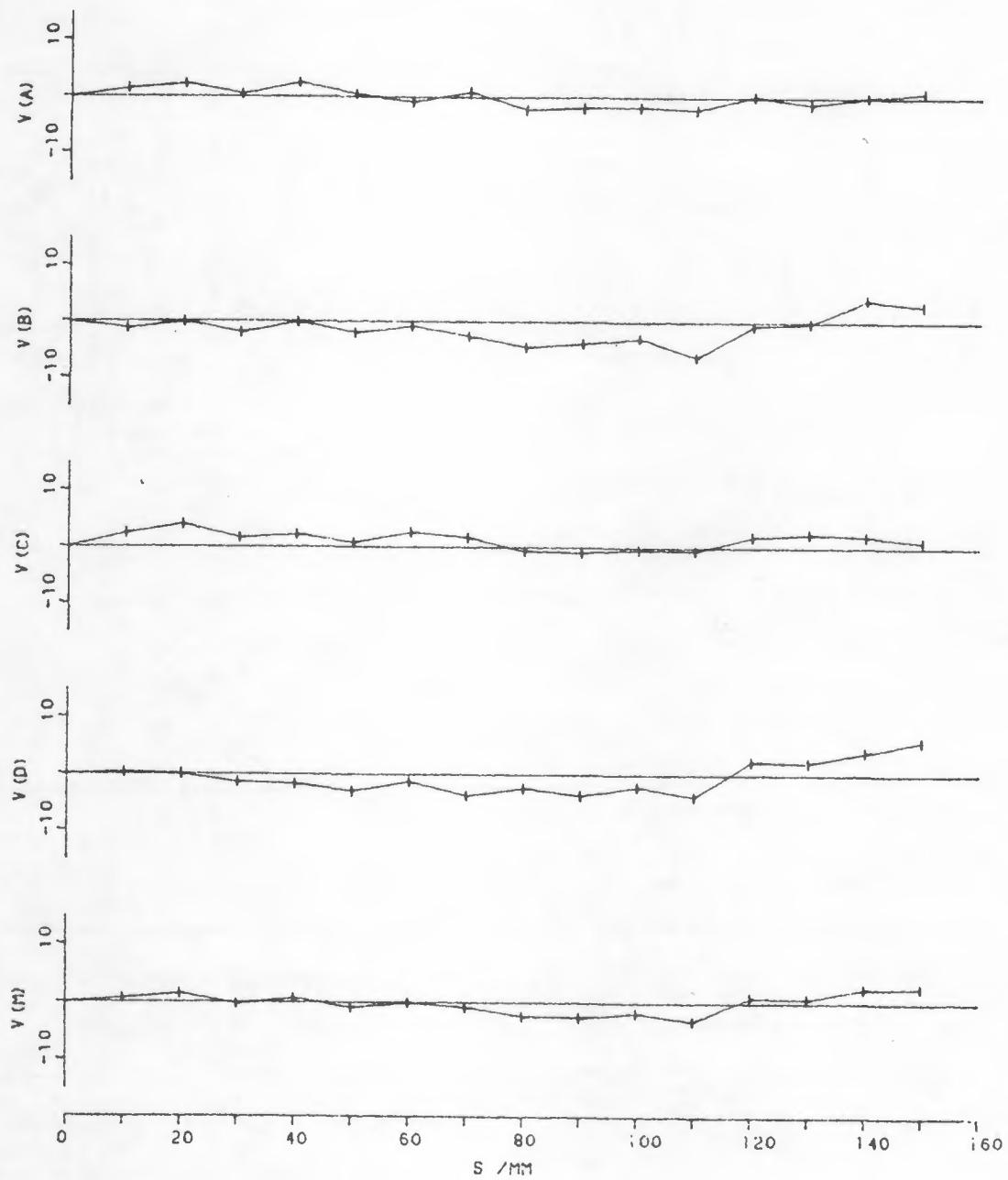
D A T E   1 1 . 0 9 . 8 0

D r . - I n g . R . Schwebel

RMK A 30/23  
TOPAR A1      5.6/305  
CFL=305.128 MM

NO. 124501  
NO. 124516

DISTORTION /0.001 MM. REFERRING TO PPS



## 2.2.2 Post-Mission Calibration

C A R L Z E I S S  
O B E R K O C H E N / W U E R T T.

### C A L I B R A T I O N C E R T I F I C A T E F O R P H O T O G R A M M E T R I C C A M E R A S

C A M E R A T Y P E : R M K A 30/23      S E R I A L N O . 1 2 4 5 0 1  
L E N S T Y P E : T O P A R A 1      S E R I A L N O . 1 2 4 5 1 6  
M A X . A P E R T U R E : F / 5 . 6      N O M . F O C A L L E N G T H : 3 0 5 M M

1) C A L I B R A T E D F O C A L L E N G T H = 3 0 5 . 1 2 3 M M

2) D I S T O R T I O N / 0 . 0 0 1 M M , R E F E R R I N G T O P . P . O F S Y M M E T R Y P P S

S / M M =	0	10	20	30	40	50	60	70	80	90	100	110	120	130	140	150
A	0	0	1	0	1	0	0	-1	-2	-2	-3	-2	1	0	1	0
B	0	1	0	1	0	0	-1	-1	-4	-2	-4	-2	1	1	1	1
C	0	1	1	2	0	1	0	0	-3	-2	-3	-2	1	1	5	3
M	0	1	3	0	1	0	1	-1	-1	-3	0	-1	0	3	1	4
A V .	0	1	1	0	0	0	0	0	-1	-2	-2	-3	-2	1	1	2

3) P . P . O F A U T O C O L L I M A T I O N A N D F I D U C I A L C E N T R E , R E F E R R I N G T O P P S

P . P . O F A U T O C O L L I M A T T O N P P A      X = . 0 5 1      Y = . 0 3 8 M M  
F I D U C I A L C E N T R E      F C      X = . 0 4 9      Y = . 0 3 3 M M

4) F I D U C I A L M A R K S , R E F E R R I N G T O P P S

X 1 = 1 1 3 . 0 4 1      Y 2 = - 1 1 2 . 9 5 5      X 3 = . 0 4 7      X 4 = . 0 5 1 M M  
Y 1 = . 0 3 1      Y 2 = . 0 3 5      Y 3 = 1 1 3 . 0 1 7      Y 4 = - 1 1 2 . 9 4 8 M M  
D I S T A N C E S      1 - 2 = 2 2 5 . 9 9 6      3 - 4 = 2 2 5 . 9 6 4 M M

5) P H O T O G R A P H I C R E S O L V I N G P O W E R , I N C Y C L E S P E R M M

A R E A W E I G H T E D A V E R A G E R E S O L U T I O N 3 9

F I E L D A N G L E / D E G = 0      7      1 4      2 4

R A D I A L L I N E S      4 5      4 4      3 4      3 6

T A N G E N T I A L L I N E S      4 5      4 4      4 2      3 7

F I L M : A V T P H O T P A N 3 0      S P E E D 2 1 D I N  
D E V E L O P E D I N U L T R A F I N 1 + 1 5

6) F I L T E R S

K L ( C L E A R ) N O .

B ( Y E L L O W ) N O .

D ( O R A N G E ) N O .

7) M A G A Z I N E P L A T E N

F K 2 4 / 1 2 0 N O .

A B T E I L U N G F U E R G E O D A E S T E U N D P H O T O G R A M M E T R I E

T . A .

*W. Lorch*

D A T E 3 0 . 0 5 . 1 9 8 4

D r . - I n g . W . L o r c h

RMK A 30/23

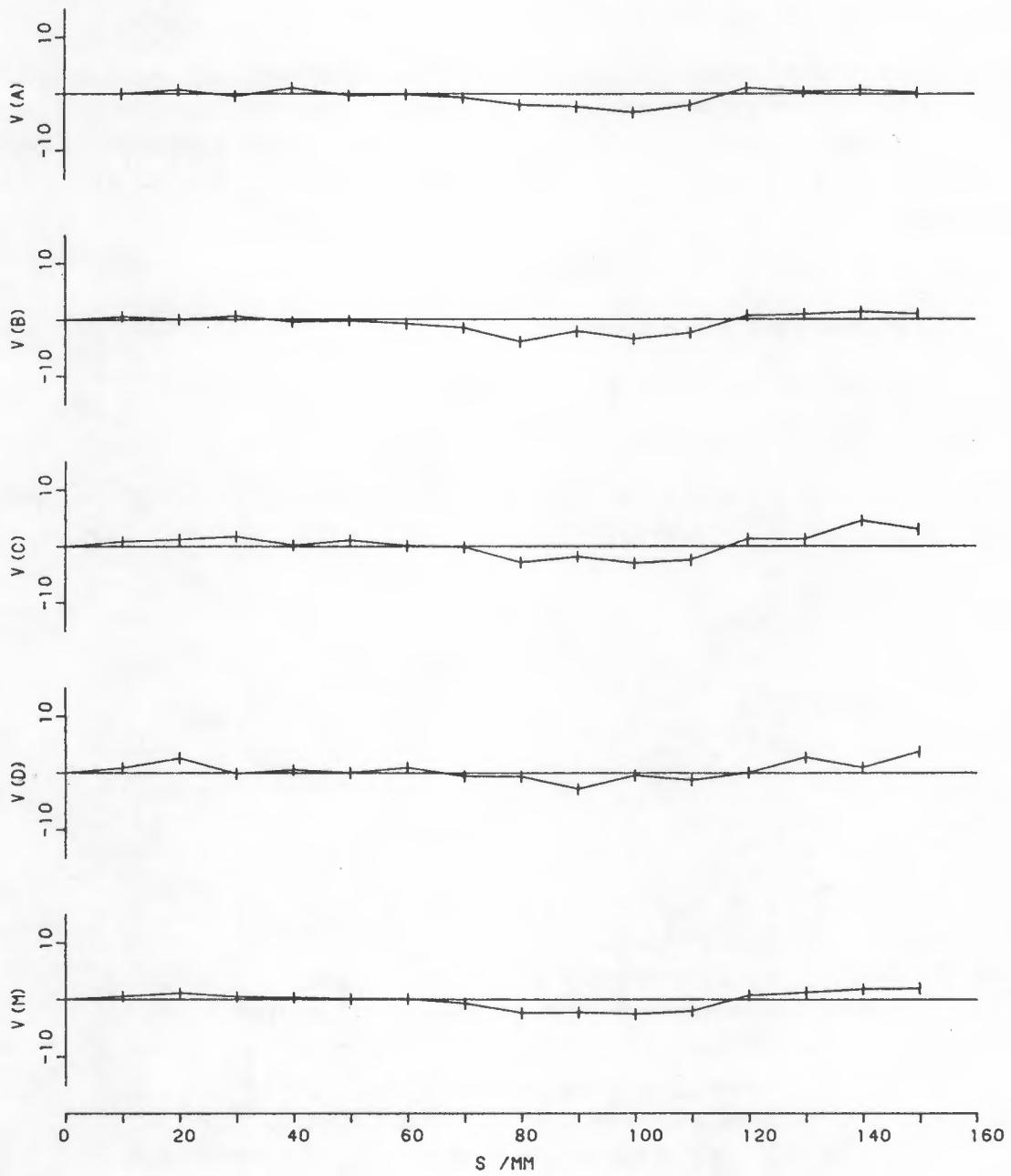
TOPAR A1 5.6/305

CFL=305.123 MM

NO. 124501

NO. 124516

DISTORTION /0.001 MM. REFERRING TO PPS



### 3. Mission Parameters

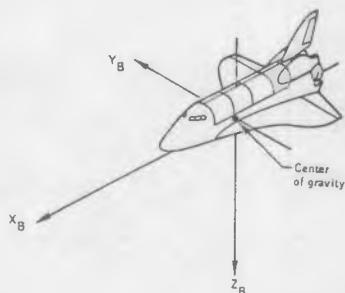
#### 3.1 Flight Attitude for Camera Operations

The orbiter body coordinate system ( $X_B, Y_B, Z_B$ ) has the positive  $X_B$ -axis through the nose, the positive  $Y_B$ -axis out the right wing, and the positive  $Z_B$ -axis through the bottom of the Orbiter. This is shown in Figure 3.1.

For operation of the camera the Orbiter flew with the open cargo bay oriented towards the earth. In this attitude the negative  $Z_B$ -axis is directed along the radius vector towards the center of the earth (NADIR). With the  $-Z_B$ -axis in earth direction four orientations of the  $X_B$ - and  $Y_B$ -axis with respect to the flight velocity vector (VV) were possible for camera operations:

1.  $+X_B$  in direction VV (Nose Forward)
2.  $-X_B$  in direction VV (Tail Forward)
3.  $+Y_B$  in direction VV (Right Wing Forward)
4.  $-Y_B$  in direction (Left Wing Forward)

Table 3.1 shows the direction of the Velocity Vector (VV) for the various camera operations.



Name:	Body Axis Coordinate System
Origin:	Center of Mass
Orientation:	<p><math>X_B</math> axis is parallel to the orbiter structural body axis; positive toward the nose.</p> <p><math>Z_B</math> axis is parallel to the orbiter plane of symmetry and is perpendicular to <math>X_B</math>, positive down with respect to the orbiter fuselage.</p> <p><math>Y_B</math> axis completes the right-handed orthogonal system.</p>

Figure 3.1: Orbiter Body Coordinate System

Table 3.1: Direction of the Shuttle Flight Velocity Vector (VV)

Ops. No.	Direction of Velocity Vector (VV)
1	$-X_B$
2	$-Y_B$
3	$-Y_B$
4	$-X_B$
5	$-Y_B$
6	$-X_B$
7	$-X_B$
8	$-X_B$
11	50° yaw of the $-X_B$ -axis
12	$-X_B$
13	$-Y_B$
14	$-Y_B$
22	$-X_B$
23	$-Y_B$
24	$-Y_B$
25	$-X_B$
29	$-X_B$
32	15° - 45° yaw of the $Y_B$ -axis
33	$+X_B$
34	$+X_B$
35	$+X_B$

The consequence of these changing flight attitudes is that the orientation of consecutive pictures differs for various flight attitudes. This is shown in Figure 3.2.

FLIGHT ATTITUDE	VELOCITY VECTOR (VV)		OPERATION NO.
	1st picture	2nd picture	
$-x_B/VV$			1, 4, 6, 7, 8, 12, 22, 25, 29
$-x_B, 50^\circ \text{ yaw}/VV$			11
$+x_B/VV$			33, 34, 35
$-y_B/VV$			2, 3, 5, 13, 14, 23, 24
$+y_B/VV$			-
$+y_B; 15^\circ \text{ yaw}/VV$			32
$+y_B; -45^\circ \text{ yaw}/VV$			32

Figure 3.2: Orientation of consecutive pictures for various Shuttle Flight Attitudes

### 3.1.1 Roll, Pitch and Yaw

To define the attitude data for roll (omega), pitch (phi) and yaw (kappa) two coordinate systems have to be introduced. The first coordinate system named  $X_0, Y_0, Z_0$  - system, is a Cartesian Coordinate System and moves along the Shuttle orbit, and the second system named Image Coordinate System ( $X_c, Y_c, Z_c$ ) , is fixed in the center of each image. Both systems are defined in Fig. 3.3 and 3.4.

The angles omega, phi, kappa for Roll, Pitch and Yaw define the orientation of the three axes of the  $X_0, Y_0, Z_0$  system with respect to the  $X_c, Y_c, Z_c$ -Image Coordinate System.

To align the  $X_0, Y_0, Z_0$ - system with the corresponding axes of the  $X_c, Y_c, Z_c$ - system three rotations in roll, pitch and yaw have to be carried out. The sequence of these rotations is as follows:

1. The  $X_0, Y_0, Z_0$ - system is rotated by the angle omega (roll) about the  $X_0$ - axis. Positive rotation is  $Y_0$  towards  $Z_0$ .
2. With the  $X_0, Y_0, Z_0$ - system in the position resulting from 1. it is then rotated by the angle phi (pitch) about the  $Y_0$ - axis. Positive rotation is  $Z_0$  towards  $X_0$ .
3. With the  $X_0, Y_0, Z_0$ - system in the position resulting from 1. and 2., it is then rotated by the angle kappa (yaw) about the  $Z_0$ - axis. Positive rotation is  $X_0$  towards  $Y_0$ .

The values for Roll, Pitch and Yaw for each picture are listed in the Data Catalogue (Chapter 6) in Columns 6 to 8.

### 3.2 Time of Exposure

The midpoint of exposure is determined by a photodiode located behind the shutter. The time for the midpoint of exposure was received from the on-board clock and was recorded with an accuracy of 1/1000 sec in Greenwich Mean Time (GMT). GMT is the local mean solar time on the Greenwich Meridian. The GMT of launch was Julian day 332 (28 Nov. 1983), 16h.

The GMT of exposure midpoint for each picture is listed in Column 3 of the Data Catalogue (Chapter 6).

Remark: Universal Time Code (UTC) = GMT

### 3.3 Geographic Coordinates

The geographic coordinates, which are listed in Column 4 and 5 of the Data Catalogue (Chapter 6) are the *geodetic coordinates* for the Shuttle's Nadir point at the midpoint of exposure. The relationship between geodetic- and geocentric latitude is given by the following equation:

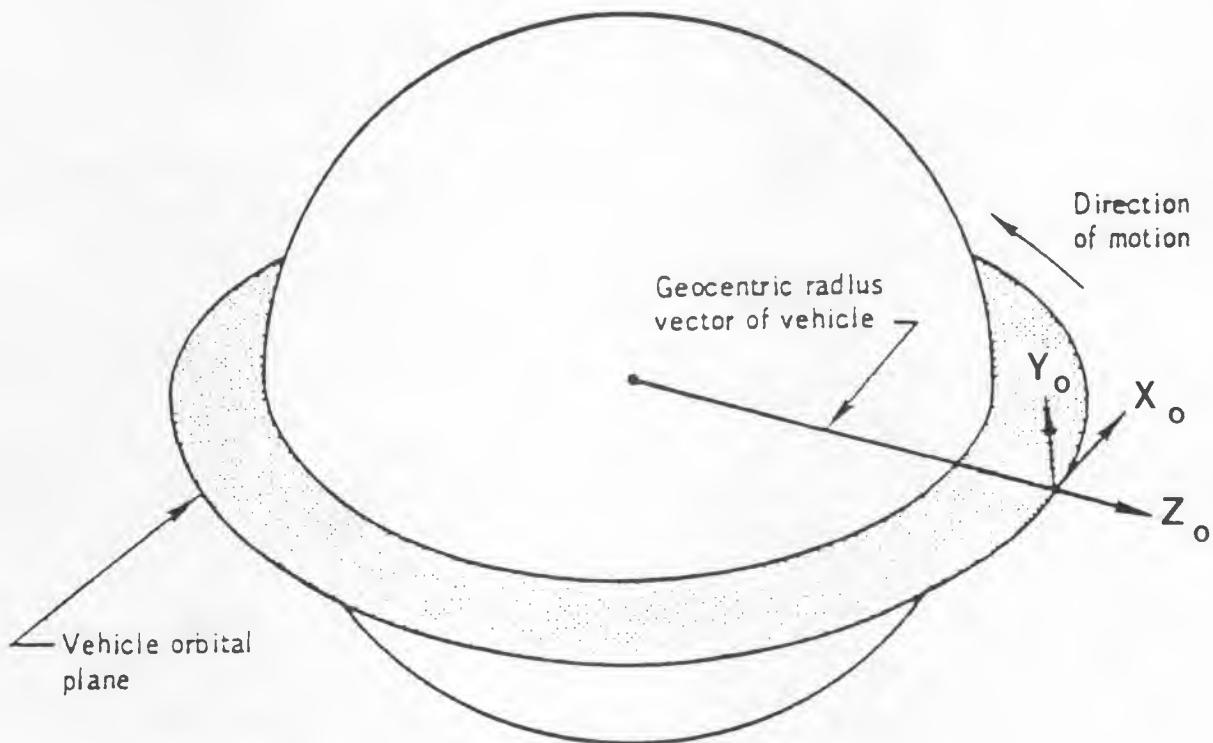
$$\tan(\varphi_D) = \frac{\tan(\varphi_C)}{(1 - \frac{1}{298.3})^2}$$

with  $\varphi_D$  : Geodetic Latitude

and  $\varphi_C$  : Geocentric Latitude

The definition of  $(\varphi_D)$  and  $(\varphi_C)$  is given in Figure 3.5.

The geographic longitude is counted positive to the East and negative to the West (see Fig. 3.5).



Name:

$X_o, Y_o, Z_o$ - Coordinate System

Origin:

Point of Interest

Orientation:

The  $X_o - Z_o$  plane is the instantaneous orbit plane at epoch. The  $Z_o$  axis lies along the geocentric radius vector to the vehicle and is positive radially outward. The  $Y_o$  axis lies along the instantaneous orbital angular momentum vector at epoch and is positive in the direction of the angular momentum vector. The  $X_o$  axis completes a right-handed system.

Characteristics:

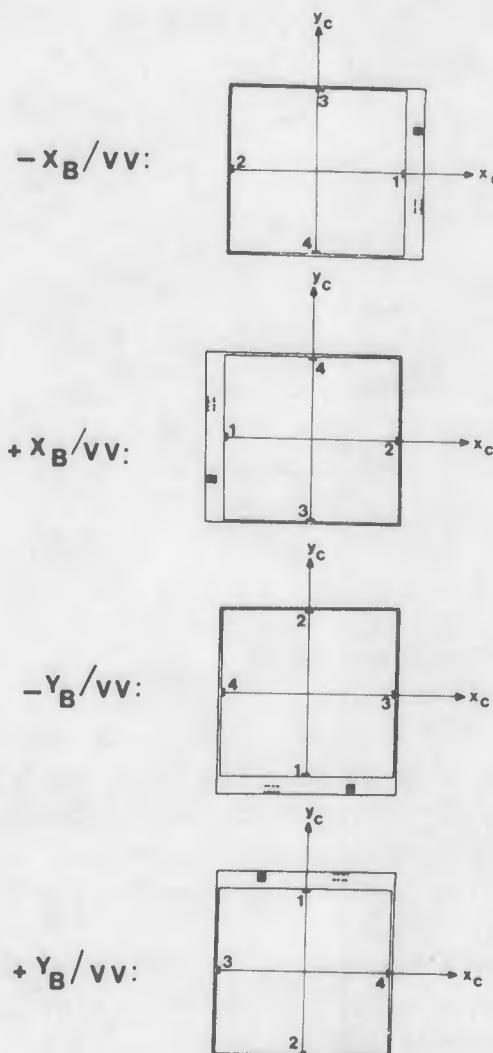
Quasi-inertial, right-handed, Cartesian coordinate system. This system is quasi-inertial in the sense that it is treated as an inertial coordinate system, but it is redefined at each point of interest.

Figure 3.3: Coordinate System  $X_o, Y_o, Z_o$

Name:  
Origin:  
Orientation:

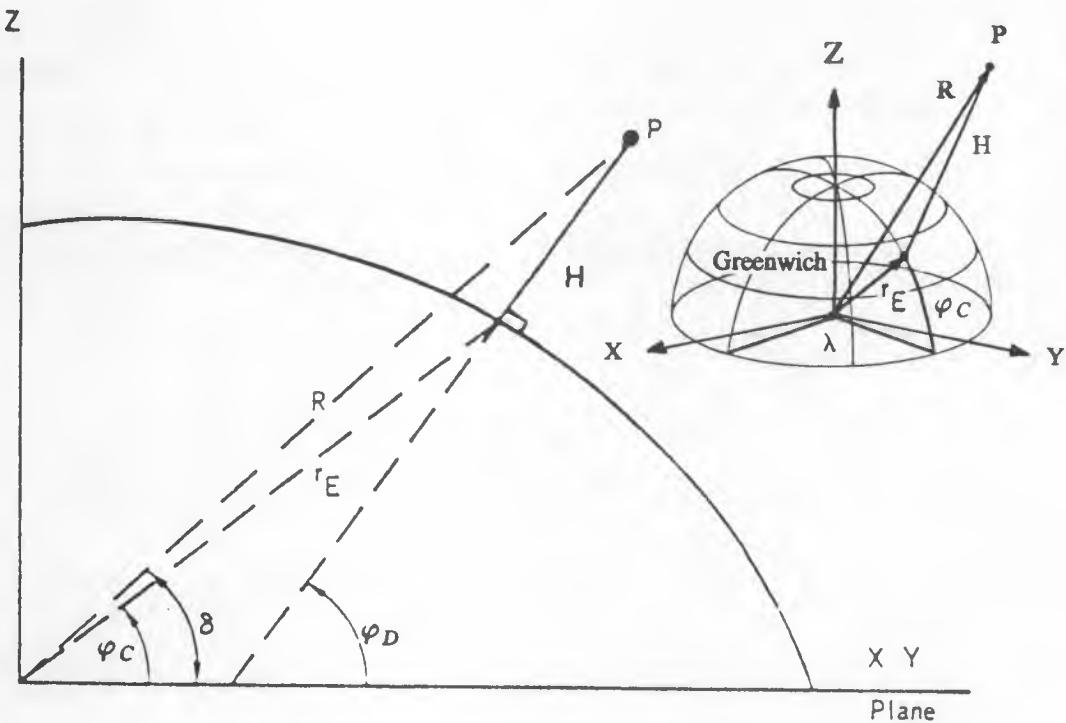
$X_c, Y_c, Z_c$  Coordinate System  
Fiducial Center in the image plane.

- The  $X_c, Y_c$ -plane is the image plane.
- The  $X_c$ -axis lies approximately along the Flight Velocity Vector (VV).
- For the four main flight attitudes for operation of the camera the direction of the  $X_c$  and  $Y_c$  axes are given by the following figures:



- The  $Z_c$ -axis is perpendicular to the image plane and completes a right-handed System.

Figure 3.4: Image Coordinate System  $X_c, Y_c, Z_c$



Name:

Geodetic Coordinate System of Point P.

Orientation:

The X-Y-plane is the earth's true of date equator. The Z-axis is directed along the earth's true of date rotational axis and is positive pointing North. The surface of the earth (geoid) is approximated by the Fischer ellipsoid modell of 1960.

Definitions:

The Point P is the instantaneous position of the Space Shuttle. The geographic longitude of Point P is the angle between the plane of the figure and the plane formed by the Greenwich meridian.

H is the altitude of Point P. Measured perpendicular from the surface of the referenced ellipsoid.

$\varphi_D$   
is the geodetic (geographical) latitude of Point P.

$\varphi_C$   
is the geocentric latitude of Point P.

$\delta$  is the angle between radius vector and equatorial plane (declination).

$\lambda$  is the longitude of Point P. Angle (+east) between plane of the figure and the plane formed by the Greenwich meridian.

Figure 3.5:

Definition of geodetic-, geocentric latitude ( $\varphi_D$ ) ( $\varphi_C$ ), geographic longitude and altitude (H)

### 3.3.1 Flight Altitude

The geodetic altitude H is the distance from the Space Shuttle (see Fig. 3.5) to the reference ellipsoid, measured along the geodetic local vertical.

The altitude for the midpoint of exposure is listed for each image in Column 9 of the Data Catalogue (Chapter 6). These values are rounded to 1/100 km. The altitude for the different camera operations varied between 235 and 255 km. Table 3.2 shows the flight altitude ranges for the different camera operations.

Table 3.2: Flight-Altitudes

Ops. No.	Altitudes (km)
1	246 - 251
2	244 - 246
3	243 - 249
4	246 - 250
5	243 - 245
6	243 - 248
7	251 - 255
8	243 - 245
11	244 - 249
13	241 - 247
14	241
22	240 - 241
23	238 - 242
24	238 - 244
25	238 - 246
29	238 - 239
32	243 - 247
33	239 - 243
34	237 - 241
35	235 - 241

### 3.4 Sun Elevation

Since the launch was originally planned for the summer months, but was delayed to 28 November 1983, the lighting conditions on the ground were very unfavorable as the sun elevation for all camera operations never exceeded  $30^\circ$ . The CIR-film was exposed at sun elevations between  $15^\circ$  and  $29^\circ$ . For the B/W film the sun elevation over Europe and North America reached only  $4^\circ$  to  $16^\circ$ .

Many B/W photographs were taken at extremely low sun angles over North America. Nevertheless, they reveal the morphology of the terrain, especially in arid zones. Table 3.3 shows the range of sun elevation angle for the different camera operations. The sun elevation angle for each picture is listed in Column 16 of the Data Catalogue (Chapter 6).

Table 3.3: Sun Elevation

Ops. No.	Sun Elevation (deg.)	Film
1	28 - 29	Colour Infrared
2	28 - 29	
3	24 - 28	
4	23 - 27	
5	27 - 28	
6	17 - 26	
7	15 - 21	
8	21 - 27	
11	22 - 26	
12	21 - 22	
13	15 - 24	
14	22 - 23	
22	15 - 16	
23	12 - 15	Black and White
24	13 - 15	
25	13 - 15	
29	13 - 14	
32	4 - 9	
33	8 - 10	
34	6 - 8	
35	4 - 8	

## 4. Film and Filters

### 4.1 Filter

A Zeiss-D Filter with a 50% transmittance cut-off wavelength at 535 nm was used with both films. The spectral transmission of the filter is shown in Figure 4.1.

The filter was positioned in front of the lens.

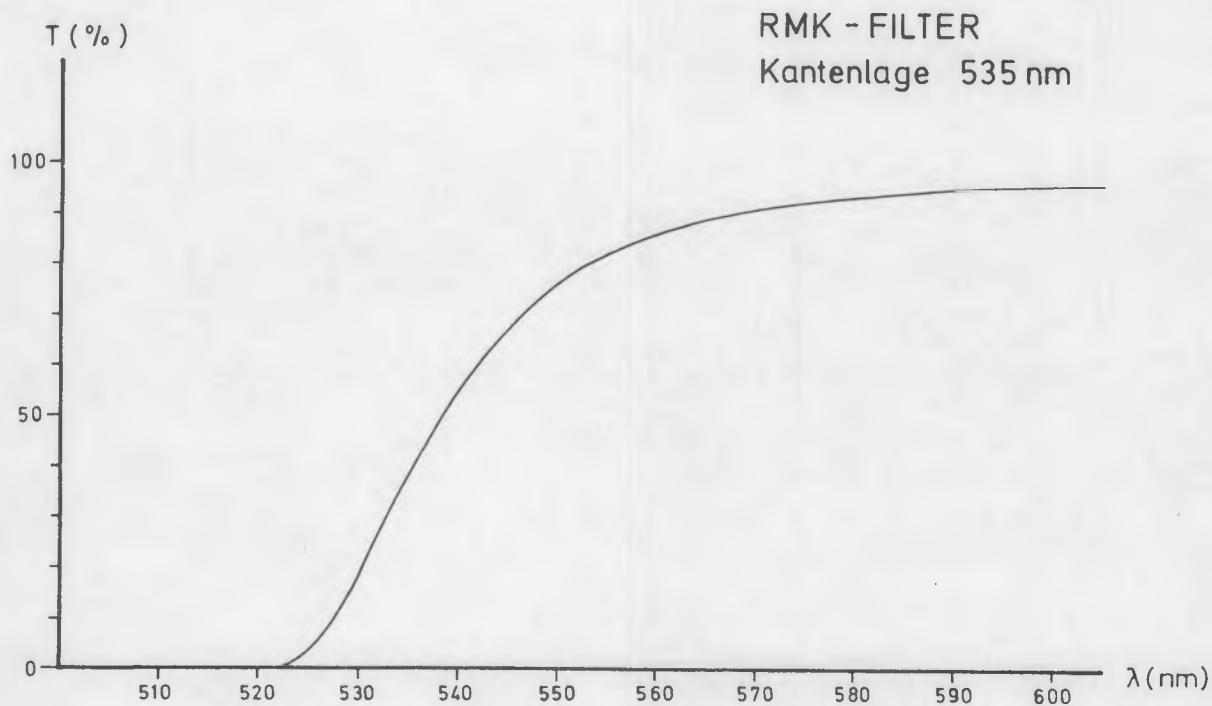


Figure 4.1: Spectral Transmission of Zeiss Filter D

## 4.2 Kodak Double - X Aerographic Film 2405

### 4.2.1 Manufacturer's Data Sheet

(From Kodak Publication M-29 'Kodak Data for Aerial Photography')

#### KODAK DOUBLE-X AEROGRAPHIC FILM 2405 (ESTAR BASE)

A panchromatic, negative camera film that has good contrast, wide exposure latitude, and extended red sensitivity for the reduction of haze effects. Its emulsion is highly hardened for high-temperature processing in continuous-processing machines. It is a medium- to high-speed film for aerial mapping, charting purposes, and general aerial photography.

<b>Base:</b>	4.0-mil ESTAR Base with a B-11 (fast-drying) backing.
<b>Sensitivity:</b>	Panchromatic, with extended red sensitivity.
<b>Safelight:</b>	None. Total darkness is required.
<b>Total Film Thickness:</b>	The nominal total thickness (unprocessed)—including emulsion (0.4 mil), base (4 mils), and fast-drying backing (nil)—is 4.4 mils.
<b>Gel/Base Ratio:</b>	0.10
<b>Weight:</b>	The weight of this film (unprocessed)—in equilibrium with 50 percent relative humidity—is 0.034 lb/ft <sup>2</sup> (0.017 g/cm <sup>2</sup> ).
<b>Aerial Film Speed:</b>	Daylight (no filter)—500 (based on development in KODAK Developer D-19 at 68°F [20°C] for 8 minutes in a sensitometric processing machine).

**Typical Camera Exposure:** A typical camera exposure for this film is approximately 1/500 second at f/11. This exposure is based on an EAFS of 400, a solar altitude of 10 degrees, an aircraft altitude of 10,000 feet, and a clear day. Processing in a KODAK VERSAMAT Film Processor, Model 11, with KODAK VERSAMAT 885 Chemicals, two developer racks, a machine speed of 10 ft/min (3 m/min), and a developer temperature of 85°F (29.5°C).

In an aerial camera equipped with an antivignetting filter, it is important to increase this typical exposure by the appropriate filter factor.

**Filter Factors:**

KODAK WRATTEN Filter	Color of Filter	Filter Factor
No. 3	Light Yellow	1.5
No. 8	Yellow	2
No. 12 (minus blue)	Medium Yellow	2.5
No. 15	Deep Yellow	2.5
No. 25	Red	4

**RMS Granularity Value:** KODAK VERSAMAT Film Processor, Model 11—26\*

**Resolving Power:** Test-Object Contrast 1000:1—125 lines/mm\*

Test-Object Contrast 1.6:1—50 lines/mm

**Reciprocity Characteristics:** Exposure times between 1/1000 and 1/10 second require no adjustment for reciprocity. For an exposure time in the range of 1/10,000 second, no lens aperture or exposure time adjustments are required—however, a 10% increase above normal in development time is recommended.

For an exposure time in the range of 1 second, a 10% decrease in development time is recommended with EITHER a lens aperture increase of +1/2 f-stop OR an exposure time increase to 1½ seconds.

\*These data are based on processing with KODAK VERSAMAT 885 Chemicals in a KODAK VERSAMAT Film Processor, Model 11, at a machine speed of 15 ft/min (4.5 m/min), with 2 developer racks and a processing temperature of 85°F (29.5°C) producing a gamma of 1.10.

## PROCESSING

This film can be processed in the KODAK VERSAMAT Film Processor, Model 11 using KODAK VERSAMAT 885 Chemicals. It also can be processed in rewind-processing machines, such as the Gordon/Morse M-10 Developing Outfit (Military Designator: B-5), using KODAK Developer D-19 or KODAK Developer DK-50.

## PROCESSING IN REWIND EQUIPMENT

The following recommendations are for processing in equipment such as the Gordon/Morse M-10 Developing Outfit (Military Designator: B-5). For complete instructions, see the "General Information" section under "Rewind Processing."

## Chemicals

KODAK Chemicals for Rewind Processing KODAK DOUBLE-X AEROGRAFIC Film 2405 (ESTAR Base)	
KODAK Developer DK-50 or KODAK HC-110 Developer	
KODAK Stop Bath SB-1a or KODAK Stop Bath SB-5	
KODAK Rapid Fixer or KODAK Fixer	
KODAK Hypo Clearing Agent* (Optional)	
KODAK PHOTO-FLO Solution	

\*Can be used to reduce washing time and conserve water.

## Development:

KODAK Developer	Length of Film (feet)	Temp	Time of Development*	Number of Cycles	Approximate Gamma	Effective Aerial Film Speed
DK-50	75	68°F (20°C)	1 pass + 15 min	8	1.1 to 1.3	250
	250		1 pass + 20 min	4		
HC-110 Dilution A	125	75°F (24°C)	1 pass + 15 min	6	0.9 to 1.0	500
	250		1 pass + 30 min	12		

\* If a pass is incomplete at the end of the recommended time, finish the pass.

**Rinse, Fix, Wash, and Dry:** See "General Information" section.

## PROCESSING IN A KODAK VERSAMAT PROCESSOR, MODEL 11

### Processing Sequence— KODAK VERSAMAT 885 Chemicals

Processing Step	Temperature	Number of Racks	Path Length	
			feet	metres
Develop	85 ± 1/2°F (29.5 ± 0.3°C)	1 or 2	4 or 8	1.2 or 2.4
Fix	85°F (29.5°C), nominal	3	12	3.6
Wash	79 to 83°F (26 to 28.5°C)	2	8	2.4
Dry	135 to 145°F (57 to 63°C)	—	8	2.4

## Sensitometric Data for KODAK VERSAMAT 885 Chemicals (85°F [29.5°C])

Machine Speed ft/min	Machine Speed m/min	No. of Dev Racks	Average Gamma	Effective Aerial Film Speed	Minimum Density
5	1.5	1	1.25	500	0.21
10	3	1	0.90	250	0.12
15	4.5	1	0.80	160	0.11
20	6	1	0.70	100	0.10
25	7.5	1	0.65	64	0.10
5	1.5	2	—	—	0.50
10	3	2	1.30	500	0.22
15	4.5	2	1.10	400	0.16
20	6	2	1.00	250	0.14
25	7.5	2	0.85	200	0.13

## MAXIMUM RECOMMENDED MACHINE SPEEDS

For Adequate Fixing	20 ft/min (6 m/min)
Washing— For Archival Records	Up to 5 ft/min (1.5 m/min)
For Adequate Drying	20 ft/min (6 m/min)

## REPLENISHMENT RATES (See Table 21, IBC)

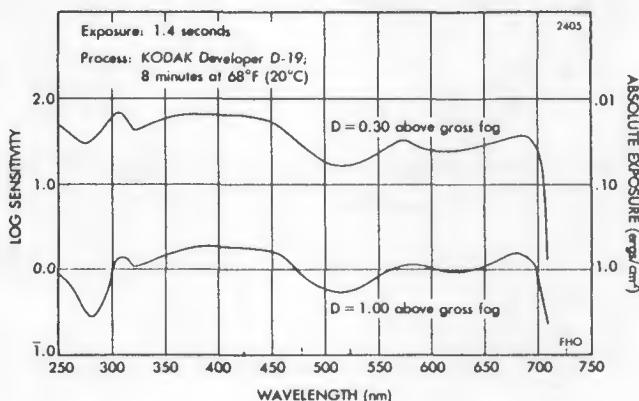
**Basic Developer Replenishment Rate:**  
0.29 mL/square inch

**Basic Fixer Replenishment Rate:** 0.45 mL/square inch

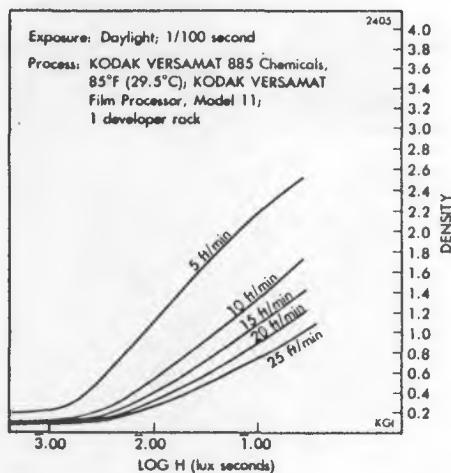
**Water Pickup in Processing:**

g/ft <sup>2</sup>	10 fpm		20 fpm	
	mg/cm <sup>2</sup>		g/ft <sup>2</sup>	mg/cm <sup>2</sup>
3.1	3.3		3.8	4.1

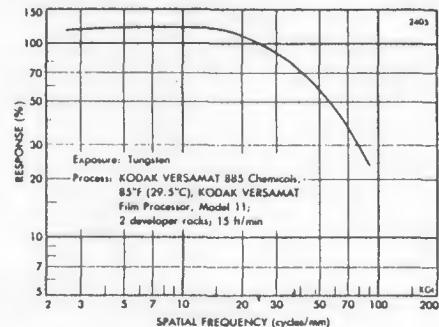
### Spectral Sensitivity Curves



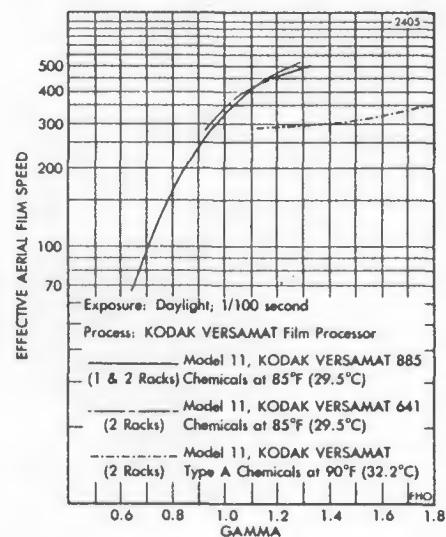
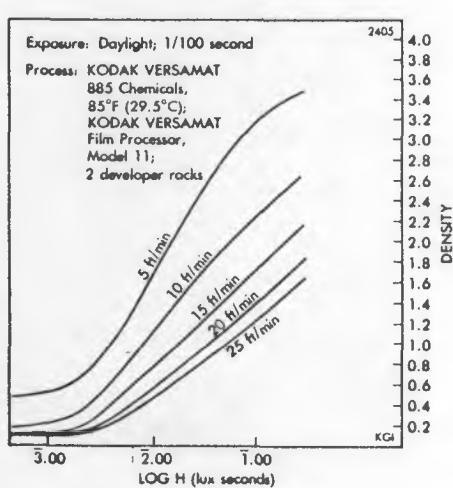
### Characteristic Curves



### Modulation-Transfer Curves



### EAFS vs. Gamma Curves



#### 4.2.2 Development and Sensitometric Control of DX 2405 - Film

Four days after completion of the mission the film was developed in the DFVLR photolab at Oberpfaffenhofen.

The film was processed in an Agfa Pakotone film processor using Kodak 885 chemicals at a temperature of 30°C and a machine speed of 28 inch/min.

For sensitometric control a 20-step grey tablet was exposed on the film with a EG & G sensitometer, Model Mark VI.

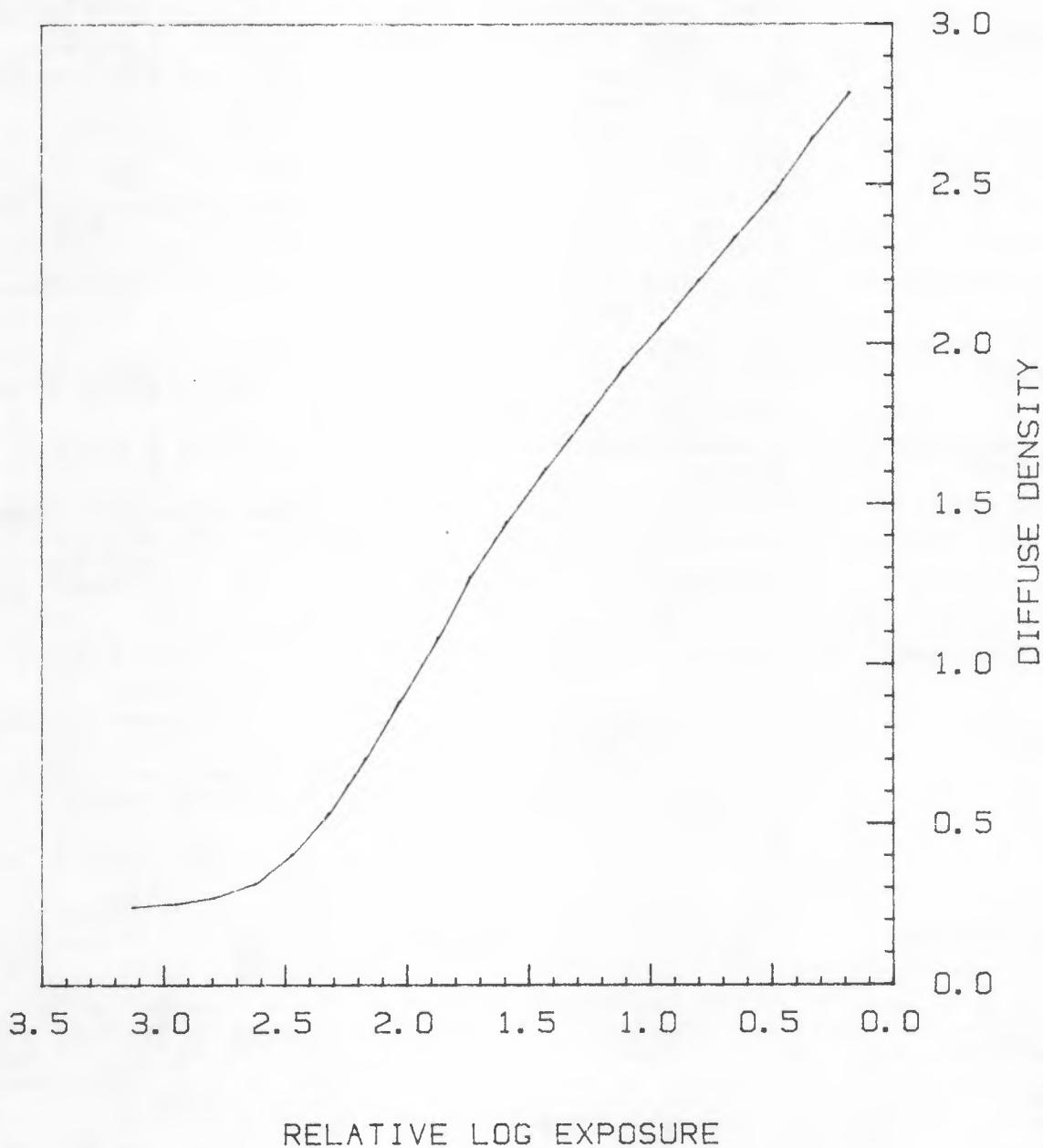
The densities of the grey step tablet and the corresponding densities measured in the film negative are given in Table 4.1.

From the values of Table 4.1 the characteristic curve for the DX-2405 film can be derived, which is shown in Fig. 4.2. From this curve the film gamma can be determined as 1.1.

Table 4.1: Sensitometric Control of DX-2405 Film

Step No.	Exposed Grey Step Tablet	Measured Density (Film Negative)
	D	D
1	0,18	2,79
2	0,33	2,65
3	0,49	2,48
4	0,65	2,34
5	0,80	2,20
6	0,95	2,07
7	1,11	1,93
8	1,26	1,78
9	1,43	1,61
10	1,59	1,45
11	1,74	1,28
12	1,87	1,09
13	2,03	0,89
14	2,17	0,77
15	2,32	0,54
16	2,47	0,41
17	2,62	0,32
18	2,79	0,27
19	2,95	0,25
20	3,13	0,24

Figure 4.2: Characteristic Curve for DX-2405-film. Gamma: 1.1



## 4.3 Kodak Aerochrome Infrared Film 2443

### 4.3.1 Manufacturer's Data Sheet

(From Kodak Publication M-29 'Kodak Data for Aerial Photography')

#### KODAK AEROCHROME INFRARED FILM 2443 (ESTAR BASE)

A "false color" reversal film for remote-sensing applications where infrared discrimination is required.

<b>Base:</b>	4.0-mil ESTAR Base with a B-11 (fast-drying) backing.
<b>Sensitivity:</b>	Sensitive to ultraviolet, visible, and infrared radiation to approximately 900 nm.
<b>Safelight:</b>	None. Total darkness is required.
<b>Total Film Thickness:</b>	The nominal total thickness (unprocessed)—including emulsion (0.8 mil), base (4 mils), and fast-drying backing (nil)—is 4.8 mils.
<b>Gel/Base Ratio:</b>	0.20
<b>Weight:</b>	The weight of this film (unprocessed)—in equilibrium with 50 percent relative humidity—is 0.035 lb/ft <sup>2</sup> (0.018 g/cm <sup>2</sup> ).
<b>Effective Aerial Film Speed:</b>	40 (based on exposure through a KODAK WRATTEN Filter No. 12, and processing in EA-5 chemicals).
<b>Typical Camera Exposure:</b>	A typical camera exposure for this film is 1/500 second at f/5.6 with a KODAK WRATTEN Filter No. 12. This exposure is based on a solar altitude of 40 degrees or above, an aircraft altitude of 10,000 feet, and a clear day. The No. 12 Filter must be used over the camera lens to absorb the blue radiation to which all three layers are sensitive.
<b>RMS Granularity Value:</b>	17
<b>Resolving Power:</b>	Test-Object Contrast 1000:1—63 lines/mm Test-Object Contrast 1:1.6—32 lines/mm
<b>Reciprocity Characteristics:</b>	Short exposure times of 1/100 to 1/1000 second require no adjustment for reciprocity. For an exposure in the order of 1/10 second, an exposure increase of 1 f-stop and the addition of a KODAK Color Compensating Filter CC20B is recommended. An exposure time of 1 second or more is not recommended.

## PROCESSING

This film can be processed in EA-5 chemicals in modern continuous processing machines such as KODAK RT Color Processor, Model 1411C, KODAK EKTACHROME RT Processor, Model 1411, and the KODAK EKTACHROME RT Processor, Model 1811.

While not a primary recommendation, this film can also be processed in rewind equipment or on stainless steel reels.

### PROCESSING IN REWIND EQUIPMENT

The following recommendations are for processing in equipment such as the Gordon/Morse M-10 Developing Outfit (Military Designator: B-5). For complete instructions, see General Information section under "Rewind Processing."

**Note on Exposure:** For best results, if 2443 Film is to be rewind-processed, expose the film in the aerial camera

with either a KODAK WRATTEN Filter No. 15 alone or a KODAK Color Compensating Filter CC10M plus the usual KODAK WRATTEN Filter No. 12 (or equivalent filters) to offset the different color balance produced by this process.

### Chemicals

KODAK Chemicals for Rewind Processing of KODAK AEROCHROME Infrared Film 2443 (ESTAR Base)
KODAK Prehardener and Replenisher, Process E-4
KODAK Neutralizer MX 875 and
KODAK Neutralizer Additive MX 870
KODAK First Developer, Process E-4
KODAK Stop Bath and Replenisher, Process E-4
KODAK Color Developer, Process E-4
KODAK Bleach and Replenisher, Process E-4
KODAK Color Film Liquid Fixer and Replenisher
KODAK Stabilizer and Replenisher, Process E-4

## Chemicals

See first cycle for the list of EA-5 chemicals.

## Processing Sequence

Processing KODAK AEROCHROME Infrared Film 2443 (ESTAR Base) in the KODAK EKTACHROME RT Processor, Model 1411 and Model 1411A (Modified with the KODAK EKTACHROME RT Speed-Up Kit, Model 1411 and Model 1411A)					
Processing Step	Tank No.	Time (seconds)	Temperature °F	Replenishment Rates	
				mL/min mL/ft <sup>2</sup>	mL/min (9½" width)
Prehardener	1 & 2	90	100 ± 1	60	260
Neutralizer	3	45	115 ± 2	60	260
First Developer	4, 5, & 6	90	120 ± ½	175	1250
First Stop	7	45	115 ± 5	200	1430
Wash	8	45	120 ± 5	2 gallons per minute	
Color Developer	9, 10, 11, & 12	90	120 ± 1	225	1760
Second Stop	13	45	120 ± 5	200	1430
Wash	14	45	120 ± 5	2 gallons per minute	
Bleach	15	45	125 ± 5	90	645
Fix	16	45	125 ± 5	90	645
Wash	17	45	120 ± 5	2 gallons per minute	
Stabilizer	18	45	Equilibrium	120	860
Dry*	—	90	135 ± 5	—	—

\*Inject EA-5 Stabilizer and Replenisher in the final wash at a rate of 50 mL/min for all film widths.

†Set air damper control knobs at 8. The temperature of the dryer may require adjustment, depending on the ambient temperature and humidity conditions in the processing area.

## Replenishment Rates

Rate in mL/ft <sup>2</sup>	Rates in mL/min for film widths of		
	70 mm	5 inches	9½ inches
60	75	135	260
175	220	400	750
225	280	510	1035
200	250	450	860
90	115	205	385

## KODAK EKTACHROME RT PROCESSOR, MODEL 1811

2443 Film can be processed in the Model 1811 in EA-5 chemicals at 9 feet per minute. At a film transport speed of 9 feet per minute, the following processing cycle provides dry-to-dry processing in 9 minutes. The film may be fed either *emulsion up* or *emulsion down* into the processor.

## Chemicals

See first cycle for the list of EA-5 chemicals.

## Processing Sequence

Processing KODAK AEROCHROME Infrared Film 2443 In the KODAK EKTACHROME RT Processor, Model 1811					
Processing Step	Tank No.	Time (seconds)	Temperature °F	Replenishment Rates	
				mL/ft <sup>2</sup>	mL/min (9½" width)
Prehardener	1 & 2	53	115 ± 1	60	430
Neutralizer	3	27	115 ± 2	60	430
First Developer	4, 5, & 6	80	120 ± ½	175	1250
First Stop	7	27	115 ± 5	200	1430
Wash	8	27	120 ± 5	2 gallons per minute	
Color Developer	9, 10, 11, & 12	107	120 ± 1	225	1760
Second Stop	13	27	120 ± 5	200	1430
Wash	14	27	120 ± 5	2 gallons per minute	
Bleach	15	27	125 ± 5	90	645
Fix	16	27	125 ± 5	90	645
Wash	17	27	120 ± 5	2 gallons per minute	
Stabilizer	18	27	Equilibrium	120	860
Dry†	—	53	145 ± 5	—	—

\*Set air damper control knobs at 8. The temperature of the dryer may require adjustment, depending on the ambient temperature and humidity conditions in the processing area.

## Replenishment Rates

Rate in mL/ft <sup>2</sup>	Rates in mL/min for film widths of		
	70 mm	5 inches	9½ inches
60	125	225	430
175	365	655	1250
200	415	750	1430
225	465	845	1760
90	190	340	645
120	250	450	860

## KODAK EKTACHROME RT PROCESSOR, MODEL 1811

### (Modified with the KODAK EKTACHROME Quick-Change Kit, Model 1811)

2443 Film can be processed in the KODAK EKTACHROME RT Processor, Model 1811 (modified with the KODAK EKTACHROME RT Quick-Change Kit, Model 1811), with EA-5 chemicals at 5.4 feet per minute. At a film transport speed of 5.4 feet per minute, the following cycle provides dry-to-dry processing in about 13 minutes. The film is fed *emulsion down* into the processor.

## Chemicals

See first cycle for a list of EA-5 chemicals.

### Processing Sequence— 9½-inch x 125-foot rolls

Processing Step	Step Time	Total Time*	Solution Temperature	
	(Minutes)		°C	°F
1. Prehardener	5	5	29.5 ± 0.6	85 ± 1
2. Neutralizer	1†	6	29.5 ± 1	85 ± 2
3. First Developer	9	15	29.5 ± 0.3	85 ± ½
4. First Stop‡	2	17‡	29.5 ± 1.1	85 ± 2
5. Wash	4	21	29.5 ± 1.1	85 ± 2
6. Color Developer	15	36	29.5 ± 1.1	85 ± 2
7. Second Stop	3	39	29.5 ± 1.1	85 ± 2
8. Wash	3	42	29.5 ± 1.1	85 ± 2
9. Bleach	10	52	29.5 ± 1.1	85 ± 2
10. Fixer	10	62	29.5 ± 1.1	85 ± 2
11. Wash	8	70	29.5 ± 1.1	85 ± 2
12. Stabilizer	1†	71	29.5§	85§
13. Dry	—	—	—	—

\*Total processing time in minutes after end of that step.

†Minimum of one pass.

‡Room lights may be turned on after completion of this step.

§Approximately.

### PROCESSING ON SPIRAL REELS (not a primary recommendation)

For best results, if 2443 Film is to be processed on spiral reels, expose the film in an aerial camera with either a KODAK WRATTEN Filter No. 15 alone, or a KODAK Color Compensating Filter CC10M plus the usual KODAK WRATTEN Filter No. 12 (or equivalent filters).

**Processing Chemicals:** KODAK Film Chemicals, Process E-4 (or equivalent), are recommended for spiral reel processing of 2443 Film.

### PROCESSING IN KODAK ROLLER-TRANSPORT PROCESSORS

The following cycles are for processing 2443 Film in EA-5 chemicals.

#### Chemicals

EA-5 Chemicals for Processing KODAK AEROCHROME Infrared Film 2443			
KODAK EA-5 Prehardener and Replenisher			
KODAK EA-5 Neutralizer and Replenisher			
KODAK EA-5 First Developer			
KODAK EA-5 First Developer Replenisher			
KODAK EA-5 First and Second Stop Bath and Replenisher			
KODAK EA-5 Color Developer			
KODAK EA-5 Color Developer Replenisher			
KODAK EA-5 Bleach and Replenisher			
KODAK EA-5 Fixer and Replenisher			
KODAK EA-5 Stabilizer and Replenisher			

### KODAK RT COLOR PROCESSOR, Model 1411C (Modified with the KODAK RT COLOR QUICK-CHANGE KIT, Model 1411C)

KODAK AEROCHROME Infrared Film 2443 can be processed in the KODAK RT Color Processor, Model 1411C (modified with the KODAK RT Color Quick-Change Kit, Model 1411C) in EA-5 chemicals at 2.7 feet per minute. At a film transport speed of 2.7 feet per minute, the following

processing cycle provides dry-to-dry processing in 21 minutes. This film is fed emulsion down into the processor.

### Processing Sequence

Processing KODAK AEROCHROME Infrared Film 2443 in the KODAK RT Color Processor, Model 1411C (Modified with the KODAK RT Color Quick-Change Kit, Model 1411C) with KODAK EA-5 Chemicals					
Processing Step	Tank No.	Time (seconds)	Temperature °F	Replenishment Rates	
				mL/ft <sup>2</sup>	mL/min (9½" width)
Prehardener	1	90	100 ± 1	60	130
Neutralizer	2	90	Equilibrium	150 mL/min for all widths	
Skip Tanks 3 & 4	—	—	—	—	—
First Developer	5	90	120 ± ½	175	375
First Stop Bath	6	90	105 ± 2	200	430
Wash	7	90	95 ± 5	2 gallons per minute	
Color Developer	8	90	120 ± 1	225	480
Second Stop Bath	9	90	110 ± 2	200	430
Wash	10	90	95 ± 5	2 gallons per minute	
Bleach	11	90	110 ± 2	90	190
Fixer	12	90	110 ± 2	90	190
Wash	13	90	95 ± 5	1 gallon per minute	
Final Wash*	14	90	95 ± 5	1 gallon per minute	
Dry†	—	180	125 ± 5	—	—

\*Add EA-5 Stabilizer and Replenisher to the final wash at a rate of 50 mL/min for all film widths.

†Set air damper control knobs at 8. The temperature of the dryer may require adjustment, depending on the ambient temperature and humidity conditions in the processing area.

### Replenishment Rates

Rate in mL/ft <sup>2</sup>	Rates in mL/min for film widths of		
	70 mm	5 inches	9½ inches
60	40	70	130
175	110	200	375
200	125	225	430
225	140	255	480
90	55	100	190

### KODAK EKTACHROME RT PROCESSOR, Model 1411 and Model 1411A (Modified with the KODAK EKTACHROME RT SPEED-UP KIT, Model 1411 and Model 1411A)

KODAK AEROCHROME Infrared Film 2443 (ESTAR Base) can be processed in the KODAK EKTACHROME RT Processor, Model 1411 or Model 1411A (modified with the KODAK EKTACHROME RT Speed-Up Kit, Model 1411 and Model 1411A), in EA-5 chemicals at 5.4 feet per minute. At a film transport speed of 5.4 feet per minute, the following processing cycle provides dry-to-dry processing in 12 minutes. The film is fed emulsion up or emulsion down into the processor.

## Processing Sequence

Processing KODAK AEROCHROME Infrared Film 2443 in the KODAK EKTACHROME RT Processor, Model 1811 (Modified with the KODAK EKTACHROME RT Quick-Change Kit, Model 1811)					
Processing Step	Tank No.	Time (seconds)	Temperature °F	Replenishment Rates	
				mL/ft <sup>2</sup>	mL/min (9½" width)
Prehardener	1 & 2	90	115 ± 1	60	260
Neutralizer	3	45	Equilibrium	60	260
Skip Tanks 4, 5, & 6	—	11	—	—	—
First Developer	7 & 8	90	120 ± ½	175	750
First Stop Bath	9	45	115 ± 5	200	860
Wash	10	45	120 ± 5	2 gallons per minute	
Color Developer	11 & 12	90	120 ± 1	225	1035
Second Stop Bath	13	45	120 ± 5	200	860
Wash	14	45	120 ± 5	2 gallons per minute	
Bleach	15	45	125 ± 5	90	385
Fixer	16	45	125 ± 5	90	385
Wash	17	45	120 ± 5	1 gallon per minute	
Final Wash*	18	45	120 ± 5	1 gallon per minute	
Dry†	—	90	135 ± 5	—	—

\*Inject EA-5 Stabilizer and Replenisher into the final wash at a rate of 50 mL/min for all film sizes.

†Set air damper control knobs at 8. The temperature of the dryer may require adjustment, depending on the ambient temperature and humidity conditions in the processing area.

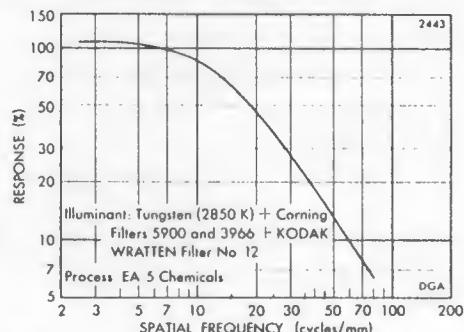
## Replenishment Rates

Rate In mL/ft <sup>2</sup>	Rates in mL/min for film widths of		
	70 mm	5 Inches	9½ Inches
60	75	135	260
175	220	400	750
200	250	450	860
225	280	505	1035
90	115	205	385

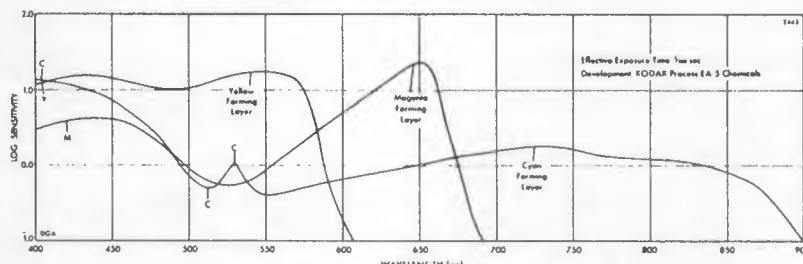
## PROCESS CONTROL STRIPS

KODAK Control Strips, Process EA-5, are available and are recommended for monitoring the processing of 2443 Film and several other color aerial films in Kodak roller-transport processors using EA-5 chemicals.

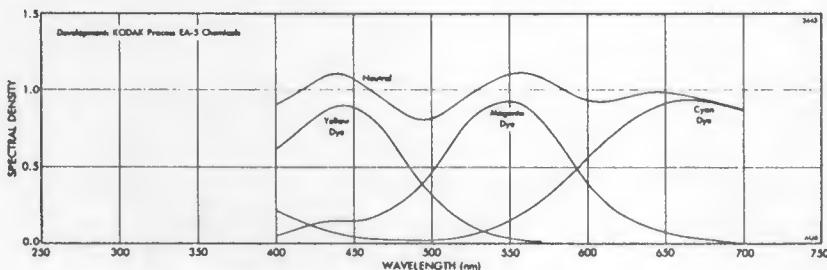
### Modulation-Transfer Curve



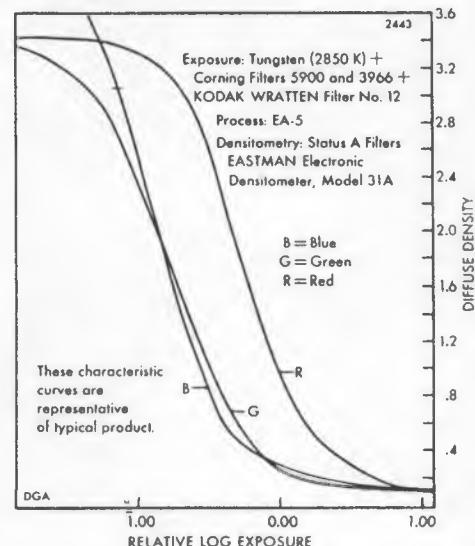
### Spectral Sensitivity Curves



### Spectral Dye Density Curves



### Characteristic Curves



#### 4.3.2 Development + Sensitometric Control of the Colour IR-Film 2443 (CIR 2443)

The Colour Infrared Film 2443 had to be loaded into the magazine 5 weeks before the mission, this being the latest point of access to the Module in the Shuttle pre-flight preparation. During these five weeks the film remained in the magazine under controlled laboratory conditions. The film was developed 7 days after completion of the mission at IGN-France photolabs in Creil using a Kodak Versamat 1811.

For sensitometric control a 21 step grey tablet was exposed with a Kodak type VI sensitometer through a Wratten 15 filter on the film before development. The Wratten 15 filter's spectral characteristic are very similar to the Zeiss D-filter, which was used in Spacelab with the CIR-2443 film. For comparison the same sensitometric control was made with a fresh film of the same emulsion, which was kept until development under controlled conditions in a refrigerator at approx.  $-20^{\circ}\text{C}$ .

The processing conditions are listed in Table 4.2. The densities of the grey step tablet and the corresponding densities measured in the film positive are given in Table 4.3. From the values of Table 4.3 the characteristic curves of the developed films can be derived and are shown in Figure. 4.3 to Figure 4.5.

From these curves it can be found:

- The IR-balance <sup>oo</sup>) of the fresh film (Fig. 4.4) was 5; this means that the film was IR-enhanced, which is very favourable for high altitude applications.
- The IR-balance <sup>oo</sup>) of the film used in Spacelab (Fig. 4.3) increased during the five weeks pre-mission storage to 20, which is still IR-enhanced.
- The characteristic curve of the blue and green layers for the Spacelab film shifted at  $D=1.0$  on the log E-axis by 0.08 to the right, which means they became approx. 1/4 f-stop slower.
- The characteristic curve of the red layer for the Spacelab film shifted at  $D=1.0$  on the log E-axis by 0.23 to the right, which means it became approx. 3/4 f-stop slower.
- The characteristic curve of the neutral density for the Spacelab film shifted at  $D=1.0$  on the log E-axis by 0,13 to the right, which means the film became approx. 1/2 f-stop slower.

---

<sup>1)</sup> Definition of Infrared-Balance:

Two points on the characteristic curves at  $D=1.0$  are established: One being at the mid-point between the green and the blue curve and the second on the IR-curve. The distance between these two points, measured parallel to the log E-axis and multiplied by 100 is a measure for the IR-balance.

(J. F. Fleming, Standardization Techniques for Aerial Colour Infrared Film, The Survey and Mapping Branch, Department of Energy Mines and Resources, Ottawa, Canada)

**Table 4.2: Processing Conditions for Colour IR-Film 2443**

	Machine	Chemicals	Transport Speed	Temperature
Kodak Recommendations	1811	EA-5	5,4 ft/min	120°F
Spacelab Film	1811	EA-5	5,6 ft/min	120°F
Fresh Film	1811	EA-5	5,4 ft/min	120°F

**Table 4.3: Sensitometric Control of the Colour IR-Film 2443  
(Measured with Macbeth TD-504)**

Step No.	D	Spacelab-Film				Fresh Film			
		Neutral	Blue	Green	Red	Neutral	Blue	Green	Red
1	0,18	0,095	0,10	0,12	0,08	0,080	0,09	0,10	0,05
2	0,34	0,105	0,11	0,13	0,09	0,085	0,09	0,10	0,06
3	0,48	0,115	0,12	0,14	0,11	0,090	0,09	0,10	0,07
4	0,62	0,130	0,14	0,14	0,14	0,100	0,11	0,11	0,08
5	0,76	0,150	0,15	0,15	0,18	0,110	0,12	0,12	0,10
6	0,91	0,180	0,18	0,16	0,23	0,120	0,14	0,12	0,12
7	1,04	0,220	0,22	0,18	0,32	0,150	0,16	0,14	0,15
8	1,16	0,285	0,26	0,21	0,47	0,180	0,21	0,15	0,21
9	1,32	0,390	0,32	0,27	0,64	0,240	0,27	0,19	0,31
10	1,46	0,610	0,47	0,48	0,94	0,395	0,39	0,34	0,47
11	1,59	0,930	0,74	0,77	1,37	0,650	0,63	0,61	0,74
12	1,72	1,290	1,11	1,08	1,84	0,960	0,94	0,88	1,03
13	1,86	1,680	1,57	1,42	2,34	1,340	1,37	1,09	1,46
14	2,00	2,045	2,03	1,74	2,68	1,775	1,82	1,56	1,98
15	2,15	2,350	2,49	2,03	2,90	2,170	2,34	1,90	2,49
16	2,29	2,590	2,84	2,28	3,04	2,470	2,68	2,16	2,82
17	2,44	2,760	3,05	2,44	3,11	2,660	2,96	2,35	3,02
18	2,58	2,860	3,23	2,58	3,14	2,760	3,11	2,45	3,08
19	2,72	2,920	3,30	2,64	3,15	2,860	3,24	2,52	3,11
20	2,87	2,975	3,30	2,69	3,15	2,905	3,30	2,59	3,15
21	3,09	2,950	3,30	2,69	3,15	2,915	3,30	2,59	3,15

Figure 4.3: Characteristic Curve for Colour Infrared  
2443 Film used in Spacelab

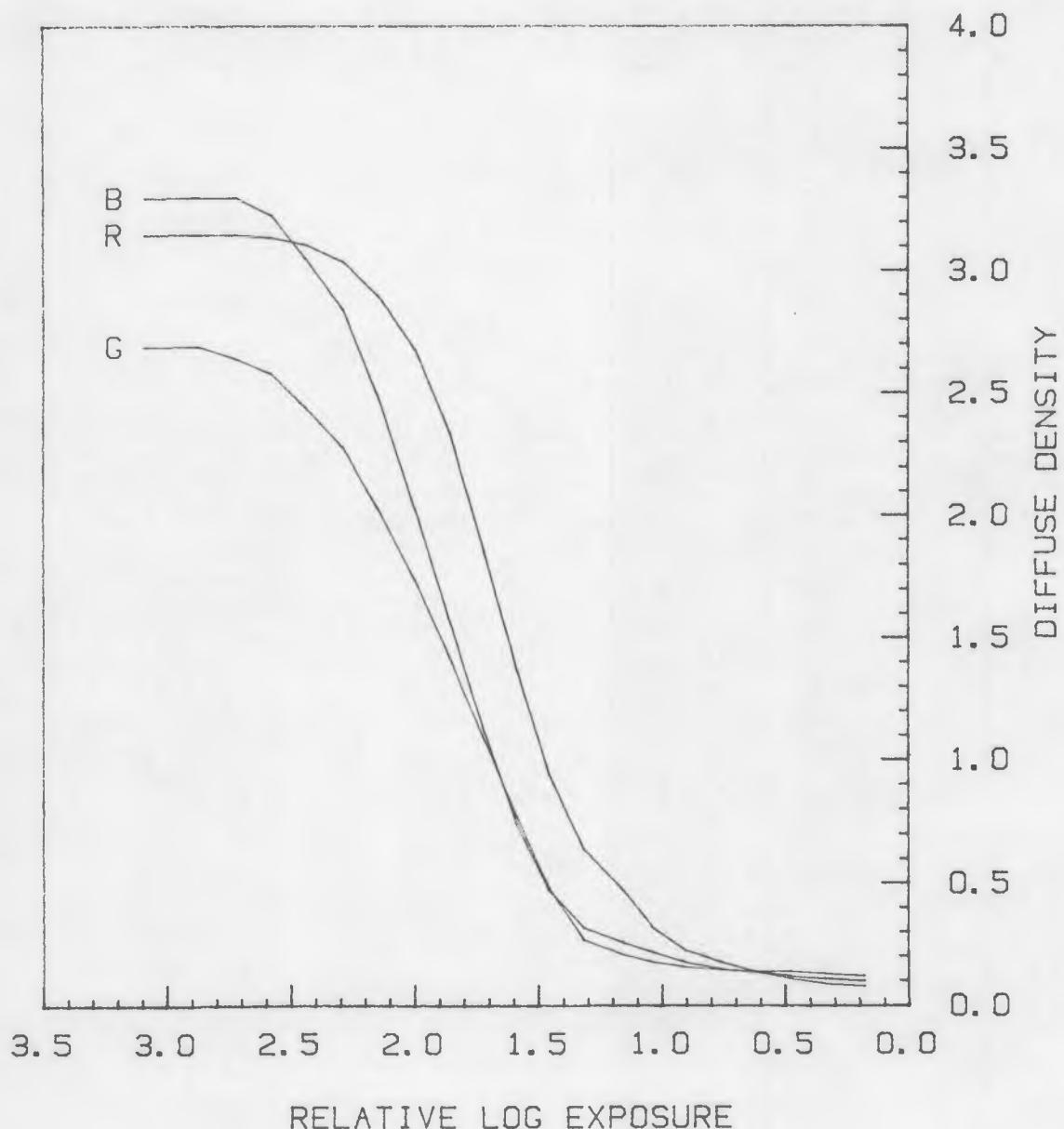


Figure 4.4: Characteristic Curve for the 'fresh' Colour Infrared Film 2443

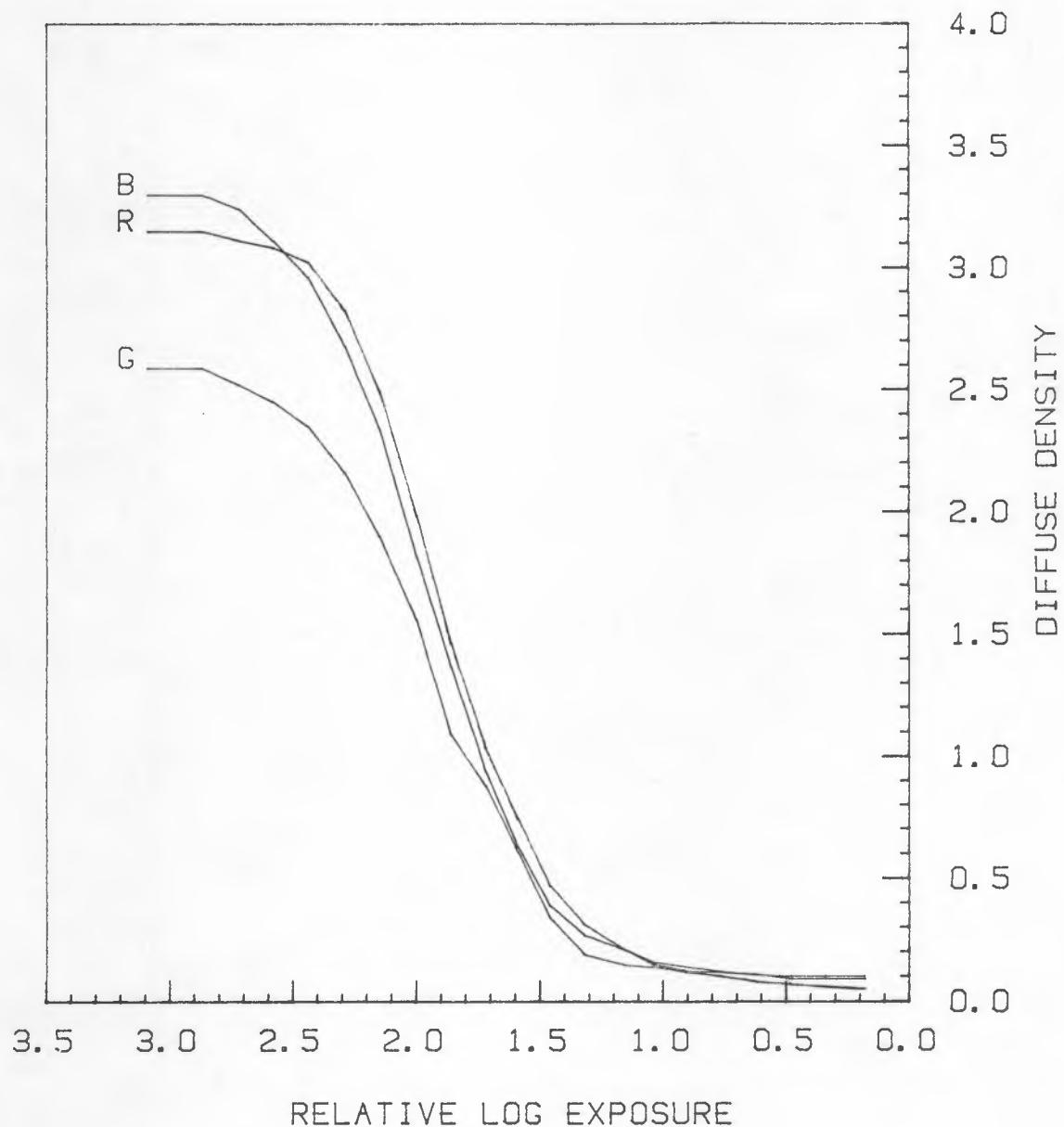
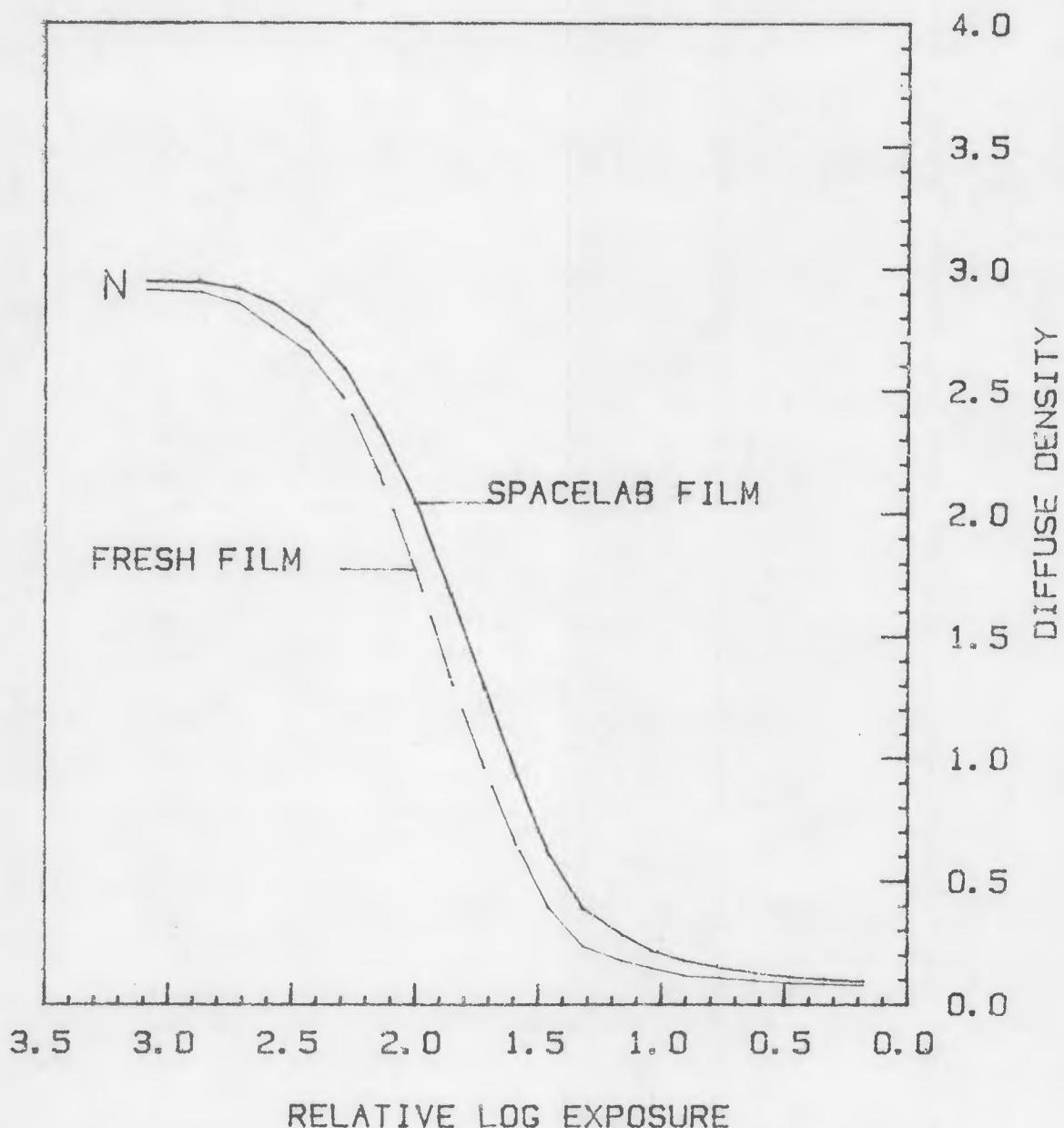


Figure 4.5: Characteristic Curve for the Neutral Density  
of the Colour Infrared Film 2443 for the film used in Spacelab  
and for the 'fresh' film



## 5. Data Products

### 5.1 Copy Material

The products delivered to the users are first generation copies of the original images, Table 5.1 shows the film or paper material that is used for the copies.

Table 5.1: Copy Material for first Generation Copies

Colour Film	Ilford-Cibachrome II CTD.F7
Colour Paper	Ilford-Cibachrome II CPS.1K
B/W-Film	Agfa-Scannerfilm S 230 P
B/W-Paper	Ilford-Ilfospeed-Multigrade II

### 5.2 Auxiliary Data

Auxiliary data were exposed on every original image immediately after film transport. These data consist of a six step grey tablet and two lines of numerical data as shown in Fig. 5.1 for a positive image.

#### Grey Step Tablet

A six step grey tablet was built into the focal plane frame of the camera and was exposed on the original image via a small lamp. The bulb was *not* a standard 'white' light source.

The densities which were measured with a Macbeth-densitometer for the grey step tablet in the original films are listed in Table 5.2. These density values can be used to determine the D/log E-curves and the film gamma for the copy film. To obtain these curves it is necessary only to plot the densities of the copy film as function of the densities of the original film.

Table 5.2: Densities of the Grey Step Tablet in the Original Film

Grey Step No.	CIR 2443				DX - 2405
	N	B	G	R	N
1	0,57	1,62	0,54	0,75	2,29
2	1,42	2,86	1,32	1,72	1,81
3	2,30	3,46	2,07	2,78	1,38
4	2,76	3,71	2,62	3,13	0,90
5	2,86	3,75	2,67	3,19	0,59
6	2,89	3,75	2,72	3,21	0,49

### Numerical Data

The first line of the numerical data gives the time in GMT for the midpoint of exposure (Fig. 5.1):

hh : Hour  
min : Minutes  
sec : Seconds  
1/10 : 1/10 Seconds  
1/100 : 1/100 Seconds  
1/1000 : 1/1000 Seconds

The second line comprises from left to right (Fig. 5.1).

- d: Day of year coded by a one digit number from 0 to 9. Table 5.3 shows to which calendar date the code number corresponds.
- Image No.: A four digit number between 0001 and 1054; it is a sequential number unique for each image.
- t: Exposure time coded by a two digit number from 01 to 31. Table 5.4 shows how these code numbers correspond to the exposure time.
- f-stop: Aperture number coded as two digit numbers from 01 to 31. Table 5.4 shows how these code numbers correspond to the f-stop.

Table 5.3: Coding of the Calendar Date

Code No.	Julian Day	Calendar Date
2	332	28 November 1983
3	333	29 November 1983
4	334	30 November 1983
5	335	1 December 1983
6	336	2 December 1983
7	337	3 December 1983
8	338	4 December 1983
9	339	5 December 1983
0	340	6 December 1983
1	341	7 December 1983

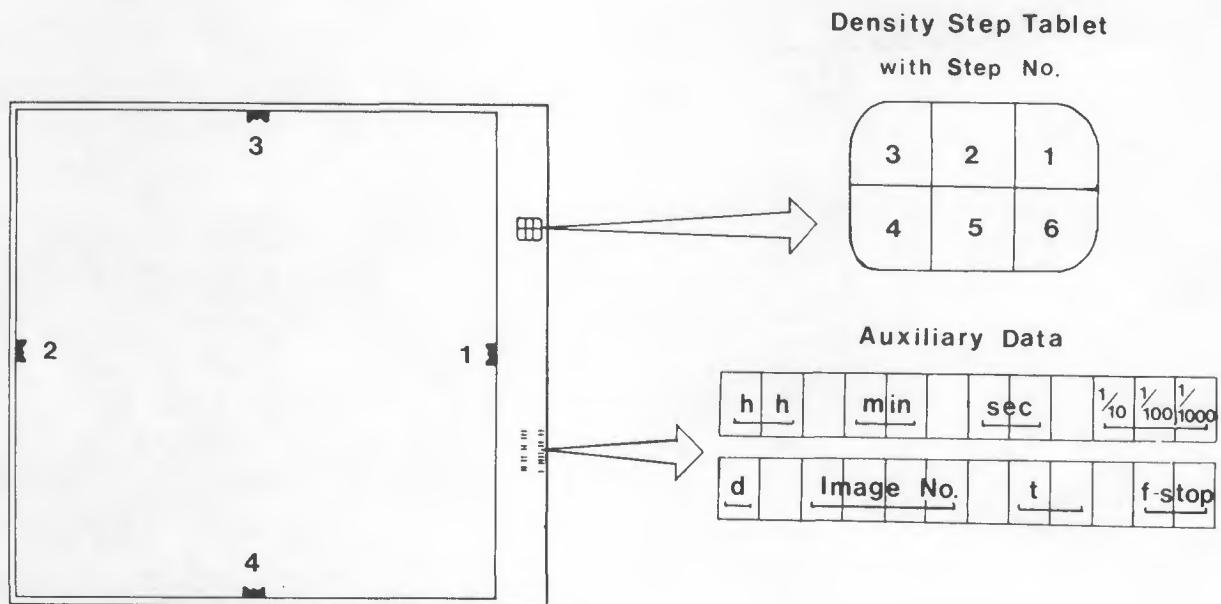


Figure 5.1: Auxiliary Data Strip

Project Identification                                  Agency Identification

STS 9/SPACELAB 1 METRIC CAMERA - Processed by DFVLR for ESA/EARTHNET  
02 Dec 83 06-25-41 GMT SUNEL 30 AZ 126 Approx. Center N 13.9 E 032.3 01-0102-03

Date & Time of Acquisition	Sun Elevation & Azimuth	Image Center Coordinates	Mission-, Image-, Operation- Number
----------------------------	-------------------------	--------------------------	-------------------------------------

Figure 5.2: Example of Annotation Strip

Table 5.4: Coding of Exposure Time (t) and f-stop

Code	t(sec)	f-stop
1	1/250	5,6
2	1/275	5,7
3	1/300	5,8
4	1/325	5,9
5	1/350	6,0
6	1/375	6,1
7	1/400	6,1
8	1/425	6,2
9	1/450	6,3
10	1/475	6,4
11	1/500	6,5
12	1/525	6,6
13	1/550	6,7
14	1/575	6,8
15	1/600	7,0
16	1/625	7,1
17	1/650	7,3
18	1/675	7,4
19	1/700	7,6
20	1/725	7,8
21	1/750	8,0
22	1/775	8,2
23	1/800	8,4
24	1/825	8,6
25	1/850	8,9
26	1/875	9,2
27	1/900	9,5
28	1/925	9,9
29	1/950	10,3
30	1/975	10,7
31	1/1000	11,3

### 5.3 Annotation

A two line annotation strip inserted either over or under the auxilliary data strip is exposed on every copy. An example of the annotation strip is shown in Fig. 5.2.

The data for the Sun Elevation and the Image Center Coordinates may differ from those given in the Data Catalogue (Chapter 6) by some tenth of a degree. The reason for this difference is that the data for the annotation strip were generated shortly after the mission with an approximated orbit programm, whereas the data in Chapter 6 (Data Catalogue) were generated from actually measured orbit data. The data in the catalogue are therefore more accurate.

#### 5.4 Microfiche Catalogue

A set of 8 microfiches has been produced which consists of a map with all ground tracks for the camera operations and all Metric Camera images from the first Spacelab Flight. Four of the microfiches are in colour for the images on CIR-2443 film and four are in black and white for the images on DX-2405 film. The Microfiche Catalogue is only available through ESA/Earthnet Frascati (see 5.5).

#### 5.5 Data Distribution

Data requests should be directed to:

DFVLR-DA  
D-8031 Wessling  
W-Germany  
Phone: (49) 8153-28(0)-673  
T/X 525419 dvlop'd

or

ESA/Earthnet, Programme Office, User Services  
ESRIN C.P. 64  
Via Galileo Gallilei  
00044 Frascati, Italy  
Phone: (39) 6-94011  
T/X: 610637 ESRIN I

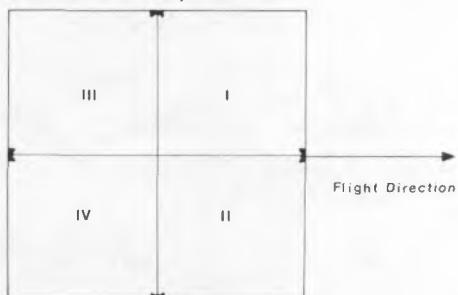


Figure 6.1: Four Quadrants of an image for Cloud Cover Estimation

## 6. Data Catalogue

### 6.1 Data List

The data catalogued in the following list identify those MC photographs which were obtained during the first Spacelab Mission on Space-Shuttle Flight No. 9. It also includes specific parameters associated with each image frame such as image No., date, Greenwich Mean Time (GMT), geographic coordinates, attitude and altitude, overlap, cloud coverage, image quality, film type, MC settings, sun elevation and the photographed country. These parameters are listed for every MC image in 17 columns which are explained below.

The data for geographical coordinates, attitude and altitude are based on the PATH (Post Flight Attitude and Trajectory History) data recorded during the mission in 1 sec. intervals. The data listed in this catalogue were derived from the PATH data by linear interpolation over the two nearest points with respect to the GMT of the exposure event.

**Column 1** - Image ID: Mission-Image Number-Operation

**Column 2** - Calendar Date

**Column 3** - Greenwich Mean Time (=UTC= Universal Time Code) for the midpoint of exposure.

**Column 4** - Geographic latitude (geodetic) of Nadir Point (see 3.3).

**Column 5** - Geographic longitude of Nadir Point (see 3.3).

**Column 6** - Roll-angle (see 3.1.1).

**Column 7** - Pitch-angle (see 3.1.1).

**Column 8** - Yaw-angle (see 3.1.1).

**Column 9** - Geodetic Altitude (see 3.3.1).

**Column 10** - Overlap:

The Metric Camera was able to acquire photography with 60% or 80% overlap in flight direction. The percent overlap indicates how much of each frame was re-photographed by the following frame.

**Column 11** - Cloud Cover:

An estimate of the percentage of cloud cover was made for each frame of photography. For this estimation the image was devided into four quadrants as shown in Fig. 6.1. For each quadrant the cloud cover is given by a digital number between 0 and 9. This number has the following meaning:

0.....0-10 % Cloud Cover

1.....11-20 % Cloud Cover

9.....91-100% Cloud Cover

Each digit represents one quadrant in the sequence of the quadrants (see Fig. 6.1).

**Column 12 - Image Quality:**

The image quality value (digital number from 0 to 9 or X) was obtained by visual inspection of first generation copies and is intended to give the user a rough indication of the overall interpretability of the image content. For scoring, the following criteria were used:

X...Quality not assessed, entire sea scene or ground not visible due to clouds.

0...Exceptional in contrast and definition (relative to all MC-1 images).

1...Exceptional, but not in all parts of the image.

3...Excellent in either contrast or definition and very good in the remaining criterium.

5...Part of the image with quality equal to 3, and part equal to 7, or overall missing contrast and details e.g. in linear features or also poor details due to the nature of the object (sand desert).

7...Ground visibility reduced to a large extend by cirrus clouds, haze or mist. Still some contrast to distinguish shore lines, rivers and/or different types of vegetation or soil.

9...Very poor contrast, image appears just grey.

**Column 13 - Film type:**

CIR...Kodak Aerochrome Infrared Film 2443

B/W...Kodak Double-X Aerographic film 2405

**Column 14 - Aperture Number or f-stop.**

**Column 15 - Reciprocal of exposure time.**

**Column 16 - Sun Elevation Angle**

**Column 17 - Name of Country or Region**



METRIC-CAMERA / SPACELAB 1  
OPERATION NO. 1

MISS-IM-OP*	DATE	TIME	* NAUTIR	* ALTITUDE	* ALTITUDE	* OVR*CLOUD*	* QUA*FILM*	* SUNEL*	COUNTRY							
NO.	* YY/MM/DD	HH:MM:SEC	* LAT.	LONG.	ROLL	PITCH	YAW	* CKM1	* LAP*COVER*LI *							
*	*	*	*	*	*	*	*	* %J	* X/10*TY *							
*	*	*	*	*	*	*	*	*STOP [SEC]	*							
01-0051-01*	83/12/2	3:35:16.593	40.27	99.55	-0.03	-0.09	0.03	250.790	60	0000	0	CIR	5.7	500	23.25*	CHINA
01-0052-01*	83/12/2	3:35:26.550	40.75	100.14	-0.04	-0.07	0.04	250.950	60	0000	0	CIR	5.7	500	23.04*	CHINA
01-0053-01*	83/12/2	3:35:36.436	41.21	100.73	-0.06	-0.06	0.04	251.100	60	0000	1	CIR	5.7	500	22.83*	CHINA
01-0054-01*	83/12/2	3:35:46.445	41.67	101.33	-0.07	-0.07	0.05	251.260	60	0000	1	CIR	5.7	500	22.62*	CHINA
01-0055-01*	83/12/2	3:35:56.390	42.13	101.94	-0.09	-0.07	0.04	251.420	60	0000	0	CIR	5.7	500	22.41*	CHINA

METRIC-CAMERA / SPACELAB 1  
OPERATION NO. 2

MISS-IM-CP*		DATE		TIME		NAIR		ATTITUDE		EXPOSURE		COUNTRY								
NO.	*	YY/MM/DD	HH:MM:SS.C	*	LAT.	LONG.	*	ROLL	PITCH	YAW	*	[KM]	*	*L%J*	X10*TY*	*	*STOP [SEC]	*	*CODEGJ*	*
U1-0056-02*	83/12/2	4:57: 2.651	16.42	56.61	-0.15	U.10	0.00	244.390	60	2121	X	CIR	6.8	500	28.66*ARAB SEA/MUSC & OM.					
U1-0057-U2*	83/12/2	4:57:12.646	16.97	56.96	-0.17	U.U9	0.02	244.490	60	1121	X	CIR	6.3	500	23.62*ARAB SEA/MUSCAT					
U1-0058-U2*	93/12/2	4:57:22.633	17.53	57.32	-0.18	U.07	0.04	244.580	60	1111	X	CIR	6.3	500	28.58*ARAB SEA/MUSCAT					
U1-0059-U2*	93/12/2	4:57:32.515	16.08	57.68	-0.19	U.07	0.05	244.680	60	0011	3	CIR	6.8	500	28.54*ARAB SEA/MUSCAT					
U1-0060-U2*	93/12/2	4:57:42.590	13.64	58.04	-0.16	U.07	0.07	244.780	60	0000	3	CIR	6.8	500	28.49*ARAB SEA/MUSCAT					
U1-0061-U2*	83/12/2	4:57:52.550	19.18	58.41	-0.15	U.07	0.08	244.890	60	0000	3	CIR	6.8	500	28.45*ARAB SEA/MUSCAT					
U1-0062-U2*	83/12/2	4:58: 2.511	19.74	58.77	-0.15	U.07	0.08	244.990	60	0000	3	CIR	6.8	500	28.39*ARAB SEA/MUSCAT					
U1-0063-U2*	83/12/2	4:58:12.474	20.28	59.14	-0.16	U.09	0.06	245.100	60	0000	0	CIR	6.9	500	28.34*ARAB SEA/MUSCAT					
U1-0064-U2*	83/12/2	4:58:22.413	20.33	59.52	-0.17	U.10	0.34	245.220	60	2000	0	CIR	6.3	500	29.28*ARAB SEA/MUSCAT					
U1-0065-U2*	83/12/2	4:58:32.563	21.37	59.89	-0.18	U.12	0.03	245.330	60	0010	0	CIR	6.0	500	28.21*ARAB SEA/MUSCAT					
U1-0066-U2*	83/12/2	4:58:42.523	21.92	60.27	-0.17	U.10	0.02	245.450	60	1100	0	CIR	6.8	500	28.14*ARAB SEA/MUSCAT					
U1-0067-U2*	83/12/2	4:58:52.503	22.46	60.65	-0.17	U.10	0.02	245.570	60	0012	0	CIR	6.9	500	28.06*ARAB SEA/MUSCAT					
U1-0068-U2*	83/12/2	4:59: 2.281	23.01	61.04	-0.17	U.09	0.01	245.690	60	0010	X	CIR	6.3	500	27.99*ARAB SEA/MUSCAT					
U1-0069-U2*	83/12/2	4:59:12.234	23.56	61.43	-0.15	U.09	0.00	245.810	60	0000	X	CIR	6.8	500	27.90*ARAB SEA/MUSCAT					
U1-0070-U2*	83/12/2	4:59:22.190	24.09	61.82	-0.16	U.10	0.00	245.940	60	0000	X	CIR	6.8	500	27.81*ARAB SEA IRAN					
U1-0071-U2*	83/12/2	4:59:32.134	24.63	62.21	-0.15	U.11	-0.01	246.070	60	0000	0	CIR	6.8	500	27.72*PAKISTAN					
U1-0072-U2*	83/12/2	4:59:42.103	25.17	62.62	-0.15	U.09	-0.02	246.200	60	0000	0	CIR	6.8	500	27.63*PAKISTAN					
U1-0073-U2*	83/12/2	4:59:52.056	25.71	63.02	-0.15	U.05	-0.03	246.330	60	0000	0	CIR	6.8	500	27.53*PAKISTAN					



STERIC-CAMERA / SPACELAB 1  
OPERATION NO. 3

MISSION-OPP*	DATE YY/MM/DD	TIME HH:MM:SEC	MAJOR			ALTITUDE [KMM]	OVERCLOUD [%]	EXPOSURE X10^6Y*	SUNEL*	COUNTRY					
			LAT.	LONG.	ROLL						PITCH	YAW	* [E] * X10^6Y *	* STOP [SEC] *	* F / 1/T * CDEG *
U1-0124-03*	83/12/2	6:29:20.404	26.09	40.62	-0.19	0.09	0.03	246.140	60	0010	2	CIR 6.8	500	27.23*	SAUDI ARABIA
U1-0125-03*	83/12/2	6:29:30.387	26.62	41.02	-0.18	0.06	0.01	246.280	60	0000	1	CIR 6.8	500	26.12*	SAUDI ARABIA
U1-0126-03*	83/12/2	6:29:40.552	27.16	41.44	-0.17	0.06	0.00	246.420	60	2000	1	CIR 6.8	500	27.01*	SAUDI ARABIA
U1-0127-03*	83/12/2	6:29:50.145	27.68	41.36	-0.17	0.05	-0.01	246.560	60	3010	1	CIR 6.8	500	26.91*	SAUDI ARABIA
U1-0128-03*	83/12/2	6:30:00.299	28.23	42.30	-0.16	0.06	-0.02	246.700	60	3230	2	CIR 6.8	500	26.79*	SAUDI ARABIA
U1-0129-03*	83/12/2	6:30:10.359	28.74	42.72	-0.15	0.06	-0.04	246.840	60	5552	2	CIR 6.8	500	26.67*	SAUDI ARABIA
U1-0130-03*	83/12/2	6:30:20.053	29.26	43.15	-0.16	0.07	-0.05	246.980	60	4444	4	CIR 6.8	500	26.56*	SAUDI ARABIA
U1-0131-03*	83/12/2	6:30:30.449	29.79	43.58	-0.15	0.07	-0.07	247.120	60	0145	5	CIR 6.7	500	26.43*SAUDI ARABIA, IRAQ	
U1-0132-03*	83/12/2	6:30:40.973	30.31	44.03	-0.16	0.04	-0.08	247.270	60	0101	7	CIR 6.7	500	26.30*	IRAQ
U1-0133-03*	83/12/2	6:30:50.086	30.34	44.48	-0.16	0.05	-0.09	247.420	60	0101	7	CIR 6.7	500	26.16*	IRAQ
U1-0134-03*	83/12/2	6:31:00.033	31.57	44.95	-0.17	0.02	-0.12	247.580	60	8823	7	CIR 6.7	500	26.03*	IRAQ
U1-0135-03*	83/12/2	6:31:10.083	31.88	45.40	-0.17	0.02	-0.12	247.720	60	9988	9	CIR 6.7	500	25.89*	IRAQ
U1-0136-03*	83/12/2	6:31:20.045	32.39	45.86	-0.18	0.02	-0.13	247.870	60	9999	X	CIR 6.7	500	25.74*	IRAQ
U1-0137-03*	83/12/2	6:31:29.996	32.90	46.33	-0.16	0.01	-0.15	248.030	60	9999	X	CIR 6.5	500	25.60*	IRAQ
U1-0138-03*	83/12/2	6:31:39.959	33.42	46.80	-0.16	0.01	-0.14	249.180	60						











METRIC-CAMERA / SPACELAB 1  
OPERATION NO. 8

MISS-IM-CP*	DATE	TIME	* NAUTR *	* ALTITUDE	* CLOUD*GUA*	FILM*	EXPOSURE	*SUNEL*
NO.	YY/MM/DD	HH:MM:SEC	LAT.	LONG.	PITCH	YAW	[CM]	1/T *CSEGJ*
*	*	*	*	*	*	*	*	*
U1-0342-08*	83/12/2	12:19:16.286	1.13	65.95	-0.07	-U.23	0.05	242.680
U1-0343-08*	83/12/2	12:19:25.366	1.70	-65.62	-U.08	-U.24	0.05	242.670
U1-0344-08*	83/12/2	12:19:36.265	2.26	-65.30	-U.09	-U.25	0.05	242.670
U1-0345-08*	83/12/2	12:19:45.150	2.82	-64.95	-0.12	-U.24	0.05	242.670
U1-0346-08*	83/12/2	12:19:55.232	3.39	64.65	-0.13	-U.26	0.05	242.680
U1-0347-08*	83/12/2	12:20:05.112	3.95	-64.33	-0.13	-U.26	0.05	242.690
U1-0348-08*	83/12/2	12:20:16.180	4.52	-64.00	-0.11	-U.26	0.05	242.710
U1-0349-08*	83/12/2	12:20:26.073	5.05	-63.68	-0.10	-U.26	0.06	242.730
U1-0350-08*	83/12/2	12:20:35.961	5.65	63.35	-0.10	-U.25	0.06	242.750
U1-0351-08*	83/12/2	12:20:46.051	6.21	-63.02	-C.10	-U.26	0.07	242.770
U1-0352-08*	83/12/2	12:20:55.931	6.77	62.70	-0.10	-U.25	0.08	242.800
U1-0353-08*	83/12/2	12:21:05.029	7.35	-62.37	-0.10	-U.25	0.07	242.840
U1-0354-08*	83/12/2	12:21:15.908	7.90	-62.04	-0.10	-U.24	0.07	242.870
U1-0355-08*	83/12/2	12:21:25.779	8.46	-61.71	-0.09	-U.24	0.07	242.910
U1-0356-08*	83/12/2	12:21:35.866	9.03	-61.39	-0.09	-U.25	0.06	242.960
U1-0357-08*	83/12/2	12:21:45.757	9.59	-61.05	-0.08	-U.24	0.07	243.000
U1-0358-08*	83/12/2	12:21:55.841	10.16	-60.71	-0.08	-U.25	0.07	243.060



METRIC-CAMERA / SPACELAB 1  
OPERATION NO. 12

NO.	* YY/MM/DD	TIME HH:MM:SEC	* LAT.	LONG.	* ROLL	PITCH	YAW *	ALTITUDE * CLOUJ*QUA*FILE* [KM]	*LAP*CAVE*LI* *[%]*	EXPOSURE *SUNEL* *F/* 1/T *SEGJ* *STOP [SECJ]*	COUNTRY	
U1-0408-12	93/12/	3:23:	3.675	35.31	90.97	-0.05	-0.21	0.05	246.970	80	0001	1
U1-0409-12*	83/12/	3:23:	8.389	35.55	91.20	-0.05	-0.18	0.07	247.140	80	0000	1
U1-0410-12*	83/12/	3:23:13	3.510	35.79	91.45	-0.02	-0.15	0.07	247.120	80	0000	0
91-0411-12*	83/12/	3:23:18	2.236	36.04	91.70	-0.01	-0.13	0.06	247.190	80	0000	0
U1-0412-12*	83/12/	3:23:23	1.158	36.28	91.96	0.02	-0.09	0.06	247.270	80	0000	0
U1-0413-12*	83/12/	3:23:28	2.273	36.53	92.23	0.03	-0.06	0.06	247.350	80	0000	0
U1-0414-12*	83/12/	3:23:33	1.184	36.78	92.49	0.06	-0.04	0.05	247.420	80	0000	0
U1-0415-12*	83/12/	3:23:38	1.03	37.02	92.74	0.08	-0.01	0.05	247.500	80	0000	0
U1-0416-12*	83/12/	3:23:43	0.923	37.27	93.00	0.10	0.01	0.05	247.570	80	0000	0
U1-0417-12*	83/12/	3:23:48	1.139	37.52	93.29	0.10	0.04	0.05	247.660	80	0000	0
U1-0418-12*	83/12/	3:23:53	0.056	37.76	93.55	0.10	0.05	0.05	247.730	80	0000	0
U1-0419-12*	93/12/	3:23:57	9.981	39.00	93.91	0.10	0.05	0.04	247.800	80	0000	0
U1-0420-12*	23/12/	3:24:	2.906	38.24	94.08	0.10	0.04	0.04	247.880	80	0000	0
U1-0421-12*	93/12/	3:24:	8.021	38.48	94.26	0.09	0.04	0.04	247.960	80	0000	0
U1-0422-12*	83/12/	3:24:12	9.933	38.72	94.63	0.09	0.02	0.04	248.040	80	1100	0
U1-0423-12*	83/12/	3:24:17	8.852	38.96	94.91	0.08	0.01	0.04	248.110	80	2200	1
U1-0424-12*	83/12/	3:24:22	7.782	39.20	95.18	0.08	0.01	0.03	248.190	80	2500	1
U1-0425-12*	33/12/	3:24:27	6.991	39.45	95.47	0.05	-0.01	0.03	248.270	80	2222	1
U1-0426-12*	83/12/	3:24:32	8.811	39.68	95.76	0.07	-0.02	0.03	248.350	80	5143	4
U1-0427-12*	83/12/	3:24:37	7.733	39.91	96.04	0.06	-0.03	0.02	248.420	80	5023	5
U1-0428-12*	83/12/	3:24:42	6.658	40.14	96.32	0.06	-0.03	0.02	248.500	80	4232	5
U1-0429-12*	23/12/	3:24:47	7.782	40.39	96.62	0.06	-0.05	0.01	248.570	80	4551	6
U1-0430-12*	63/12/	3:24:52	7.701	40.62	96.91	0.07	-0.06	0.01	248.650	80	6651	7
U1-0431-12*	33/12/	3:24:57	6.619	40.85	97.20	0.05	-0.07	0.01	248.730	80	9754	7





METRIC-CAMERA / SPACELAB 1  
OPERATION NO. 13

MISSION-OP#	DATE NO.	TIME HH:MM:DD	NADIR LAT. °	ROLL °	ALTITUDE PITCH °	YAW °	CLOUDS [KM]	EXPOSURE *SUNEL *F/ 1/T * [DEG] *STOP [SEC]	COUNTRY	
										*
U1-0532-13*	83/12/3	4:50:39.519	30.00	63.40	-0.11	-0.01	245.320	60	0.000 0	CIR 6.2 500 23.16* AFGHANISTAN
U1-0534-13*	83/12/3	4:50:49.597	30.52	63.84	-0.11	-0.01	245.460	60	0.000 0	CIR 6.2 500 23.07* AFGHANISTAN
U1-0534-13*	83/12/3	4:50:59.460	31.05	64.30	-0.11	0.00	245.610	60	0.000 0	CIR 6.2 500 22.99* AFGHANISTAN
U1-0535-13*	83/12/3	4:51:59.509	31.56	64.74	-0.09	0.02	245.750	60	0.000 0	CIR 6.2 500 22.90* AFGHANISTAN
U1-0535-13*	83/12/3	4:51:19.366	32.08	65.21	-0.09	0.01	245.900	60	0.000 0	CIR 6.2 500 22.81* AFGHANISTAN
U1-0537-13*	83/12/3	4:51:29.247	32.59	65.67	-0.08	0.04	246.050	60	0.000 0	CIR 6.2 500 22.71* AFGHANISTAN
U1-0538-13*	83/12/3	4:51:39.317	33.10	66.15	-0.07	0.05	246.200	60	0.000 0	CIR 6.2 500 22.62* AFGHANISTAN
U1-0539-13*	83/12/3	4:51:49.375	33.62	66.63	-0.07	0.04	246.350	60	0.000 0	CIR 6.2 500 22.51* AFGHANISTAN
U1-0540-13*	83/12/3	4:51:59.262	34.12	67.11	-0.07	0.05	246.500	60	0.000 0	CIR 6.2 500 22.41* AFGHANISTAN
U1-0541-13*	83/12/3	4:52:59.323	34.64	67.61	-0.07	0.02	246.650	60	0.000 0	CIR 6.2 500 22.29* AFGHANISTAN
U1-0542-13*	83/12/3	4:52:19.172	35.13	68.10	-0.08	0.00	246.800	60	2.000 0	CIR 6.2 500 22.19* AFGHANISTAN
U1-0543-13*	83/12/3	4:52:29.240	35.64	68.61	-0.08	-0.03	246.960	60	1.110 1	CIR 6.2 500 22.07* AFGHANISTAN
U1-0544-13*	83/12/3	4:52:39.121	36.13	69.12	-0.10	-0.05	247.110	60	1.210 1	CIR 6.2 500 21.96* AFGHANISTAN
U1-0545-13*	83/12/3	4:52:49.193	36.63	69.65	-0.11	-0.07	247.260	60	0.002 1	CIR 6.2 500 21.84* AFGHANISTAN
U1-0546-13*	83/12/3	4:52:59.051	37.12	70.16	-0.11	-0.08	247.420	60	5.300 1	CIR 6.2 500 21.71* AFGHANISTAN

METRIC-CAMERA / SPACELAB 1  
OPERATION NO. 14

MISS-IM-OP*	DATE NO.	TIME YY/MM/DD	HH:MM:SEC	* LAT.	LONG.	* ROLL	PITCH	YAW	* ALTITUDE [KM]	* CLOUDS QUA	* FILM LAP	* OVERFLI X/10*TY	* SUNEL*	EXPOSURE *F/ 1/T *DEGJ*	COUNTRY	
																* [L%]*
01-0548-14*	83/12/ 3	6:11: 2.475	0.00	21.30	-1.07	-0.38	-0.09	241.350	60	9978	X	B/W	5.7	950	22.36*	REP.OF CONGO
01-0549-14*	83/12/ 3	6:11:12.351	0.36	21.63	-0.94	-0.63	-0.07	241.340	60	8788	X	B/W	5.7	975	22.34*	REP.OF CONGO
01-0550-14*	83/12/ 3	6:11:12.322	1.43	21.95	-0.97	-0.63	-0.04	241.330	60	8898	X	B/W	5.7	950	22.43*	REP.OF CONGO
01-0551-14*	83/12/ 3	6:11:12.202	1.99	22.27	-1.00	-0.63	-0.02	241.320	60	7288	B	B/W	5.7	975	22.53*	REP.OF CONGO
01-0552-14*	83/12/ 3	6:11:12.257	2.56	22.60	-1.02	-0.62	0.00	241.320	60	9572	B	B/W	5.7	950	22.63*	REP.OF CONGO
01-0553-14*	83/12/ 3	6:11:52.186	3.12	22.92	-1.01	-0.60	0.03	241.320	60	8695	9	B/W	5.7	1000	22.71*	REP.OF CONGO
01-0554-14*	83/12/ 3	6:12: 2.140	3.69	23.24	-1.01	-0.58	0.05	241.320	60	6486	B	B/W	5.7	1000	22.79*	REP.OF CONGO
01-0555-14*	83/12/ 3	6:12:12.183	4.26	23.57	-1.00	-0.55	0.08	241.330	60	0075	5	B/W	5.7	1000	22.87*R.0	CONGO+C.AF.REP
01-0556-14*	83/12/ 3	6:12:12.139	4.82	23.89	-1.00	-0.52	0.11	241.340	60	0011	2	B/W	5.7	1000	22.96*R.0	CONGO+C.AF.REP
01-0557-14*	83/12/ 3	6:12:32.086	5.38	24.22	-1.00	-0.50	0.13	241.350	60	0000	1	B/W	5.7	1000	23.04*R.0	CONGO+C.AF.REP
01-0558-14*	83/12/ 3	6:12:42.045	5.95	24.55	-0.99	-0.46	0.16	241.370	60	0100	3	B/W	5.7	1000	23.11*R.0	CONGO+C.AF.REP
01-0559-14*	83/12/ 3	6:12:52.002	6.51	24.87	-0.99	-0.41	0.19	241.400	60	0001	5	B/W	5.7	1000	23.17*	C.AFR REP.
01-0560-14*	83/12/ 3	6:13: 1.960	7.07	25.20	-0.99	-0.37	0.22	241.420	60	3100	7	B/W	5.7	1000	23.24*	C.AFR REP.
01-0561-14*	83/12/ 3	6:13:11.914	7.64	25.53	-0.98	-0.32	0.25	241.450	60	1251	8	B/W	5.7	975	23.30*	C.AFR REP.+SUDAN

METRIC-CAMERA / SPACELAB 1  
OPERATION NO. 22

- 83 -

MISS-IM-OP*	DATE	TIME	* NAIR	* LAT.	LONG.	* ROLL	PITCH	YAW *	ATTITUDE	*ALTITUDE	*OVR*CLOUD*QUA*FILM*	EXPOSURE	*SUNEL*	COUNTRY
NO.	YY/MM/DD	HH:MM:SEC	*	*	*	*	*	*	[KMH]	*LAP*COVER*LI*	*[C%]* X/10*TY*	*F/	1/T	*[DEG]*
U1-0596-22*	83/12/5	4:26:2.572	24.52	53.40	0.41	-0.54	0.22	239.640	60	2400	4	B/W	5.7	650
U1-0597-22*	83/12/5	4:26:13.209	25.04	53.79	0.20	-0.67	0.15	239.740	60	6825	3	B/W	5.7	650
U1-0598-22*	83/12/5	4:26:22.289	25.58	54.20	0.20	-0.61	0.09	239.840	60	8845	X	B/W	5.7	650
U1-0599-22*	83/12/5	4:26:32.215	26.12	54.60	0.22	-0.54	0.04	239.950	60	0785	5	B/W	5.7	650
U1-0600-22*	83/12/5	4:26:42.122	26.65	55.01	0.21	-0.54	0.02	240.060	60	0017	1	B/W	5.7	650
U1-0601-22*	83/12/5	4:26:52.058	27.18	55.42	0.18	-0.44	0.01	240.160	60	0000	1	B/W	5.7	650
U1-0602-22*	83/12/5	4:27:1.983	27.72	55.84	0.14	-0.34	0.01	240.280	60	0000	0	B/W	5.7	650
U1-0603-22*	83/12/5	4:27:11.906	28.24	56.26	0.11	-0.25	0.01	240.390	60	0000	0	B/W	5.7	650
U1-0604-22*	83/12/5	4:27:21.957	29.78	56.70	0.07	-0.16	0.00	240.500	60	0000	0	B/W	5.7	650
U1-0605-22*	83/12/5	4:27:31.865	29.30	57.13	0.02	-0.06	0.01	240.620	60	0000	0	B/W	5.7	650
U1-0606-22*	83/12/5	4:27:51.780	29.82	57.56	-0.03	0.01	0.01	240.730	60	0000	0	B/W	5.7	650
U1-0607-22*	83/12/5	4:27:51.704	30.35	58.01	-0.07	0.02	0.00	240.850	60	0000	0	B/W	5.7	650



METRIC-CAMERA / SPACELAB 1  
OPERATION NO. 23

MISSION-OP*	DATE	TIME	* NADIR	ALTITUDE	* OVR*CL QUD*QUA*FLN*	EXPOSURE	* SUNEL*	COUNTRY								
								NO.	* YY/MM/DD	HH:MM:SEC	* LAT.	LONG.	* ROLL	PITCH	YAW	[KM]
U1-0658-23*	83/12/5	5:57:20.099	31.05	35.95	-0.11	-0.02	0.00	240.930	60	5979	5	B/W	5.7	625	15.35*	JORDAN
U1-0659-23*	83/12/5	5:57:30.138	31.57	36.41	-0.12	0.01	-0.01	241.060	60	6949	5	B/W	5.7	625	15.36*	JORDAN
U1-0660-23*	83/12/5	5:57:40.007	32.09	36.86	-0.13	-0.01	0.00	241.180	60	6969	5	B/W	5.7	625	15.35*	JORDAN
U1-0661-23*	83/12/5	5:57:50.045	32.60	37.33	-0.14	-0.02	0.00	241.300	60	2669	7	B/W	5.7	625	15.36*	JORDAN,SYRIA
U1-0662-23*	83/12/5	5:58:0.074	33.13	37.82	-0.13	-0.03	0.01	241.430	60	5999	7	B/W	5.7	625	15.36*	SYRIA
U1-0663-23*	83/12/5	5:58:9.939	33.62	38.29	-0.13	-0.04	0.01	241.560	60	5778	8	B/W	5.7	625	15.36*	SYRIA
U1-0664-23*	83/12/5	5:58:19.976	34.13	38.78	-0.12	-0.04	0.02	241.690	60	1567	7	B/W	5.7	625	15.36*	SYRIA,IRAQ
U1-0665-23*	83/12/5	5:58:29.961	34.64	39.26	-0.11	-0.05	0.03	241.820	60	3116	5	B/W	5.7	625	15.35*	SYRIA,IRAQ
U1-0666-23*	83/12/5	5:58:39.988	35.14	39.77	-0.11	-0.03	0.04	241.950	60	0222	3	B/W	5.7	625	15.34*	SYRIA,IRAQ
U1-0667-23*	83/12/5	5:58:49.924	35.65	40.27	-0.10	-0.03	0.05	242.080	60	4211	3	B/W	5.7	625	15.33*	SYRIA,IRAQ
U1-0668-23*	83/12/5	5:58:59.791	36.15	40.79	-0.11	-0.02	0.04	242.220	60	3033	3	B/W	5.7	625	15.32*	SYRIA,IRAQ



METRIC-CAMERA / SPACELAB 1  
OPERATION NO. 24

MISS-IM-CP*	DATE NO.	TIME YY/MM/DD	TIME HH:MM:SEC	NAUTR * LAT. * LONG.	* ROLL	PITCH	YAW	* ALTITUDE [KM]	* OVR*CLOUD*QUA*FILM*	EXPOSURE *LAP*COVER*I * [L%]* X/10*TY*	*SUNEL* 1/T *COEGJ* *STOP [SECJ]*	COUNTRY
01-0719-24*	93/12/5	7:27:20.277	33.38	15.40	-0.03	0.06	0.03	241.430	60	9999	X	B/W
01-0720-24*	83/12/5	7:27:30.506	33.90	15.38	-0.02	0.09	0.05	241.550	60	9999	X	B/W
01-0721-24*	83/12/5	7:27:40.150	34.40	16.36	-0.02	0.02	0.05	241.680	60	9999	X	B/W
01-0722-24*	83/12/5	7:27:50.158	34.91	16.36	-0.02	0.08	0.05	241.810	60	9999	X	B/W
01-0723-24*	83/12/5	7:28:00.178	35.43	17.38	-0.02	0.06	0.06	241.940	60	9999	X	B/W
01-0724-24*	83/12/5	7:28:10.152	35.91	17.88	-0.05	0.05	0.03	242.080	60	9999	X	B/W
01-0725-24*	83/12/5	7:28:20.034	36.41	18.39	-0.03	0.05	0.05	242.210	60	9999	X	B/W
01-0726-24*	83/12/5	7:28:29.992	36.90	18.91	-0.02	0.06	0.03	242.340	60	9999	X	B/W
01-0727-24*	83/12/5	7:28:40.105	37.40	19.45	-0.04	0.06	0.00	242.480	60	9999	X	B/W
01-0728-24*	83/12/5	7:28:50.057	37.89	19.39	-0.05	0.07	0.00	242.610	60	9999	X	B/W
01-0729-24*	83/12/5	7:29:00.016	38.39	20.55	-0.06	0.08	-0.03	242.740	60	7999	X	B/W
01-0730-24*	83/12/5	7:29:09.331	38.85	21.03	-0.07	0.09	-0.04	242.870	60	7989	X	B/W
01-0731-24*	83/12/5	7:29:19.882	39.33	21.64	-0.07	0.09	-0.05	243.010	60	9999	X	B/W
01-0732-24*	83/12/5	7:29:29.004	39.81	22.21	-0.07	0.06	-0.04	243.140	60	9989	X	B/W
01-0733-24*	83/12/5	7:29:39.913	40.29	22.79	-0.06	0.05	-0.02	243.280	60	9989	X	B/W
01-0734-24*	93/12/5	7:29:44.863	40.76	23.38	-0.05	0.03	-0.02	243.410	60	9999	X	B/W
01-0735-24*	83/12/5	7:29:59.935	41.23	23.99	-0.04	0.01	0.00	243.550	60	9999	X	B/W







METRIC-CAMERA / SPACELAB 1  
OPERATION NO. 29

MISSION-OP#	DATE	TIME	NADIR	ALTITUDE	EXPOSURE	SUNEL*	COUNTRY	QUA*FILM*					
								NO.	* YY/MM/DD	HH:MM:SEC	* LAT.	LONG.	* ROLL
U1-0899-29*	83/12/	5 13:19:	2.530	16.32 -88.21	0.01 -U.15 -0.07	238.260	60	5271	5	B/W	5.7	525	12.81*
U1-0900-29*	83/12/	5 13:19:12.310	16.87 -87.86	0.02 -U.13 -0.07	238.320	60	3162	5	B/W	5.7	525	12.88*	
U1-0901-29*	83/12/	5 13:19:22.205	17.42 -87.50	0.03 -U.12 -0.07	238.390	60	6321	5	B/W	5.7	550	12.95*	
U1-0902-29*	83/12/	5 13:19:32.217	17.98 -87.14	0.05 -U.12 -0.06	238.450	60	2141	4	B/W	5.7	550	13.02*	
U1-0903-29*	83/12/	5 13:19:42.065	18.52 -86.78	0.08 -U.12 -0.06	238.520	60	1211	4	B/W	5.7	550	13.09*	
U1-0904-29*	83/12/	5 13:19:52.088	19.08 -86.42	0.07 -U.11 -0.05	238.590	60	1221	X	B/W	5.7	550	13.15*	
U1-0905-29*	83/12/	5 13:20:2.105	19.63 -86.05	0.07 -U.09 -0.04	238.570	60	1211	X	B/W	5.7	550	13.21*	
U1-0906-29*	83/12/	5 13:20:1.955	20.17 -85.68	0.07 -U.11 -0.04	238.740	60	1222	X	B/W	5.7	550	13.27*	
U1-0907-29*	83/12/	5 13:20:21.902	20.72 -85.31	0.06 -U.11 -0.04	238.820	60	3222	X	B/W	5.7	550	13.33*	
U1-0908-29*	83/12/	5 13:20:31.029	21.22 -84.96	0.06 -U.12 -0.04	238.900	60	3252	5	B/W	5.7	550	13.39*	
U1-0909-29*	83/12/	5 13:20:41.045	21.77 -84.59	0.07 -U.12 -0.04	238.980	60	0543	4	B/W	5.7	550	13.44*	
U1-0910-29*	83/12/	5 13:20:51.058	22.36 -84.17	0.07 -U.13 -0.04	239.080	60	0304	3	B/W	5.7	550	13.50*	
U1-0911-29*	83/12/	5 13:21:1.683	22.90 -83.79	0.06 -U.14 -0.03	239.160	60	0303	3	B/W	5.7	550	13.55*	



METRIC-CAMERA / SPACELAB 1  
OPERATION NO. 33

MISS-NUM-OP#	DATE	TIME	NAUTR	ALTITUDE	PATCH	YAW	COVER	EXPOSURE	SUNEL	COUNTRY	
U1-0953-35*	83/12/ 7	8:33:16.624	38.19	-7.52	0.09	0.02	0.00	239.090	60	9597	3
U1-0954-35*	83/12/ 7	8:33:26.332	38.66	-6.99	0.07	0.01	0.02	239.240	60	9995	3
U1-0955-35*	83/12/ 7	8:33:36.286	39.15	-6.43	0.05	0.00	0.02	239.380	60	9889	3
U1-0956-35*	83/12/ 7	8:33:46.271	39.63	-5.87	0.05	-u.u1	0.04	239.520	60	4699	3
U1-0957-35*	83/12/ 7	8:33:56.194	40.10	-5.29	0.02	-u.u2	0.05	239.670	60	0656	1
U1-0958-35*	83/12/ 7	8:34: 6.103	40.57	-4.71	0.01	-u.u2	0.06	239.810	60	0406	1
U1-0959-35*	83/12/ 7	8:34:16.005	41.03	-4.12	-0.02	-u.u2	0.06	239.950	60	004	1
U1-0960-35*	83/12/ 7	8:34:25.893	41.50	-3.53	-0.02	-u.u3	0.08	240.100	60	001	1
U1-0961-35*	83/12/ 7	8:34:36.097	41.97	-2.90	-0.04	-u.u3	0.09	240.250	60	5000	1
U1-0962-35*	83/12/ 7	8:34:46.030	42.43	-2.28	-0.06	-u.u2	0.07	240.390	60	7040	1
U1-0963-35*	83/12/ 7	8:34:55.953	42.88	-1.65	-0.08	-u.u1	0.06	240.530	60	9170	1
U1-0964-35*	83/12/ 7	8:35: 5.060	43.23	-1.07	-0.09	u.u1	0.05	240.670	60	8790	1
U1-0965-35*	83/12/ 7	8:35:15.791	43.76	-0.37	-0.07	u.u3	0.03	240.810	60	1796	3
U1-0966-35*	83/12/ 7	8:35:25.679	44.20	0.00	-u.u7	u.u6	0.44	240.960	60	0298	3
U1-0967-35*	83/12/ 7	8:35:35.882	44.65	0.97	-0.05	u.u6	0.02	241.110	60	2303	3
U1-0968-35*	83/12/ 7	8:35:45.757	45.07	1.64	-0.04	u.u9	0.01	241.240	60	8423	3
U1-0969-35*	83/12/ 7	8:35:55.595	45.50	2.33	-0.11	u.u8	0.01	241.380	60	9983	3
U1-0970-35*	83/12/ 7	8:36: 5.753	45.93	3.04	-0.01	u.u7	-0.01	241.520	60	7998	9
U1-0971-35*	83/12/ 7	8:36:15.610	46.34	3.75	0.00	u.u7	-0.02	241.660	60	7989	9
U1-0972-35*	83/12/ 7	8:37:15.125	48.72	8.26	0.08	u.u5	-0.08	242.470	60	9948	7
U1-0973-35*	83/12/ 7	8:37:24.663	49.09	9.02	0.10	u.u6	-0.09	242.590	60	9998	7
U1-0974-35*	83/12/ 7	8:37:34.813	49.46	9.84	0.09	u.u6	-0.10	242.720	60	6999	7
U1-0975-35*	83/12/ 7	8:37:44.693	49.82	10.66	0.09	u.u7	-0.11	242.850	60	9969	7
U1-0976-35*	83/12/ 7	8:37:54.772	50.19	11.51	0.08	u.u8	-0.12	242.980	60	7789	7





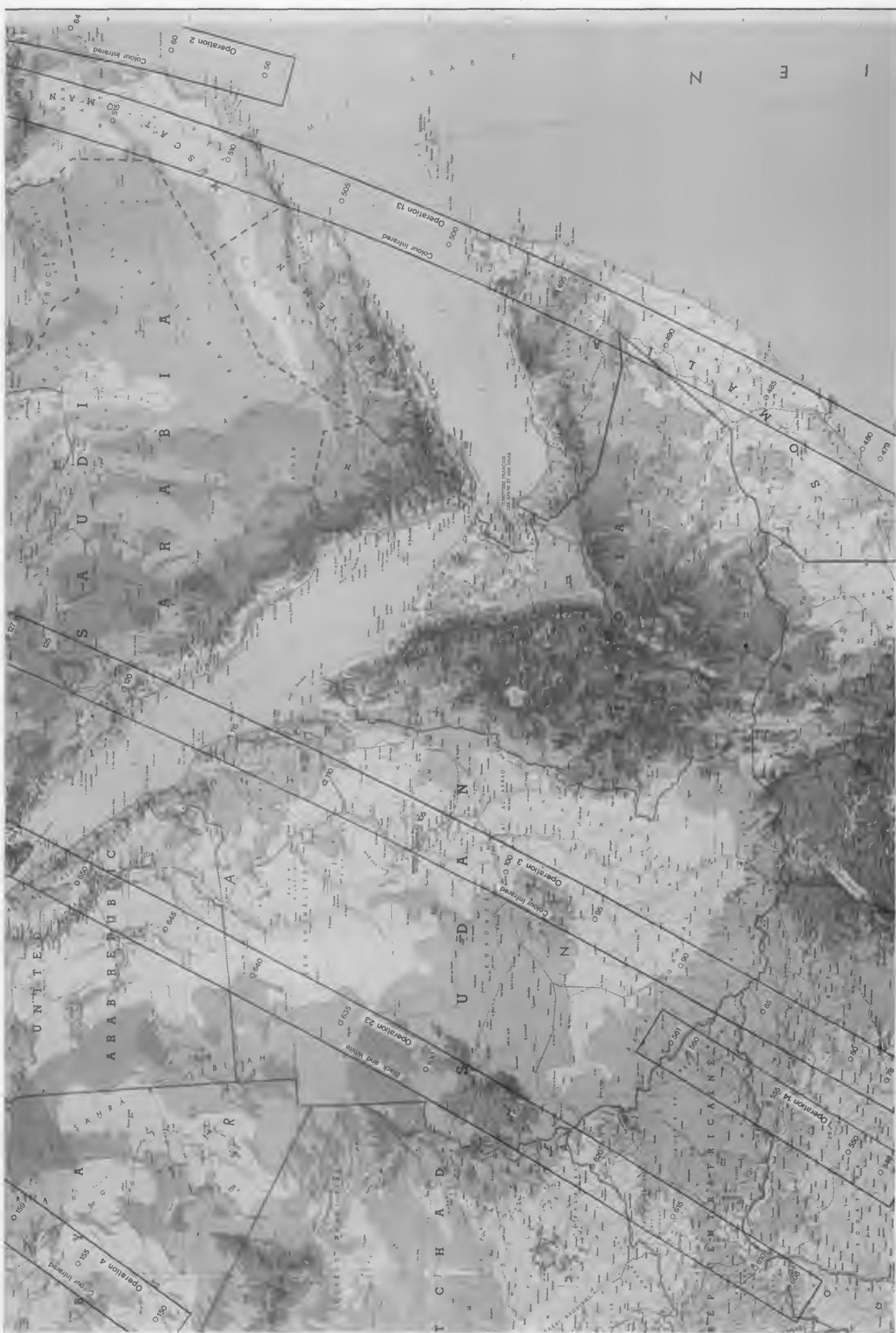
## 6.2 Geographical Reference

In the following 10 map sheets, the photographed ground strips are drawn as follows; each strip is characterized by the operation no. and the film type used for this operation (Colour Infrared or Black and White). The center of every fifth image frame is indicated inside each strip by a small circle and the corresponding image no; in the same manner, the first and the last image of each strip are also marked.

These maps can be used for easy identification of the geographical location belonging to a certain image no.

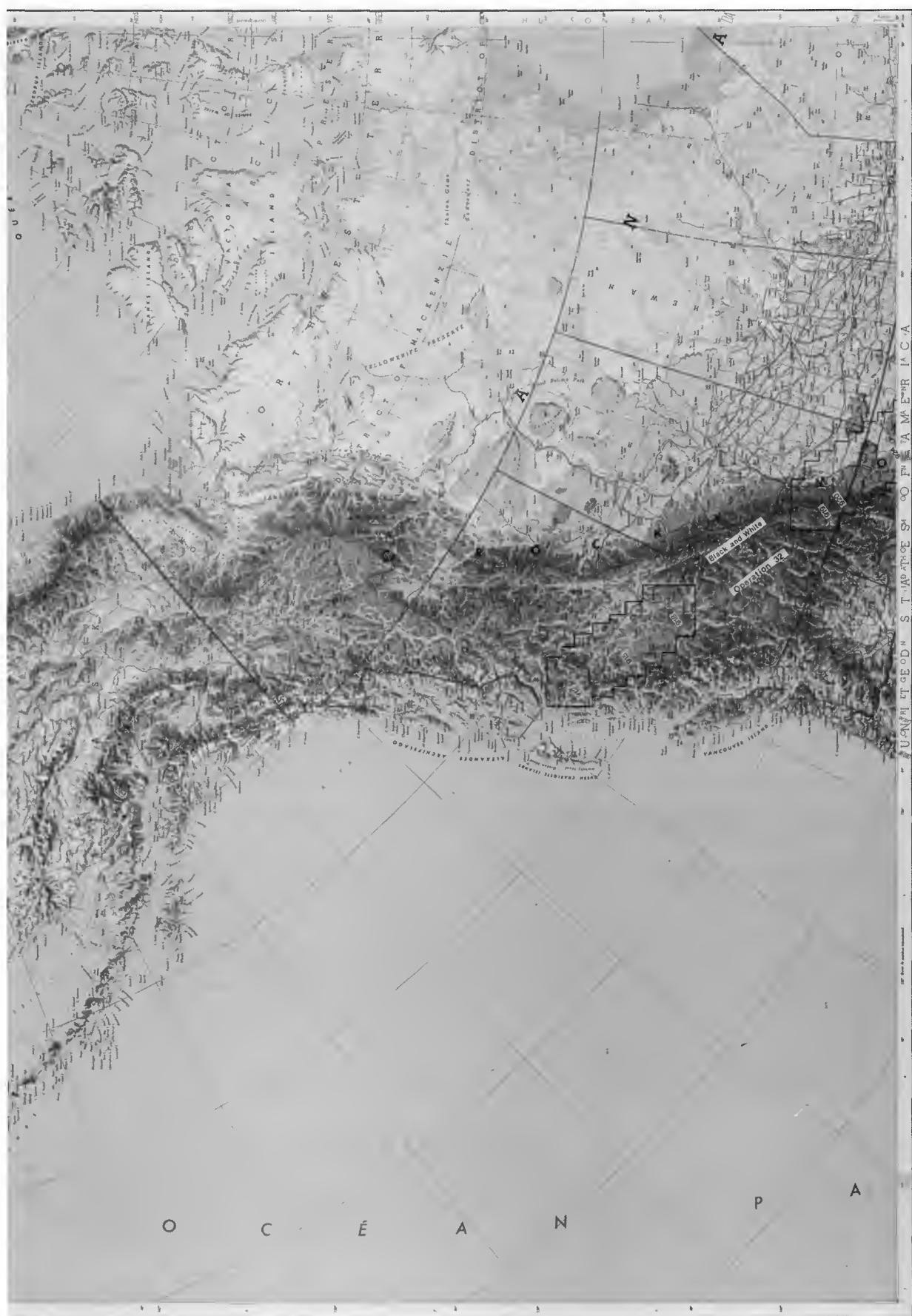
Remark: Image No. 74-76 of Operation No. 3 are not displayed.











U.S. GEOLOGICAL SURVEY MAP

Scale 1:625,000

